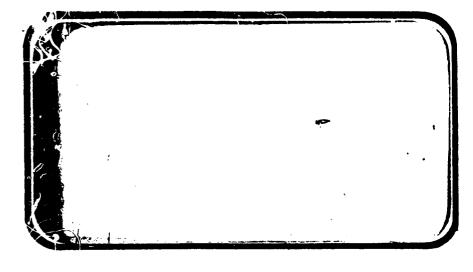
HOUSTON, TEXAS





SPACE SHUTTLE

LEROTHERMODYNAMIC DATA REPORT

JOHNSON SPACE CENTER

DATA MANagement services

DMS-DR-2082 NASA-CR-128,800

EFFECTS OF REACTION CONTROL SYSTEM JET SIMULATION

ON THE STABILITY AND CONTROL CHARACTERISTICS OF A

0.015-SCALE SPACE SHUTTLE ORBITER MODEL IN THE

AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL

(OA73)

Ву

T. J. Dziubale and John Marroquin, Rockwell International J. W. Gleary and J. A. Mellenthin, NASA Ames

Prepared under NASA Contract Number NAS9-13247

Ву

Data Management Services Chrysler Corporation Space Division New Orleans, La. 70189

for

Engineering Analysis Division

Johnson Space Center National Aeronautics and Space Administration Houston, Texas

WIND TUNNEL TEST SPECIFICS

Test Number:

ARC 3.5-167

NASA Series Number: 0A73

Test Dates:

July 11 to July 18, 1973

Model Number:

42-0

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Chrysler Corporation Space Division assumes no responsibility for the data presented herein other than its display characteristics. ON THE STABILITY AND CONTROL CHARACTERISTICS OF A

O.015-SCALE SPACE SHUTTLE ORBITER MODEL IN THE

AMES RESEARCH CENTER 3.5-FOOT HYPERSONIC WIND TUNNEL (0A73)

By T. J. Dziubala and John Marroquin J. W. Cleary* and J. A. Mellenthin*

ABSTRACT

An experimental investigation was performed in the Ames Research Center 3.5-Foot Hypersonic Wind Tunnel (Test 0A73) to obtain detailed effects which interactions between the RCS jet flow field and the local orbiter flow field have on orbiter hypersonic stability and control characteristics. Six-component force data were obtained through an angle-of-attack range of 15 to 35 degrees with 0° angle of sideslip. The test was conducted with yaw, pitch and roll jet simulation at a free-stream Mach number of 10.3. These data simulate two (2) SSV re-entry flight conditions at Mach numbers of 28.3 and 10.3. Fuselage base pressures and pressures on the non-metric RCS pods were obtained in addition to the basic force measurements. Model 42-0 was used for these tests.

* NASA Ames

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TABLE OF CONTENTS

	Page
ABSTRACT	111
INDEX OF MODEL FIGURES	2
INDEX OF DATA FIGURES	3
NOMENCLATURE	4
INTRODUCTION	7
CONFIGURATIONS INVESTIGATED	11
TEST FACILITY DESCRIPTION	13
DATA REDUCTION	14
REFERENCES	16
TABLES	
I TEST CONDITIONS	18
II DATA SET COLLATIONS	
A. OIL FLOW DATA	19
B. RCS ON FORCE DATA	20
C. RCS OFF FORCE DATA	22
III MODEL DIMENSIONAL DATA	24
IV RCS DIRECT IMPINGEMENT FORCE DATA	37
V NOZZLE CALABRATION CURVES	38
FIGURES	
MODEL	43
DATA	51
APPENDIX - TABULATED SOURCE DATA	

THE RESERVE THE PROPERTY OF THE PARTY OF THE

INDEX OF DATA FIGURES

FIGURE	Title	COEFFICIENTS SCHEDULE	CONDITIONS VARYING	PAGES
. व	Effects of RCS jet flowfield interactions (yaw), Epsilon = 1.159	⋖	Elevon and body	1-12
۶.	Effects of RCS jet flowfield interactions (roll), Epsilon = 1.159	⋖	Elevon and body flap deflections	13-72
•	Effects of RCS jet flowfield interactions (pitch up), Epsilon = 1.159	⋖	Elevon and body flap deflections	13-90
7.	Effects of RCS jet flowfield interactions (pitch down), Epsilon = 1.159	⋖	Elevon and body	91-150
ထံ	Effects of RCS jet flowfid. interact. True M = 10.29, Yaw Sim., Epsilon = 10.62	⋖	Elevon and body flap deflections	121-144
•	Effects of RCS jet flowfld. interact. Alt. Roll mode, Epsilon = 1.159	⋖	Elevon and body flap deflections	145-156

COEFFICIENTS SCHEDULE:

(A) CH, CLM, CA, CY, CYH, AND CEL vs. Alpha

- 10 mmの 10 m

NOMENCLATURE General

SYMBOL	Sadrac Symbol	DESTRICTION
		speed of sound; m/sec, ft/sec
Cp	CP	pressure coefficient; (p ₁ - p _m)/q
M	MACE	Mach number; V/a
D		prossure; I/m², pef
q	Q(NSM) Q(PSF)	dynamic pressure; 1/2, 1/2, 1/m ² , pef
RM/L	RM/L	unit Reynolds number; per m, per ft
٧		velocity; m/sec, ft/sec
•	ALPHA	angle of attack, degrees
β	BETA	angle of sideslip, degrees
ψ	PSI	angle of yaw, degrees
•	PHI	angle of roll, degrees
P		mass density; kg/m3, slugs/ft3
	ļ	Reference & C.G. Definitions
Ab.		bese eres; m ² , ft ²
b	RIF	wing spen or reference spen; m, ft
c.g.		center of gravity
REF	iref	reference length or wing meen aerodynamic chord; m, ft
8	SRET	wing area or reference area; m2, ft2
	MRP	moment reference point
Xcg	XXX	moment reference point on X exis
	MRP	ment reference point on Y axis
Z _{cs}	ZRP	moment reference point on Z axis
SUBSCR!	CPTS	
b		base
1.		10001
9 👂		static conditions
ŧ		total conditions free streem
-		TEGS DIFEST

NOMENCIATURE (Continued) Body-Axia System

SYMBOL	SADSAC SYMBOL	DEFINITION
C _M	CIN	normal-force coefficient; normal force
C _A	CA	exial-force coefficient; exial force
C _Y	CY	side-force coefficient; side force
c_{A_b}	CAB	base-force coefficient; base force
		-A _b (p _b - p ₂)/q8
CAf	CAP	forebody exial force coefficient, c_A - c_{A_b}
C _{IRI}	CIM	pitching-moment coefficient; pitching moment
c _n	CYN	yawing-moment coefficient; yawing moment qSb
c ₁	CBL	rolling-moment coefficient; rolling moment
, .		Stebility-Axis System
C _L	CL	lift coefficient; 11ft q8
CD	CD	drag coefficient; drag
c_{D_D}	CDB	base-drag coefficient; base drag
$c_{D_{\underline{r}}}$	CDF	forebody drag coefficient; CD - CDb
C _Y	CY	side-force coefficient; side force
C _m	CIM	pitching-moment coefficient; pitching moment quicking
c _n	CLN	yawing-moment coefficient; yawing moment q8b
c _L	CSL	rolling-moment coefficient; rolling moment
r/p	T/D	lift-to-drag ratio; $c_{ m I}/c_{ m D}$

The second secon

NOMENCLATURE (Continued)

ADDITIONS TO STANDARD LIST

	PLCT	
SYMBOL	SYMBOL	DEFINITION
A _{DM}		OMS pod base area, ft ²
Pt		freestream total pressure, psia
Pe	PC	model RCS plenum chamber pressure, psia
Tt		freestream total temperature, °R
R/ft	RN/L	freestream unit Reynolds number, per foot
X _{CP} /£ _B	xcp/L	longitudinal center of pressure location, fraction of body length
⁶ BF	BDFLAP	body flap deflection angle, deg.
δ _e	ELEVON	elevon deflection angle, deg.
δ _{SB}	SPDBRK	speed brake deflection angle, deg.
E	epsilon	model OMS nozzle expansion ratio

Ë

INTRODUCTION

Investigations were performed to determine interaction effects of Reaction Control System (RCS) flow on the aerodynamic characteristics of the Space Shuttle Vehicle (SSV) orbiter. These tests were performed in the Ames Research Center (ARC) 3.5-Foot Hypersonic Wind Tunnel on a 0.015-scale model of the SSV Configuration 3L orbiter. Orbiter model 42-0 was used for this test.

Nominal freestream test conditions were Mach = 10.3 and a unit Reynolds number of 1.74 million per foot.

These test data are applicable to two points in the reentry trajectory:

M	q~psf	*R x 10 ⁶	Altitude -f
28.3	20	0.688	253,000
10.3	106.7	5 .89	168,000

Complete simulation of the RCS jet/free-stream interaction would require exact duplication of both the above conditions and the mass flow ratio, momentum, pressure, thrust and plume shape of the RCS jets. The simulation or all these conditions in a scaled-model test is not possible and, therefore, those conditions which were considered of primary significance were simulated.

The Hypersonic Mach Number Independance Principle (reference 12, which states that at very large values of M_{∞} the flow pattern and pressure coefficients on a body are independent of M_{∞}) was used as a basis for applying results obtained at M = 10.3 to the M = 28 case.

Based upon the Secondary Injection Momentum Principle for injection * Reynolds number based on orbiter length (107.5 ft.)

of a jet perpendicular to the free-stream, the dominant parameters affecting interaction forces are the jet momentum and jet pressure. Mass flow rate ratio and jet plume shape are less important parameters. Thus, the design of the model nozzles was based entirely on matching jet to free-stream pressure ratio and momentum ratio.

RCS flow was simulated by blowing jets of cold air from non-metric nozzles, attached to the model support sting, in proximity to the fuselage base. Nozzle combinations which represented pitch, roll, and yaw controls were tested in conjunction with various elevon and body flap control settings. Pitch-up and pitch-down control was simulated with jets flowing only on one side of the model on the assumption that the induced effects for two sides blowing would be twice as great.

The RCS nozzle hardware was designed, built and calibrated by the Convair Division of General Dynamics, Inc. at San Diego, California.

Nozzle thrusts were measured using a single-component strain-gauge balance. All nozzles except N₁₉ were calibrated at ambient atmospheric conditions and corrected to vacuum conditions. The N₁₉ nozzle was calibrated under near vacuum conditions because of its high expansion ratio. Mass flow rates were measured using a calibrated orifice meter. Plots of the thrust calibration data, and corresponding theoretical variations, are presented in table V.

For the in-tunnel tests, six-component force data were measured using the ARC/Task MK XIV B, 1.0-inch diameter internal balance which was supported by ARC sting No. Al3911060. Pressure taps were located within

the RCS plenum, at five points on the plenum base and at one point on the fuselage base (see figure 2d). Model RCS plenum pressure was set to obtain desired momentum ratio and pressure ratio on the basis of the thrust calibrations provided by General Dynamics.

Normal force static check calibrations; obtained prior to and immediately following each run, indicated minimal output drift but both side force gauges and the rolling moment gauge exhibited consistent, positive shifts, on the order of 1/2% of full scale output throughout the test. The character of these shifts is indicative of thermal stresses induced by heating of the model and balance during the course of each run. Axial force zero shifts were generally within 1/2% of full scale; however for runs 17, 18, 20, 22 to 25, and 27 zero shifts of magnitudes ranging from 1/2% to nearly 7% of full scale occurred. The cause of these shifts was not determined but, since the axial force data were of secondary importance to this test, the balance was not replaced. For these runs the level of axial force was adjusted so that with the air of the RCS jets off the axial force conformed with data for other runs known to be valid. This adjustment was made primarily to improve the estimates of pitching moment since pitching-moment data is influenced secondarily by the axial force when moments are transferred to the center of gravity.

Prior to each run, data were recorded with RCS jets flowing (no tunnel flow) to determine direct impingement effects. With the tunnel flowing, data were then recorded with RCS flow both fi and on at each 5° increment of angle of attack in the range from 15° to 35°. Thus, even

though the balance exhibited significant shifts the incremental data due to RCC flow with the possible exception of the axial force data, should be valid since corresponding jet off and jet on points were recorded within a relatively short time span.

Surface flow patterns of the combined tunnel and RCS flows on the surface of the model were obtained using a mixture of titanium dioxide and oil. Shadowgraph pictures were taken at selected test points.

Seven oil-flow runs and 26 valid force runs were made in the interim of July 11 to 18, 1973.

CONFIGURATIONS INVESTIGATED

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The test article, provided by Rockwell, was a 0.015-scale model (number 42-0) or the 7170-000139B SSV orbiter Configuration 3. A three-view sketch of the model, showing the principal dimensions, and photographs of the model installed in the tunnel and the RCS hardware are shown in figures 2b and 3.

The model was constructed of Armco 17-4 stainless steel and was comprised of the following parts: fuselage, canopy, wing and cuff, vertical tail and orbital maneuvering system (OMS) pods. Elevon brackets for 0°, +15°, -20° and -40°, body flaps with deflections of 0°, +13.75° and -14.25°, and a rudde: with a simulated 40° speed brake deflection were tested.

The RCS plenum was clamped to the sting at the base of the model; air loads acting on it and forces produced by the RCS jets were not measured by the balance. Five interchangeable nozzles, simulating pitch, yaw and roll controls as defined in figure 2c, were built and calibrated by the Convair division of General Dynamics.

The following nomenclature was used to designate the model comcomponents:

Component	Definition
B ₁₉	Vehicle configuration 3 (139B) fuselage of the SSV orbiter configuration (VL70-000139B)
c ₇	Basic vehicle configuration 3(139) canopy (VL70-900139)
E ₂₃	Elevon on vehicle configuration 3 (139B) wing (VL70-000139B)

F ₅	Basic vehicle configuration 3 (139) body flap (VL70-00C139)
^M 6	Modified OMS-RCS pod for the Rockwell International SSV configuration 3 (VL70-000139B)
^N 19	Twin IH yaw nozzle sized to simulate the center two prototype 3 configurations (VL70-000140A) RCS yaw engines when tunnel Mach No. equals M for prototype trajectory
^N 20	Twin LH yaw nozzle sized to simulate the center two prototype 3 configurations (VL70-000140A) RCS yaw engines at $M_{\infty} = 28.3 q_{\infty} = 20 \text{psf}$ with tunnel Mach No. equal to 10.3
N ₂₁	Twin IH pitch down nozzle sized to simulate the forward two prototype configuration 3 (VL70-000140A) aft RCS pitch down engines at M = 28.3, q = 20 psf with tunnel Mach No. equal to 10.3. (Nozzles are canted 12° aft and 20° outboard)
	are canted to are and so officerd)
_N 25	Same as N ₂₁ , except nozzles are pointed straight down
N ₂₃	
	Same as N_{21} , except nozzles are pointed straight down Twin RH pitch up nozzle sized to simulate the forward two prototype 3 configurations (VL70-000140A) aft RCS pitch up engines at M_{200} 28:3, q_{100} = 20 psf with tunnel Mach No. equal to 10.3. (Nozzle are
N 23	Same as N{21} , except nozzles are pointed straight down Twin RH pitch up nozzle sized to simulate the forward two prototype 3 configurations (VL70-000140A) aft RCS pitch up engines at M_28.3, q_ = 20 psf with tunnel Mach No. equal to 10.3. (Nozzle are pointed straight up) Complete orbiter configuration consisting of B_{10} C_7
N ₂₃	Same as N ₂₁ , except nozzles are pointed straight down Twin RH pitch up nozzle sized to simulate the forward two prototype 3 configurations (VL70-900140A) aft RCS pitch up engines at M ₂ 28:3, q = 20 psf with tunnel Mach No. equal to 10.3. (Nozzle are pointed straight up) Complete orbiter configuration consisting of B ₁₉ C ₇ F ₅ M ₆ V ₇ R ₅ W ₁₀₇ E ₂₃ Pasic vehicle configuration 3 (139) rudder for

TEST FACILITY DESCRIPTION

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The test program was conducted in air in the Ames 3.5-Foot Hypersonic Wind Tunnel. This facility is a blowdown-type tunnel that utilizes a pebble-bed heater to heat the air, and axisymmetric contoured nozzles to provide flow Mach numbers of 5.3, 7.4, and 10.4. The nozzle walls are insulated from the hot air stream by injecting helium into the nozzle boundary layer through annular slots upstream of the throat. The tunnel is equipped with a model quick-insert mechanism for quickly moving models into and out of the air stream.

A high-speed, analog-to-digital data acquisition system is used to record test data on magnetic tape. The present system is equipped to measure and record the outputs from 80 thermocouples and/or other types of transducers in addition to 20 channels of tunnel parameters.

DATA REDUCTION

Force and moments measured by the balance were resolved about body and stability axes and reduced to dimensionless coefficients by standard ARC data reduction methods. Corrections applied to the data include model static weight tare, balance and sting deflections and tunnel flow inclination. No adjustments were made to axial-force or drag coefficients for model base drag. Direct impingement force data, obtained without tunnel flow, were reduced to coefficients using the average dynamic pressure of the corresponding tunnel-on run.

Center-of-pressure location was computed in percent of body length by:

$$x_{CP}/\ell_B = (x_{CG} - \frac{c_m \epsilon}{c_N})/\ell_B$$

where

 X_{CG} = location of reference center of gravity..aft of model nose.

 I_R = body length, inches

Reference Dimensions and Constants are as follows:

Symbol	Definition	Value
A _b	Fuselage base area, OMS pods on	0.045 n ²
	Fuselage base area, OMS pods off	0.047 ft ²
A _{DM}	RCS pod area (two pods)	0.019 ft ²
ъ	Span, wing	14.050 in.
XCG	Reference C.G.	12.58 in.

DATA REDUCTION (Concluded)

Symbol	Definition	Value
z _{cc}	Reference C.G.	FRL (2=6.00)
CL BAL X	Center, balance force, measured from $X_0 = 0$, See Figure 8.	16.63 in.
CL BAL Z	Centerline, balance	W.L. 5.85 in.
ē	MAC, wing	7.122 in.
I B	Reference body length	19.35 in.
S	Area, wing (ref.)	0.605 st ²

REFERENCES

	Drawings Rockwell Drawings No.	<u>Title</u>
ı	VL70-000140A	Oribter Configuration Control.
2	VL70-000094A	Lines Control Aft Body and OMS Pod.
3	VL70-000139B	Orbiter Lines.
4	SS-A-00106	Model Assembly and Details 139 and 139B Lines SSV Orbiter.
5	SS-A-00107	Details and Assembly Wing and Vertical 0.015-Scale SSV.
6	ARC Drawing No. A13911D60	3.5-Foot Hypersonic W.T. Open Throat Model Support 1.0-Inch Balance Sting.
7	G/D Convair WT-72-108101	Tunnel Installation 0.015-Scale RCS Power Orbiter - Ames RC 3.5-Foot.
8	WT-72-108102	Assembly and Details RCS Power Orbiter Force Model 0.015-Scale.
9	Reports Report No. SD73-SH-0140	Title Pretest Information for Force Tests of the 0.015-Scale Space Shuttle Vehicle Orbiter Configuration 3A in the Ames Research Center 3.5-Foot Hypersonic Wind Tunnel (0A50), 25 May 1973.
10	G/D Convair TN-73-AE-02	Pretest Report Wind Tunnel Tests of a 0.015-Scale Space Shuttle Orbiter Model in the NASA-ARC 3.5-Foot Hypersonic Tunnels to Determine Effects of RCS Jet-Flow Field Interactions on the Aerothermodynamic Characteristics (MAS) January 1973.
11	NACA TMX-682	Holdaway, George H.; Polek, Thomas E; and Kemp, Joseph H. Jr.: Aerodynamic Characteristics of a Blunt Half-Cone Entry Configuration at Mach Numbers of 6.2, 7.4, and 10.4. NASA TM X-682, 1963.

REFERENCES (Concluded)

12 Textbook

Hayes, Wallace D.; Probstein, Ronald-F.: Hypersonic Flow Theory, Academic Press, New York, 1959.

	TEST CO	NDITIONS	
MACH NUMBER	REYNOLDS NUMBER (per unit length)	DYNAMIC PRESSURE (pounds/sq. inch)	STAGNATION TEMPERATUR (degrees Fahrenheit)
.10.29	1.72 × 106	2.43	1540
		 	
			
		-	
BALANCE UTILIZED:	ARC/Task 1.0-1	n. Die. MK XIV B	
	CAPACITY:	ACCURACY:	COEFFICIENT TOLERANCE:
NF	800 16.	1/2 %	*.020
SF SF	400 16.	± 3/4 7 ₆	± .c15
AF	100 ··· Hb.	<u>► 1/2 7.</u>	1.0025
		·	
			,
PM RM YM COMMENTS: Normal	1600-m.lb. 250 in.lb. 660 in.lb. balance sensitivity	2 1/2 % 2 1/2 % 2 1 70 u 1/2 % of full co	±.000 ±.000 ±.002 pacify; higher

TABLE IIA. OIL FLOW DATA

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1. REPENT RUN I DEE TO FLUCTUATION OF RCS PLENUM PRESURE Z. OIL PLOW N.G. DUG TO DRY OIL MIXTURE Notes:

MODEL INSERTED MTD TUNNEL STREAM DEFORE STRRTING RCS FLOW RCS FLAUING BEFARE MSERTIAN OF MODEL INTO TWINEL STREAM

RCS ON FORCE DATA TABLE IIB.

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11

DATA SET/RUN NUMBER COLLATION SUMMARY	PARAMETERS/VALUES	M Broke But & Why RUNS R PAM 6	1 350 1800 2000 176.	ı		763			EST	275	309	MBEI	75		878			275		314 10.62	43 49 55 61 67 75.76	أمل متمريل بييميرا يتيمينا بيينيا بنيدانة يتاليف
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RCS ON FORCE DATA (Concluded) TABLE 118.

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29	29	L	26				1		2		32			\dashv				_		\dashv		
29 M20	29	L	27	Nu	Ness						33			\dashv				_	37	30		
29 Mgo	29 M20	<u> </u>	28	MZI							34			\dashv				_	\exists	-	\exists	
1 7 113 19 25 31 37 43 49 55 61 67 111 111 111 111 111 111 111 111 1	1 7 13 19 25 31 37 43 49 55 OCEFFICENTS a of B A: A= 15° 20° 25′ 30° 35°	_	29	Mzo		E	F		E		35									_		
19 25 31 37 43 49 55 61 67 1 7 113 19 25 31 37 43 49 55 61 67 1 0 0 P B A: \$\times 15.20.25.30.35^{\text{cents}}\$	1 7 13 19 25 31 37 43 49 55 COEFFICENTS	Ĺ					-	<u> </u>	_											_		
13 19 25 31 37 43 49 55 61 67 A: \$\alpha = 15' 20' 25' 30' 35' \text{11} \text{12} \text{12} \text{12} \text{13} \text{14} \text{15} \text{14} \text{16} \text{19} \text{16} \text{16} \text{16} \text{16} \text{19} \text{16} \t	13 19 25 31 37 43 49 55 A: \$\alpha = 15. 20. 25. 30. 35.				T	╁	╀-	\vdash	_		\vdash	\vdash				_						
13 19 25 31 37 43 49 55 61 67 A: \$\alpha = 15. 20. 25. 30. 35.	13 19 25 31 37 43 49 55 A: $\alpha = 15^{\circ}$ 20° 25° 30° 35°				\dagger	╁	╀	+	_		\vdash								_	_		
13 19 25 31 37 43 49 55 61 67 A: \$\alpha = \frac{15}{2} \cdot 25 \cdot 30 \cdot 35 \cdot 35 \cdot \frac{15}{15} \text{TopAR (1) IDVAR (2)}}	13 19 25 31 37 43 49 55 A: $\alpha = 15^{\circ}$ 20° 25′30° 35°					╀╌	╀	\vdash	-		\vdash	\vdash										
13 19 25 31 37 43 49 55 61 67 A: \$\alpha = 15' 20' 25' 30' 35' A: \$\alpha = 15' 20' 25' 30' 35'	13 19 25 31 37 43 49 55 A: $\alpha = 15^{\circ}$ 20° 25° 30° 35°				1	╁	╀	-	-		-	\vdash										
13 19 25 31 37 43 49 55 61 67 A: \$\text{COEFFICENTS}\$ A: \$\text{CASTS 30.35}\$	13 19 25 31 37 43 49 55 A: \$\alpha = 15\cdot 20\cdot 25\cdot 30\cdot 35\cdot 40\cdot 55\cdot 25\cdot 30\cdot 35\cdot 40\cdot 55\cdot 60\cdot 60\cdot 55\cdot 60\cdot 55\cdot 60\cdot 55\cdot 60\cdot 55\cdot 60\cdot 55\cdot 60\cdot 55\cdot 60\cdot 6	1				+	╄	+	1			\vdash						_				
13 19 25 31 37 43 49 55 61 67 A: \$\alpha = 15.20.25.30.35\$	13 19 25 31 37 43 49 55 A: \$\alpha = 15^2 20^2 25^2 30^2 35^2\$	<u> </u>			-	╀╌	╀	╀	-		\vdash	T	1									
13 19 25 31 37 43 49 55 61 67 A: \$\alpha = 15.20.25.30.35\$	13 19 25 31 37 43 49 55 COEFFICENTS A: \$\alpha = 15^2 20^2 25^2 30^2 35^2	1				十	-	╀	_													
13 19 25 31 37 43 49 55 61 67 A: \$\inf\$ = 15.20.25.30.35\$	13 19 25 31 37 43 49 55 COEFFICENTS A: \$\alpha = \{5\cdot 25\cdot 30\cdot 35\cdot \}}					-	├-	-	-											\dashv	\dashv	
13 19 25 31 37 43 49 55 61 67 1	13 19 25 31 37 43 49 55 			-			Н	\vdash				H					凵	4	4	-	\dashv	٦
A: \$\infty\$ 20° 25'30'35'	A: \$ = 15, 20, 25, 30.	<u> </u>	Ĺ		61		%		31		37	7		2		SS		5		6		十
A: &= /5 20 23 30	A: \$ = 15 20 22 20	1	411	4	4	43		1;			Little SENTS	4	1	1]	4	1	40	1 44 /	-	VAR 12]
			80	4	2/2			7	1													

TABLE 11C. RCS OFF FORCE DATA

		—						TE	ST	RUN	NUM	BER	5	_		7	7		1		ř.	3	<u> </u>
	RCS	P. PSIG 6 1	6517 0																	10.62	67	444	1) 15.7 AR 12:
		12	0																		61	والمعالم	IDVAR III
	O _Z	P/EL RUNS	1 20/2		·														•		55	بعبيان	
SUMMARY		Bost Rout Tex	350 1800 2000																		••	11111	
DATA SET/RUN NUMBER COLLATION SUMMARY	PARANETERS/VALUES																·				93	ببينات	, ,
NUMBER C	P.SRAMET	W May	7 10.	8	6	//3	14	1/5	9/	1.1	18	61	20	21	22	23	24	25	. 52	27	37		COEFFICENTS
SET/RUN		50, 20	0 40			, z		132	0			13.7	-/4.2				0	0	-W.z	-/4.2	31	1	"ין
DATA	SCHO.	Ŷ	0 -20	51	-40	40-142	02-	1/2/	0			1 15	/- 02-	240	-40	-20	0	0	1-10-	1/- 02/-	25	-	25, 30
1/2/	l		o A		•	N2 N23				10	.			-	N23	•		N20			•	4444	15.20
TEST: 0473 (ARC 3.5 = 16.7)		CONFIGURATION	Orasa Nao			1/2				N20	1/31				, A			7			13	4	A: A=
: 0473	DATA SET			0.2	03	20	80	60	0/	//	. 21	67	¥	15	*	21	8/	19	20	21	,	. easter	G OR B
	8	DEN	RBSF01									2									٥	4	- 0S

TABLE IIC. RCS OFF FORCE DATA (Concluded)

See Kear P. Par	Sa K. x 9 PS 20 PS 2000 17	40 28 (R. 21) 350 (800 2000 1724 0 10.622 30 30 30 30 30 30 30	DATA SET COLL ATION SUMMARY
29 (N. 27) 350 (600 2000 172	29 (N. 27) 350 (600 2000 172	29 (N. 2) 350 (600 2000 1774 / 0 10.62) 30 33 34	Se 152F
30 31 32 33 34 35 35 37 30 31 31 32 33 34 35 35 37 30 31 31 32 33 34 35 37 37 38 39 30 30 30 30 30 30 30 30 30 30	30	30 (1/59) 32 34 (1/69) 35 35 (1/69) 37 43 49 55 (1/69) COEFFICENTS (10vaRH) (EVARH)	0 -20 -42
32 34 46 47 47 47 47 47 47 47 47 47 47 47 47 47	32	32 (1/59 / 1/59	15 13.7
33 37 49 55 61 67 10.0 R 10.0	32 1/59	32	0 0
32 34	35	33 37 43 40 55 61 67 12.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.	Naz Nz3 0 0
37 43 49 55 61 67	33	33	-40-14.2
37 43 46 55 61 67	35	35 34	N21 N23
35 43 49 55 61 EVARIED INVARELLE INDVARELLE INVARELLE IN			M21.
37 43 40 55 61 67 10.2.AR 12)	31 37 43 40 55 61 67 COEFFICENTS	37 49 SS 61 67 COEFFICENTS 10VAR 611 1CVAR 121 35.	/20
37 43 49 55 61 67	31 37 43 49 SS 61 67 COEFFICENTS COEFFICENTS 10 11 11 11 11 11 11 11 11 11 11 11 11 1	31 37 43 48 55 61 67 COEFFICENTS	
37 43 40 55 61 67	31 37 43 49 55 61 67 11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	31 37 43 49 SS 61 67 COEFFICENTS 10 VAR (1) 1 CVAR (2) 35.	
37 43 49 55 61 67	31 37 43 49 55 61 67 COEFFICENTS COEFFICENTS 10 ARR (1) 1 CVAR (2) 1 CVAR	31 37 43 49 55 61 67 COEFFICENTS	
37 43 49 55 61 67	33 49 SS 61 67 COEFFICENTS COEFFICENTS 10 49 SS 61 67 11 11 11 11 11 11 11 11 11 11 11 11 11	31 37 43 49 55 61 67 11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	
37 43 40 55 61 67	31 37 43 49 55 61 67 11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	31 37 43 49 55 61 67 COEFFICENTS 10 VAR (1) 1 CVAR (2) 35.	
37 43 49 55 61 67	31 37 43 49 55 61 67	31 37 43 49 55 61 67 11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	
37 43 49 55 61 67	31 37 43 49 55 61 67 11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.	31 37 43 49 55 61 67 1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	
37 43 49 55 61 67 11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	31 37 43 49 55 61 67 1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	31 37 43 49 55 61 67 COEFFICENTS 35.	
37 43 49 55 61 67 COEFFICENTS	31 37 43 49 SS 61 67 Littelitate Laterial Laterial COEFFICENTS 35.	31 37 43 49 55 61 67 11.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1.1	
37 43 49 55 61 67	31 37 43 49 55 61 67 Littelitation languages languages COEFFICENTS 10van (1) 15van (2) 35°	3) 37 43 49 55 61 67 Littellitiellitiellitiellitielliteri COEFFICENTS 35°	
LANGE TATABLE SATE AND SOVER CO. SEVAR (2)	IDVAR (1) ICVAR (2)	IDVAR (1) ICVAR (2)	13 19 25 3

TABLE III. - MODEL DIMENSIONAL DATA

MODEL COMPONENT: BODY - B19		
GENERAL DESCRIPTION: Fuselage, Configu	ration 3, per Rockwell	Linos
NOTE: Identical to B17 except for	orebody.	
Model Scale = .015		
DRAWING NUMBER: V170-000139	DB	
DIMENSIONS:	FULL-SCALE	MODEL SCALE
Length - IN.	1290.3	19.35450
Max. Width - IN.	267.6	4.0140
Max. Depth - IN.	244.5	3.66750
Fineness Ratio	4.82175	4.82175
Area - FT ²		
Max. Cross-Sectional	386.67	0.08700
Planform	•	-
Hetted.		
Base	•	

And the state of t

MODEL COMPONENT:	Canopy - C7	!	
		.	
GENERAL DESCRIPTION:	Configuration 3	per Rockwell Lines	VL70-000139
			·
Model Scale = .015			
DRAWING NUMBER	<u> </u>		-
DIMENSION:	•	FULL SCALE	MODEL SCALE
Length (Xo = 433 t	to X ₀ = 670) - in.	FS <u>237</u>	3.5550
Max Width			
Max Depth $(Z_0 =$	to $2_0 = 501) -$	in PS	
Fineness Ratio		· •	
Area		·	
. Max Cross-Sec	tional	<u> </u>	
Planform	•		
Wetted			•
. Base		•	•

MODEL COMPONENT: ELEVON - E23	•	
GENERAL DESCRIPTION: Configuration 3 per W	107 Rockwell Lines	
V170-0001393, data for (1) of (2) sides	·	
Model Scale = .015	· · · · · · · · · · · · · · · · · · ·	
•		
DRAWING NUMBER: VL70-000139B		•
DIMENSIONS:	FULL-SCALE	MODEL SCALE
Area - FT ²	205.52	0.04624
Span (equivalent) - IN.	353.34	5,30010
Inb'd equivalent chord	114.78	1.72170
Outb'd equivalent chord	55.00	0.8250
Ratio movable surface chord/ total surface chord	·	
At Inb'd equiv. chord	.208	
At Outb'd equiv. chord		400
Sweep Back Angles, degrees	. • •	
Leading Edge	0.00_	0.00
Tailing Edge	-10.24	-10.24
. Hingeline	0.00	0.00
Area Moment (Normal to hinge line) - FT ³ Product of Area Moment	1548.07	0.00522

MODEL COMPONENT:	5 Body Flap		
GENERAL DESCRIPTION:	3 Configuration	per Rockwell Lines	VL70-000139
Scale Model = .)15			
DRAWING NUMBER	V170-00013	9	
DIMENSION:		FULL SCALE	MODEL SCALE
length - in		84.70	1,2705
Max Width - in		267.6	4.0140
Max Depth	•		
Fineness Ratio			
Area - Ft ²		•	•
Max Cross-Section	onal _.		
Planform	•	142.5	0.03207
Wetted			
Base		38.0958	0.00857

MODEL DIMENSIONAL DATA

MODEL COMPONENT : OMS pod (M6)	·	
GENERAL DESCRIPTION Basic Configu	eration 3A OMS pode	with non-metric
RCS engine housing and nozzles. Same	geometry as M4.	
DRAWING NUMBER : VL70-000139B		
		•
DIMENSIONS :	5 111.	
DIMENSIONS:	FULL SCALE	MODEL SCALE
Length	346.0	5.1900
Max Width	108.0	1.620
Max Depth	113.0	1.695
Fineness Ratio		
. Area		
Mux. Cross-Sectional		
Planform ·		
Wetted	***************************************	·
Base		•

MODEL COMPONENT: NOZZLES - N19	•	
GENERAL DESCRIPTION: Basic configuration 3A- (VL70-000	0139B) OMS Nozzles v	ith Cold Jet
Simulation of Yaw Control (Lateral Thrust) at Mach 10.3	Entry Condition	
MODEL SCALE = 0.015		
DRAWING NO.		
DIMENSIONS	FULL SCALE	MODEL SCALE
Freestream Mach No. 10.3	•	
• • • • • • • • • • • • • • • • • • •	<i>:</i>	
		e - '
No. of nozzles (Left Side Only)		2
Expansion Ratio	-	10.81
•		
Diameter~in.	Direct Scaling	•
Excit.	not Applicable	0.1440
. Throat		0.0437
	• , ,	
Area $\sim 10^2$.		
Exit		.01629
Throat		.00151
<u> </u>		
Thrust Centerline	•	
x	1533.0	22.995
T	-	
	472.5 -	7.087

MODEL COMPONENT: NOZZLES - 1/20		
GENERAL DESCRIPTION: Basic Configuration 3A (VL70-	-000139B) ONS Nozzles	with Cold Jet
Simulation of Yaw Control (Lateral Thrust) at Mach 28	3.3 Entry Condition	
MODEL SCALE = 0.015		
DRAWING NO.		
DIMENSIONS	PULL SCALE	MODEL SCALE
Freestream Mach No. Simulation 28.3		
The common of th	•.	
No. of nozzles (Left Side Only)	2	- 2
Expansion Ratio	****	1,159
Diameter~in.	Direct Scaling	
Exit	Not Applicable	0.1440
Throat		0.1338
	*	
Area ~ In ² .		
Exit		0.1629
Throat		0.01405
Thrust Centerline	• • • •	,
X F.S.	1533.0	22.995
Y		
Z W.L.	472.5 -	7.087

MODEL COMPONENT: NOZZLES - N21		
GENERAL DESCRIPTION: Basic Configuration 3A (VL70-000	0139B) ONS Nozzles wi	th Cold Jet
Simulation of Combined Yaw/Roll Control (Thrust Canted	12 Degrees Aft and 20) Degrees
Outboard) at Fach 28.3 Entry Condition		
MODEL SCALE = 0.015		
DRAWING NO.		
DIMENSIONS	FULL SCALE	MODET SCALE
Freestream Mach No. Simulation 28.3	·	
•		•
No. of nozzles (Left Side Only)		
Expansion Ratio		1.159
•		
Diemeter~in.	Direct Scaling Not Applicable	0.1440
Exit	Not Applicable	0.1338
Throat		0.1)
Area ~ In 2.		0.01629 Ni ²
Exit		0.C14.05 E;2
Throat		
Thrust Centerline	• •	_
X F.S.	1533.0	22.995
Y B.P.	116.7	1.750
Z W.L.	472.5 -	7.087

MODEL COMPONENT: NOZZLES - N22		
TENERAL DESCRIPTION: Basic Configuration 3A (VL70-000)	(398) Nozzles with	Cold Jet
Simulation of Roll Control (Vertical Thrust) at Mach 28	3.3 Entry Condition	Left Side
Nozzles		•
MODEL SCALE = 0.015		
DRAWING NO.		•
DIMENSIONS	FULL SCALE	MODEL SCALE
Freestream Mach No. Simulation 28.3	•	
· · · · · · · · · · · · · · · · · · ·		
		•
No. of nozzles (Left Side Only)	2	
Expansion Ratio	· • • • • • • • • • • • • • • • • • • •	1.159
Diameter~in.		
Exit	Direct Scaling Not Applicable	0.1740
Throat		0.1338
•	***	
Area ~ In ² .		
Ecit	•	.01629
Throat		•01405
-		
Thrust Centerline	• • •	
X F.S.	1533.0	22.995
T B.P.	116.7	1.750
2 .		

MODEL COMPONENT: NOZZLES - N 23			
GENERAL DESCRIPTION: Basic Confi	guration 3A (VI.70-0001	39B) Nozzles with	Cold Jet Sim-
lation of Roll Control (Vertical T	Thrust) at Mach. 28.3 E	ntry Condition. Ri	ght Side
Nozzles			
MODEL SCAIE = 0.015			
DRAWING NO.	•		
DIMENSIONS		FULL SCALE	MODEL SCALE
Freestream Mach No. Simulati	ion 28.3	·	
e e e e e e e e e e e e e e e e e e e	·		
No. of nozzles (Right Side (Only) .		2
Expansion Ratio			1_159
Diameter~in.		Direct Scaling	
Exit		Not Applicable	0.1440
Throat			0.1338
		• <u> </u>	
Area ~ 1n ² .	· · .	Ì	
Exit		•	01629
Throat		<u> </u>	.011.05
Thrust Centerline	•	• • • •	
x	•	1533.0	22.995
Y	:	.116.7	1.750
z	•	-	

MODEL COMPONENT: RUDDER - R5								
GENERAL DESCRIPTION: 2A, 3 and 3A Configuration per Rockwell Lines VL70-000095								
Model Scale = .015								
DRAWING NUMBER: V170-000095								
DIMENSIONS:	FULL-SCALE	MODEL SCALE						
. Area - FT ²	106.38	0.024						
Span (equivalent) - IN.	201.0	3.015						
Inb'd equivalent chord	91,585	1:374						
Outb'd equivalent chord.	50.833	0.762						
Ratio movable surface chord/ total surface chord								
At Inb'd equiv. chord	0.400	0.400						
At Outb'd equiv. chord	0.400	0.400						
Sweep Back Angles, degrees	•	•						
Leading Edge .	34.83	34.83						
Tailing Edge	26.25	26.25						
Hingeline	34.83	34.83						
Area Moment (Normal to hinge line) - FT ³ : Product of Area and Mean Chord	526.13	0.0028						

The state of the s

MODEL COMPONENT: VERTICAL - V 7		
GENERAL DESCRIPTION: Centerline vertical ta	il, doublewedge sirfo	oil with
rounded leading edge.		
NOTE: Same as V5, but with manipulator hous	ing removed.	
Model Scale = .015		
DRAWING NUMBER: V170-000139		
DIMENSIONS:	FULL-SCALE	MODEL SCALE
TOTAL DATA		
Area (Theo) ft ² Planform	425.92	0.09583
Span (Theo) In	315.72	4,73580
Aspect Ratio Rate of Taper	1.675	1.675
Taper Ratio	0.404	0.404
Sweep Back Angles, degrees Leading Edge	45.000	45.000
Trailing Edge	26.249	26.249
0.25 Element Line Chords:	41.130	41.130
Root (Theo) WP	268.50	4.02750
Tip (Theo) WP	108.47	1.62705.
MAC Fus. Sta. of .25 MAC	199.81 1463.50	2.99 <u>15</u> 21.9525
W. P. of .25 MAC	635.522	9.53283
B. L. of .25 MAC Airfoil Section	0.00	0.00
Leading Wedge Angle Deg	10,000	10.000
Trailing Wedge Angle Deg	14.920	14.920
Leading Edge Radius Void Area - +2	2.0	0.0300
Blanketed Arca	13.17	0.00296

MODEL COMPONENT: WING-WOOD GENERAL DESCRIPTION: Configuration 3 per-Rochwell Lines VI/70-0001398 Same as W103, except cuff, airfoil and incidence angle. Model Scale = .015 TEST NO. DWG. NO. VI.70-0001398 DIMENSIONS: FULL-SCALE MODEL SCALE TOTAL DATA Ft² Area (Theo.) Planform 0.60525 Span (Theo In. 936.68 14.050520 Aspect Ratio 2.265 2,265 1.177 Rate of Taper 1,177 Taper Ratio 0.200 0.200 Dihedral Angle, degrees (@ TE of Elevon) 3,500 3,500 Incidence Angle, degrees 0.500 0.500 Aerodynamic Twist, degrees +3.000 +3.CCO Sweep Back Angles, degrees 45.000 Leading Edge 45.000 Trailing Edge -10.24 0.25 Element Line 35.209 Chords: Root (Theo) B.P.O.C. Tip. (Theo) B.P. 10.33860 137.85 2.06775 MAC 474.8). 7.12215 Fus. Sta. of .25 MAC W.P. of .25 MAC B.L. of .25 MAC 1136.89 17.05335 299,20 4.4880 162.13 2.73195 EXPOSED DATA Ft² 1752.29 Area (Theo) Span, (Theo) 0.39426 In. 8P108 720.68 10.81020 2.058 2.053 Aspect Ratio 0.2451 Taper Ratio 0.2451 Chords Root BP108 562.40 8.4360 Tip 1.00 b 137.85 2.06775 393.03 185.31 300.20 17-77965 Fus. Sta. of .25 MAC W.P._of .25 MAC B.L. of .25 MAC Afrfoil Section (Rockwell Mod NASA) 4.5030 XXXX-64 0.10 0.10 Root b 0.12 0.12 Tip b = Data for (1) of (2) Sides Leading Edge Cuff Planform Area Ft2 18.333 0.02662 Leading Edge Intersects Fus M. L. . Sta 7.5000 Leading Edge Intersects Wing @ Sta 1083.4 16,2510

Table IV. RCS Direct Impingement Force Data

RJN NQ.	A C _M	A CA	A 0 F	₽C.	ΔCA	¹ 28
7	0.0	0.0	0017	٥٠٠	.0002	1000
6	0.0	.0003	.0003	0.0	0.0	.0001
15	0110	0042	0500	.0106	.0025	0019
16	0120	0030	0047	.0116	8100.	0022
17	0023	0.0	0016	0002	.0002	0.0
19	0121	0026	0012	.0128	.0011	0012
20	0148	0004	*000	+ .0130	+ .0004	0012
21	0143	0017	0010	+ .0133	+ .0003	0012
22	+ .0008	+ .0001	+ .0021	0005	9000. +	♦ 000 -
23	+ .0010	+ .0002	0017	0004	6000. +	0005
24	+ .0014	0.0	0031	0007	+ .0010	0005
25	+ .0001	+ .0001	+ .0004	0001	+ .0001	0.0
26	1000	0003	0.0	0.0	+ .0002	4
27	0.0	0001	+ .0007	4	0.0	
28	+ .0001	+ .0001	+ .0003	→	+ .0001	
29	+ .0001	0.0	90000 -	0.0	0001	→
30	+ .0002	+ .0001	0.0	+ .0001	0.0	0.0
31	0005	0001	0039	+ .0244	+ .0023	0031
32	+ .0005	0002	0033	+ .0211	+ .0019	0022
33	0118	0020	0051	+ .0133	+ .0024	0022
34	0 .0164	0026	0019	+ .0156	9000° +	0014
35	+ .0003	+ .0003	0010	0001	+ .0002	0001

KE 10 X 10 TO THE CENTIMETER 46 1817
LO X 20 CH - ALBANENEO
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TABLE V. - NOZZIE CALIBRATION CURVES

Theigh Earth Po - FRIA 38

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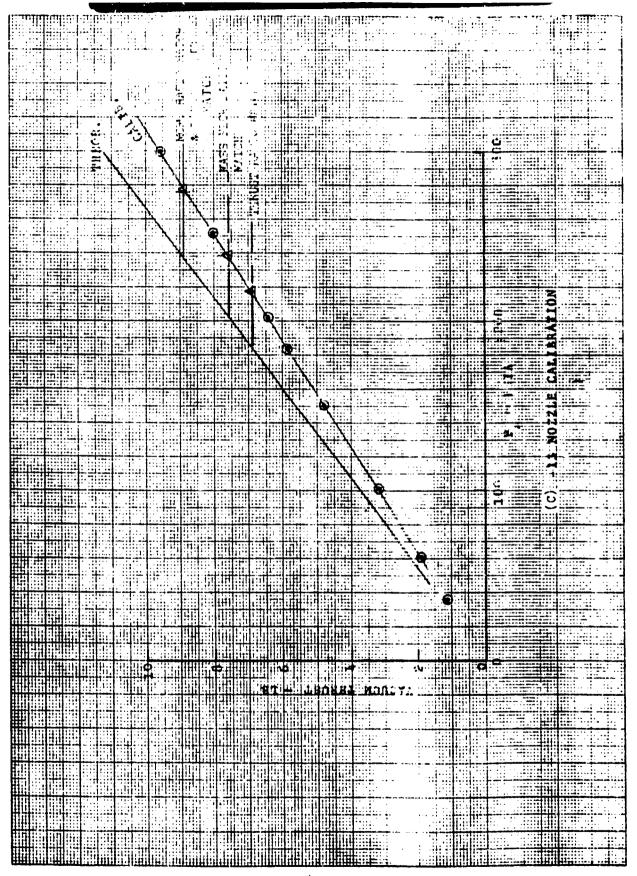
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Continued

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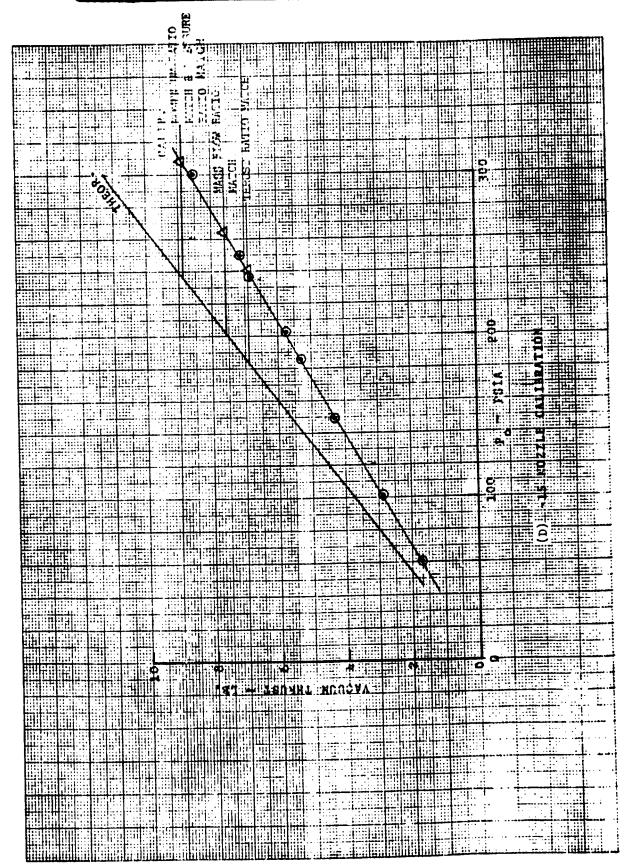
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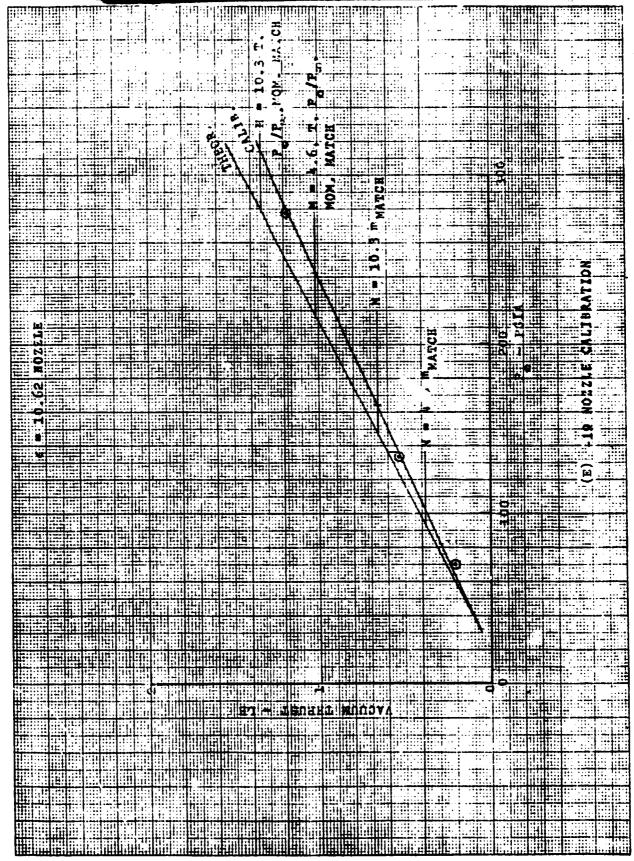
TABLE V. - Continued.

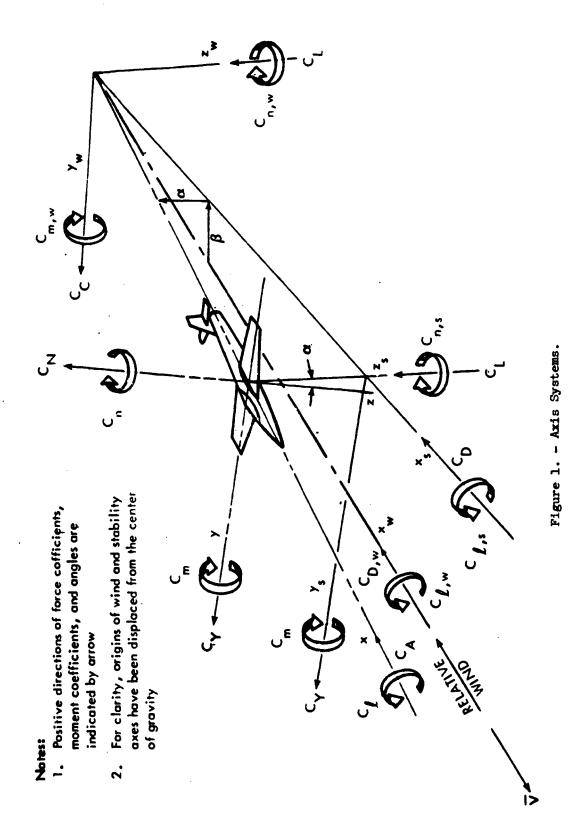


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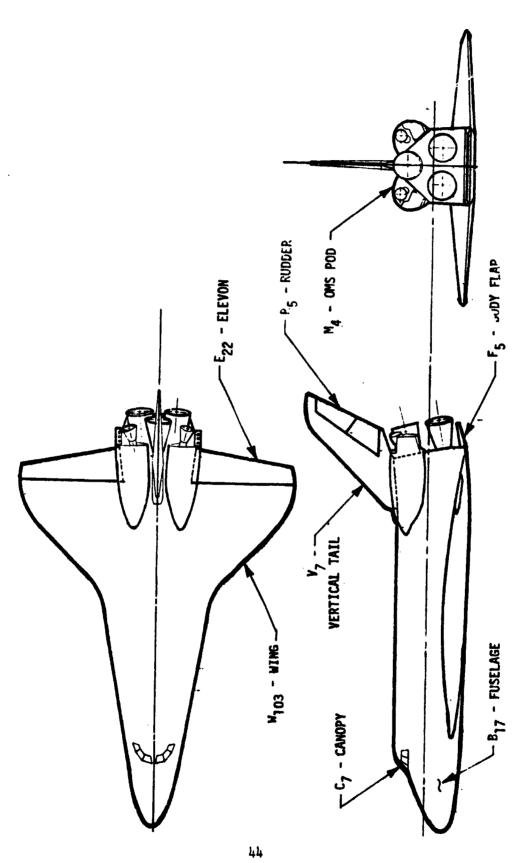
TABLE V. - Concluded.





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(a) SSV Orbiter VL70-000139B Model Nomenclature

Figure 2. - Model sketches.

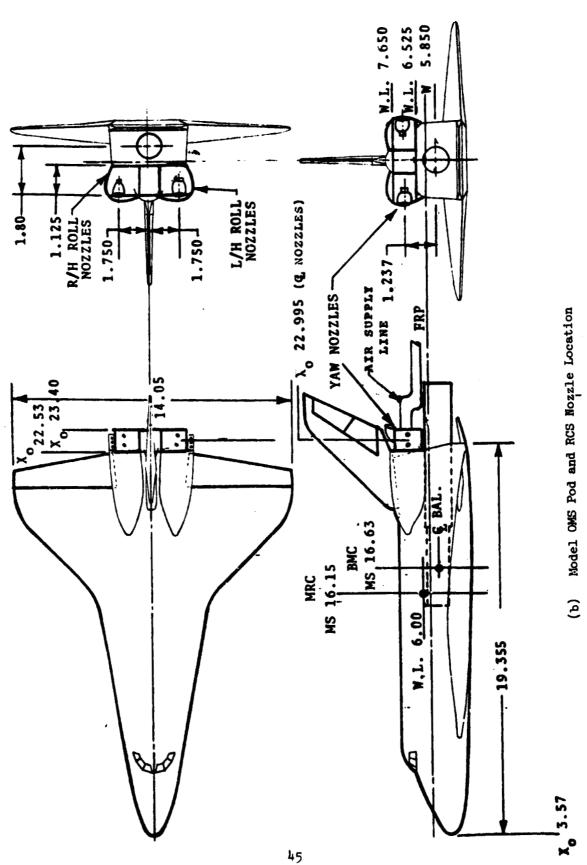


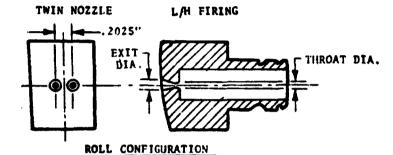
Figure 2. - Continued.

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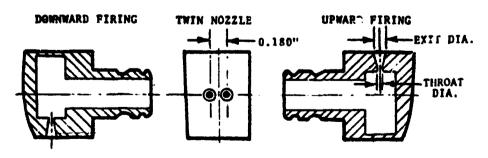
YAW CONFIGURATION

		THROAT.		EXIT			
DASH NO.	NO. OF	DIA. (IN.)	(IN. ²)	· DIA. (IN.)	AREA (IN. ²)	EXPANSION RATIO	NOTES
-19	2	0.0437	0.00151	0.1440	0.01629	10.81	L/H FIRING NOZZLES SIMULATING Mac = 10.3 FLIGHT CONDITIONS
- 20	2	0.1338	0.01405	0.1440	0.01629	1.159	SIMULATING Me = 28.3 FLIGHT CONDITIONS



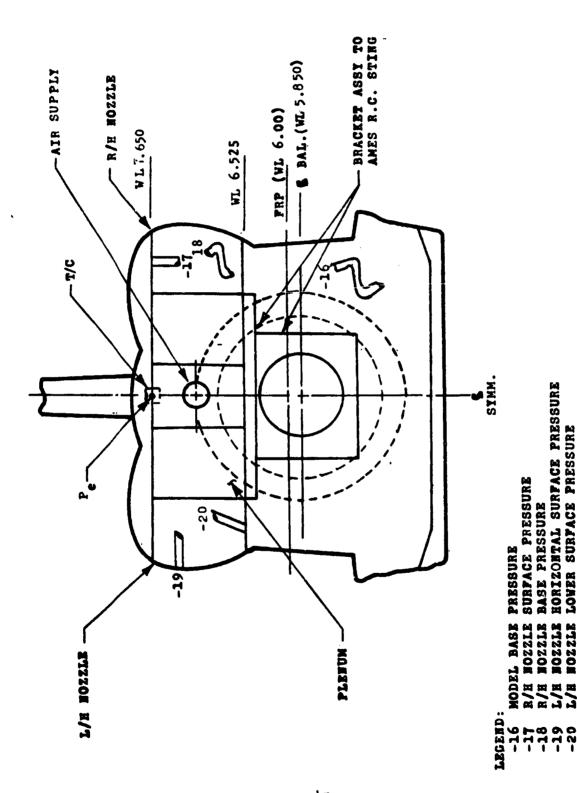
DASH NO. OF	THEOAT		AIT		EXPANSION		
NO.	NOZZLES	DIA. (IN.)	AREA	AIG (15.)	AREA (IN. 2)	RATIO	NOTES
-21	2	0.1338	0.01405	0 1440	0.01629	1.159	DOWNWARD FIRING L/H NOZZLE CANTED 12° AFT & 20° OUTBOARD*
-22	2	0.1338	0.01405	0.1440	0.41629	1.159	DONNWARD FIRING L/H NOZZLES POINTED STRAIGHT DOWN*
- 23	2	0.1338	0.01405	0 1440	0.01629	1.159	UPWARD FIRING E/H NOZZLES POINTED STRAIGHT UP

*SIMULATES
M = 28.3 FLIGHT
CONDITIONS



(c) Details of RCS Nozzle Geometry

Figure 2. - Continued.



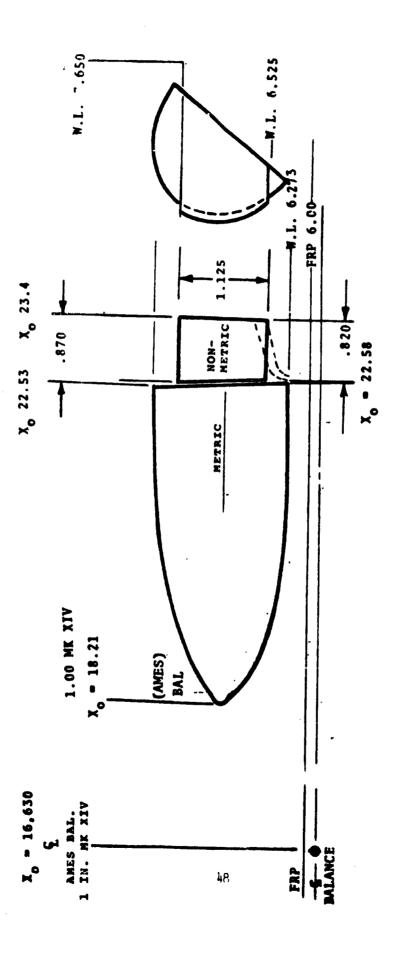
Arrangement of Fuselage and RCS Plenum Base Pressures (q)

HORIZONTAL SURFACE PRESSURE LOWER SURFACE PRESSURE

BASE PRESSURE

HOZZIE HOZZER HOZZTE Figure 2. - Continued.

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(e) M₆ OMS Pod

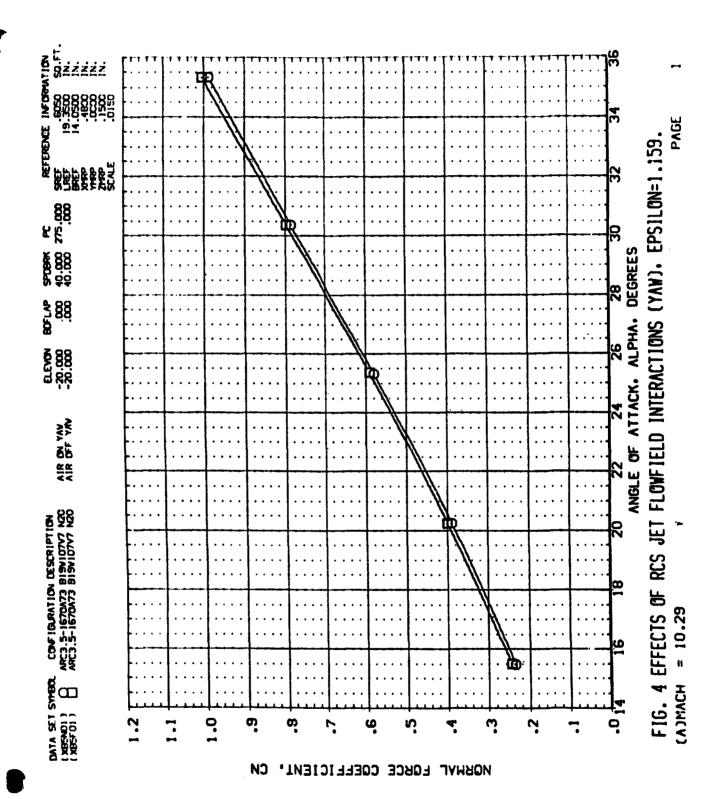
Figure 2. - Concluded.



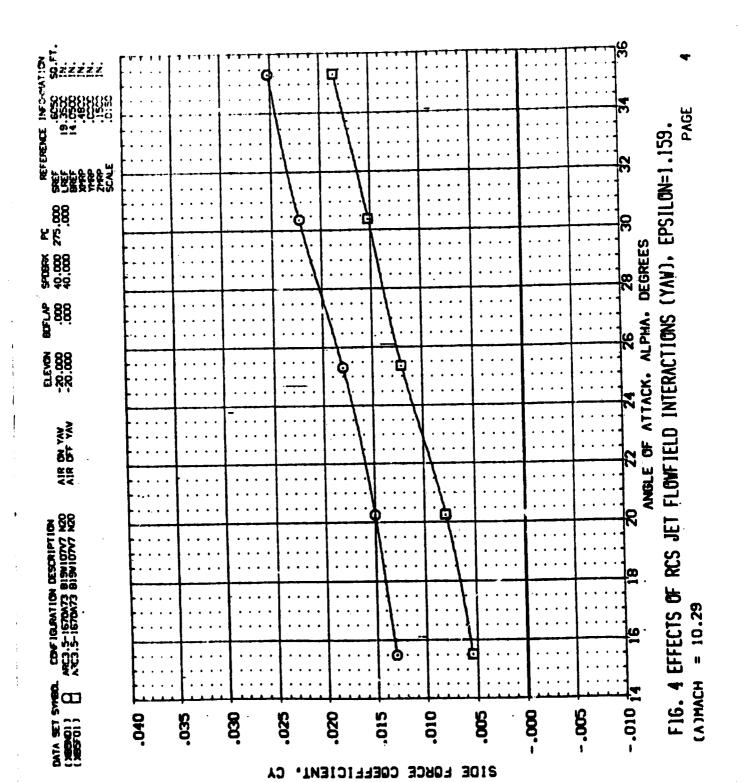
(a) 0.015-scale orbiter model equipped with non-metric OMS pods in the Ames 3.5-Foot Hypersonic Wing Tunnel Figure 3. - Installation photographs.

(b) RCS nozzies and related hardware Figure 3. - Concluded.

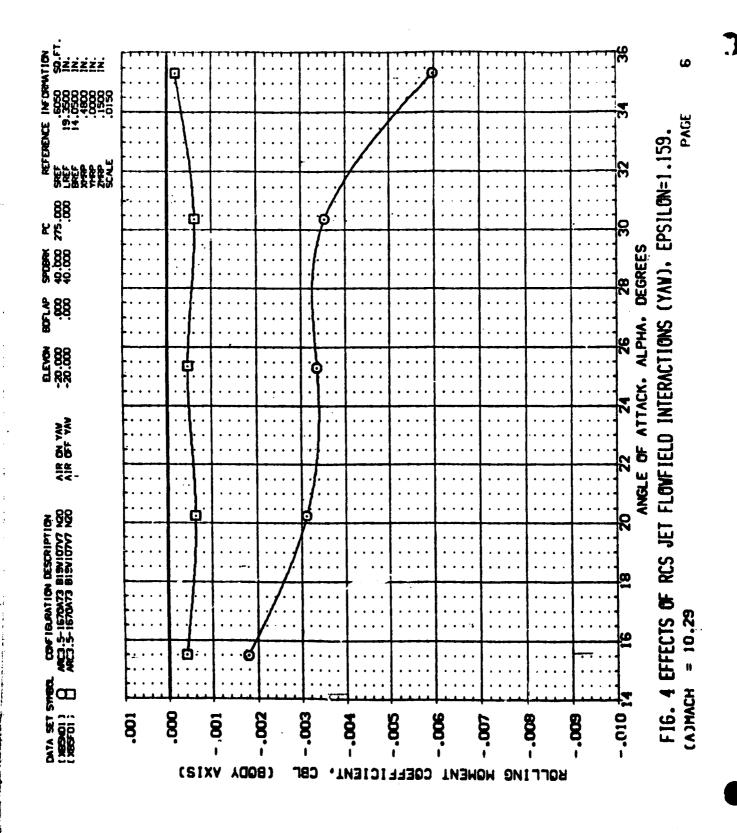
DATA FIGURES



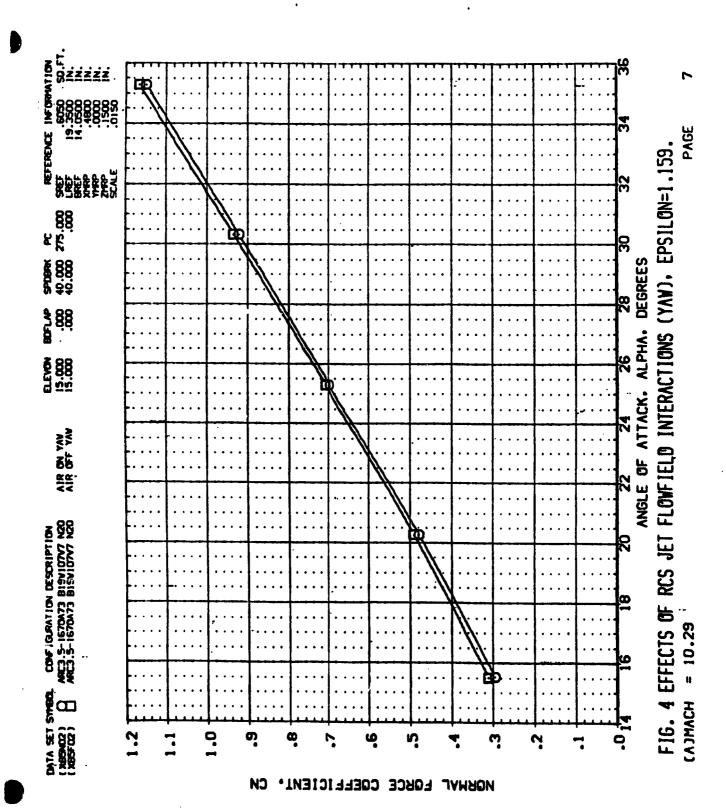
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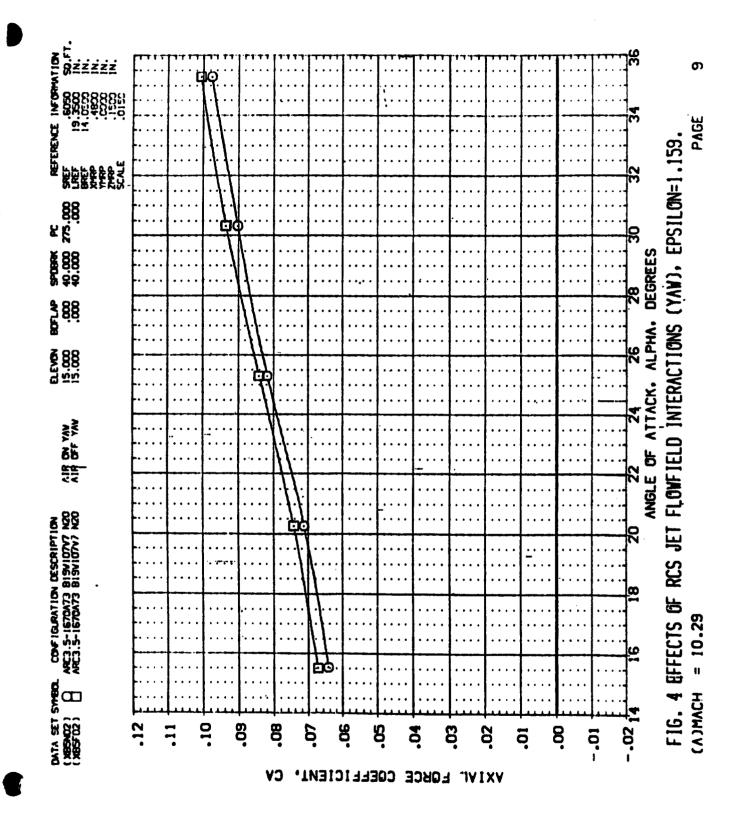


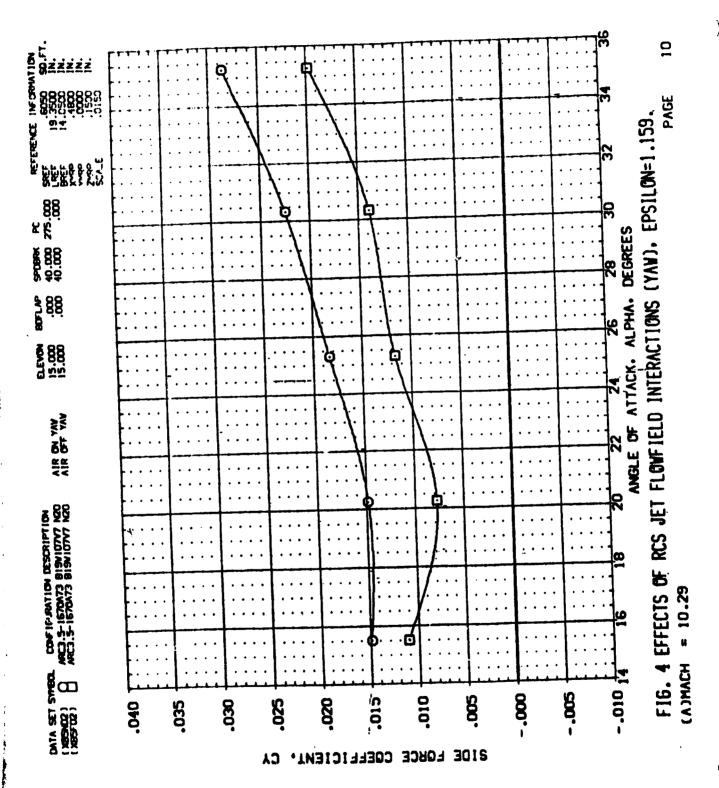
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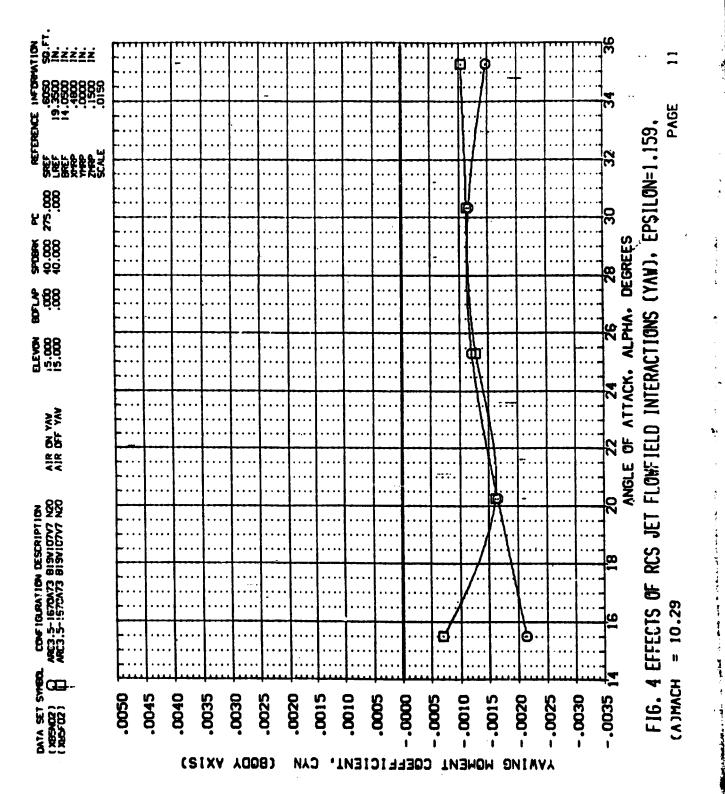
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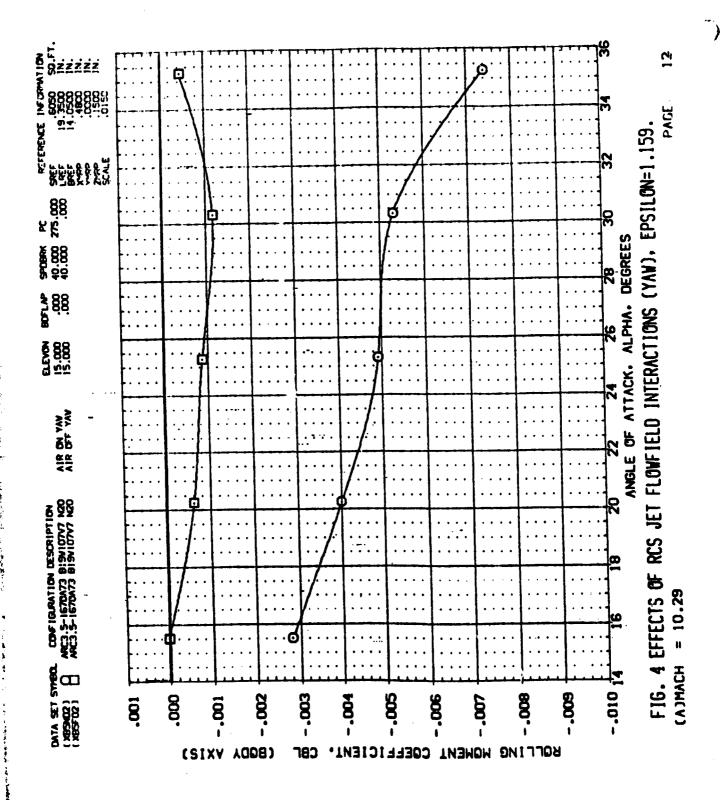
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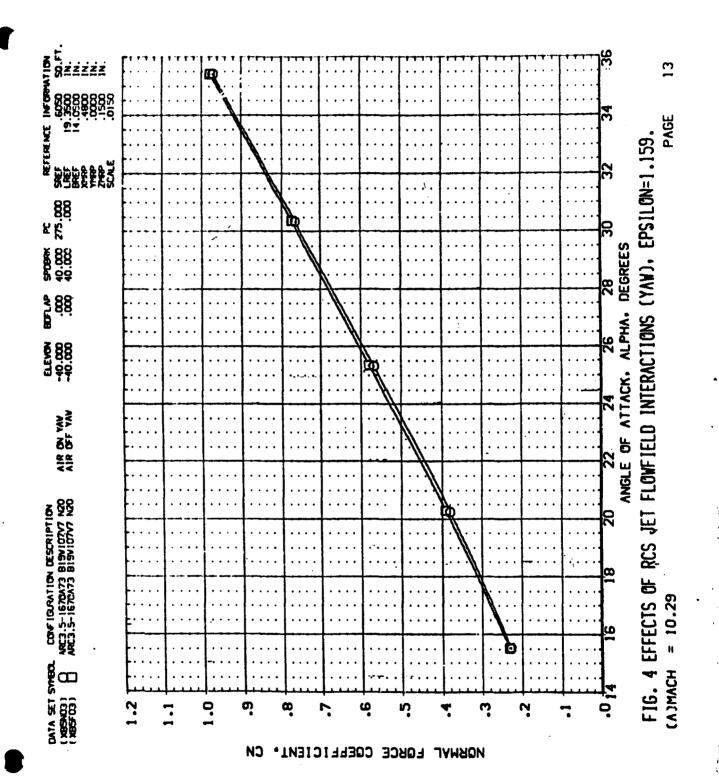




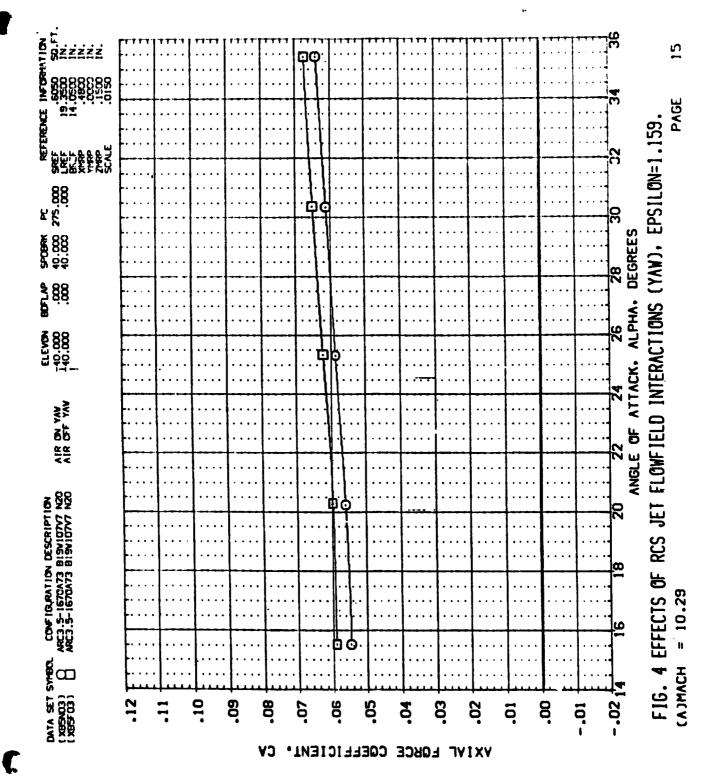
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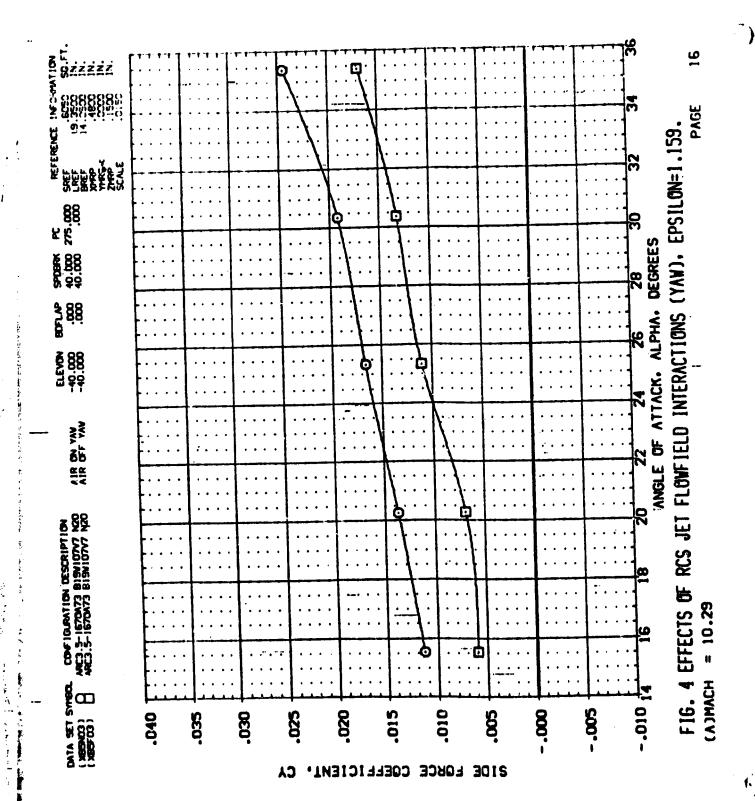


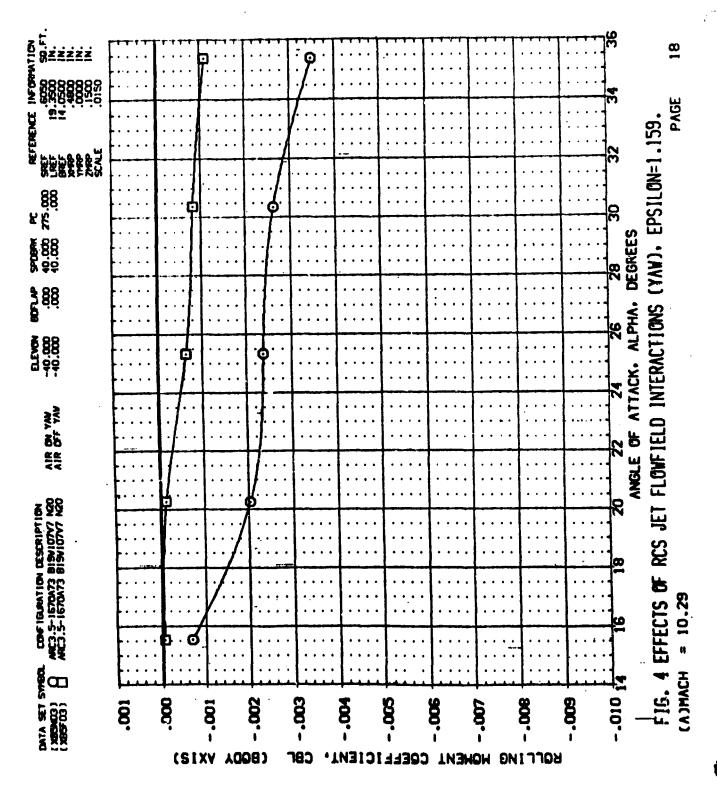


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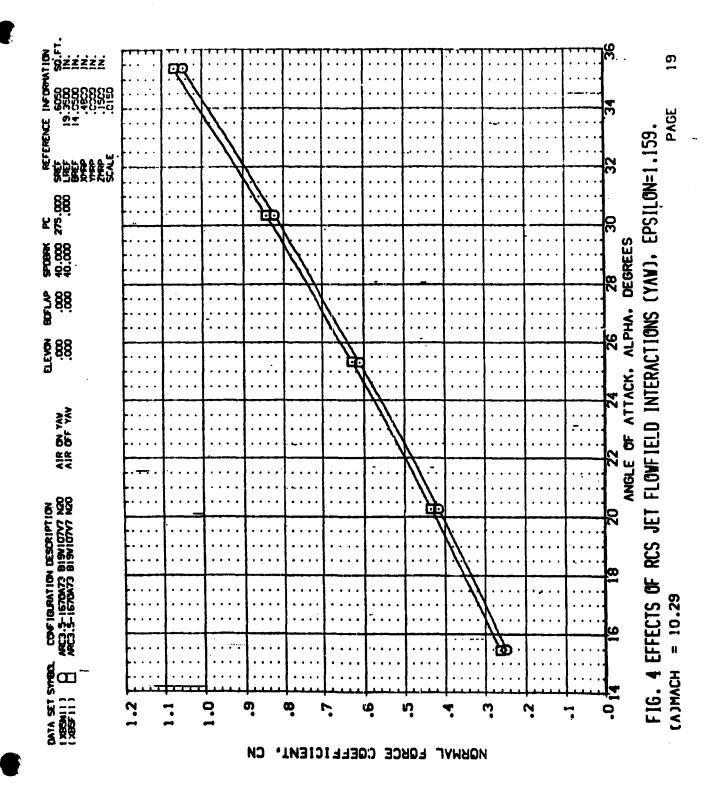


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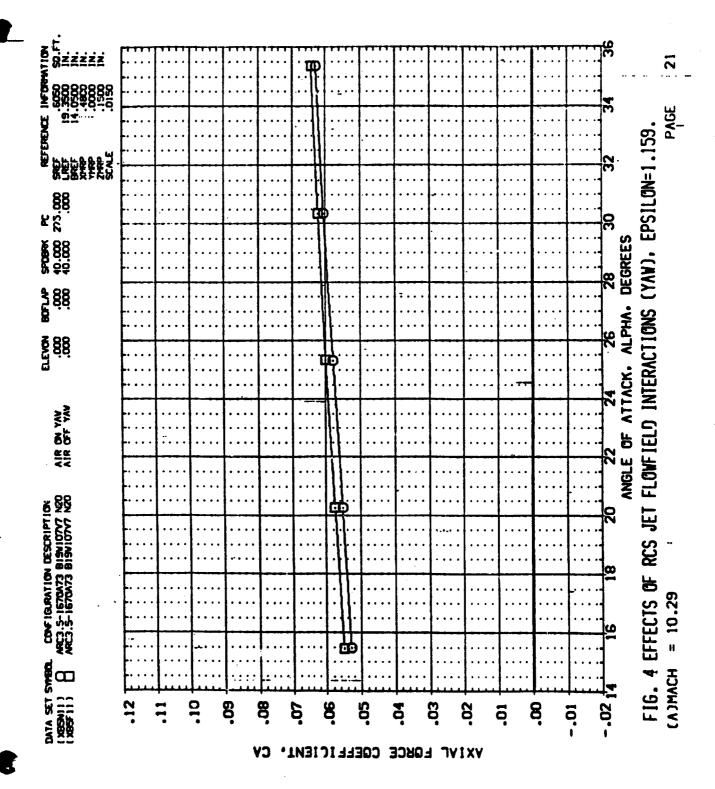




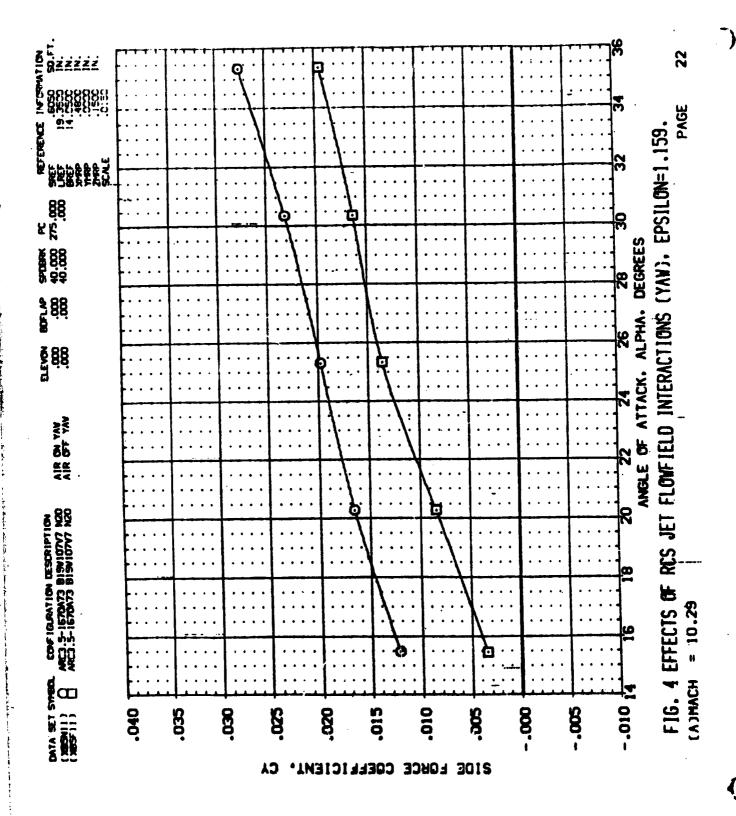
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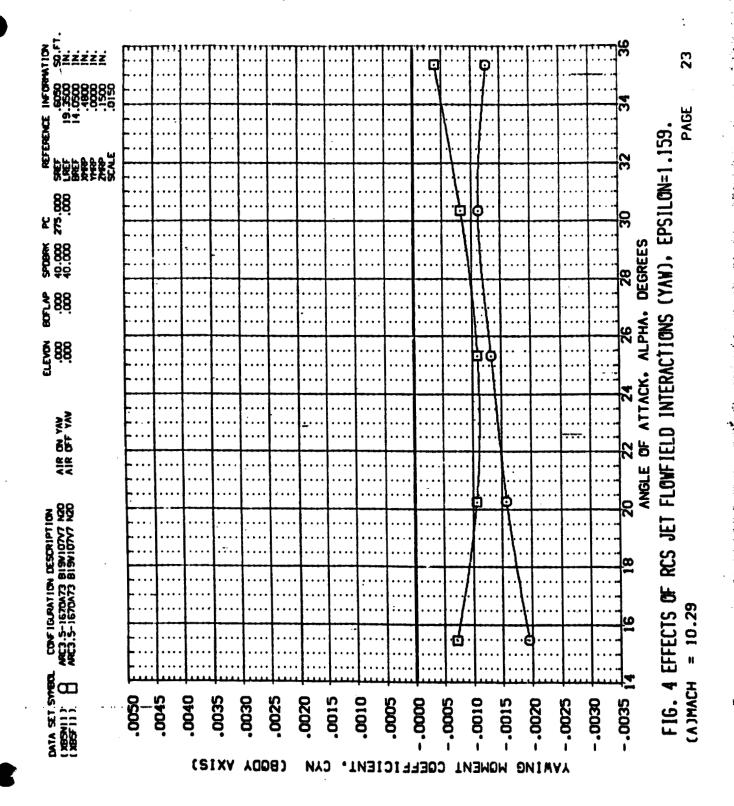


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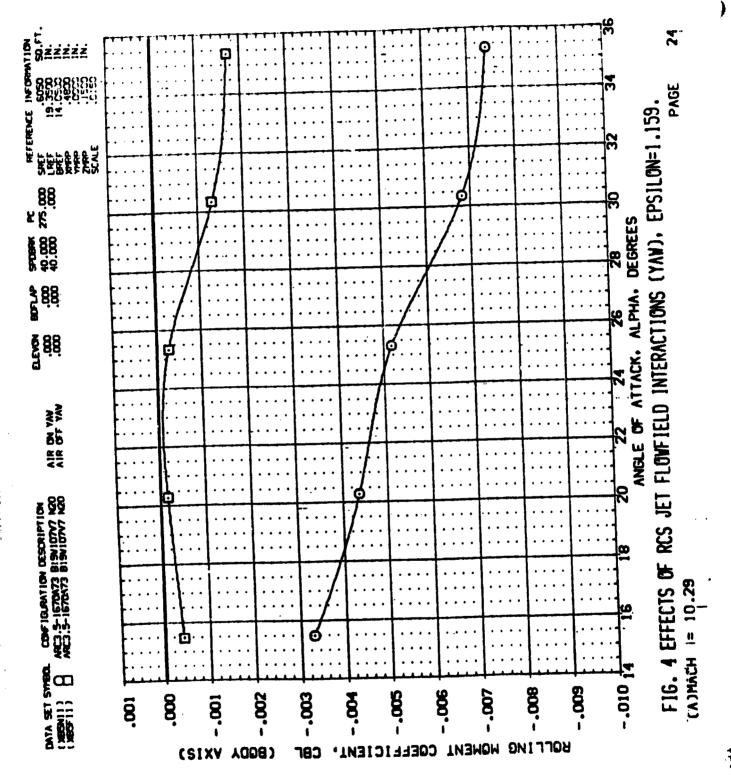


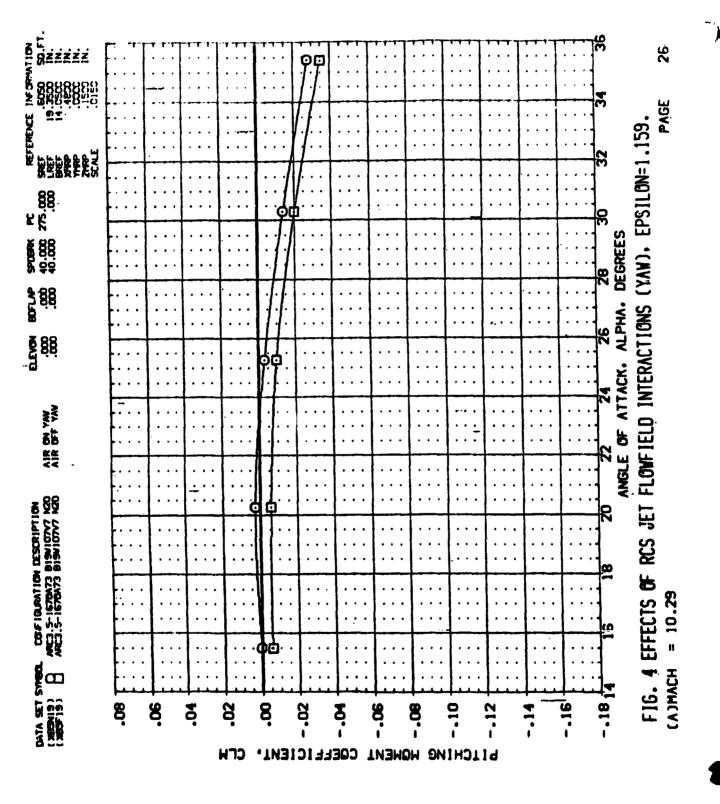
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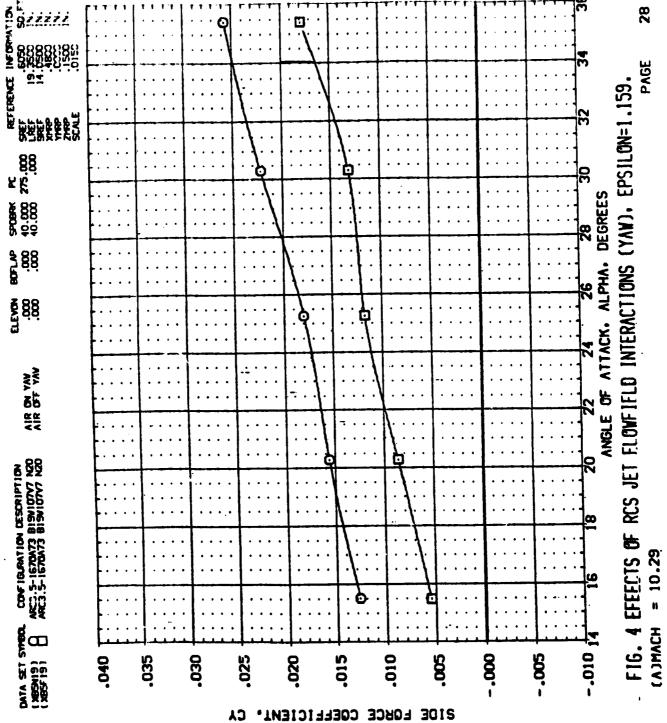


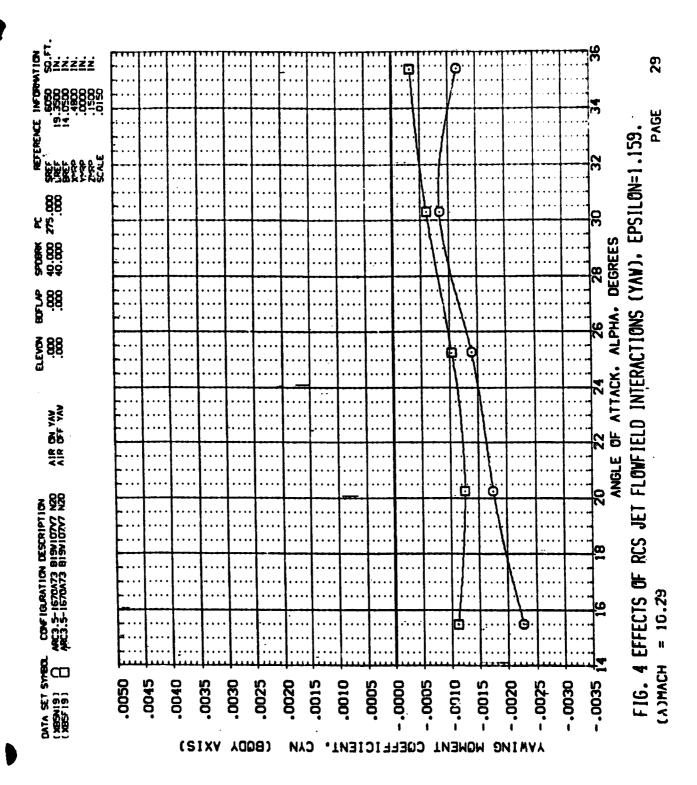


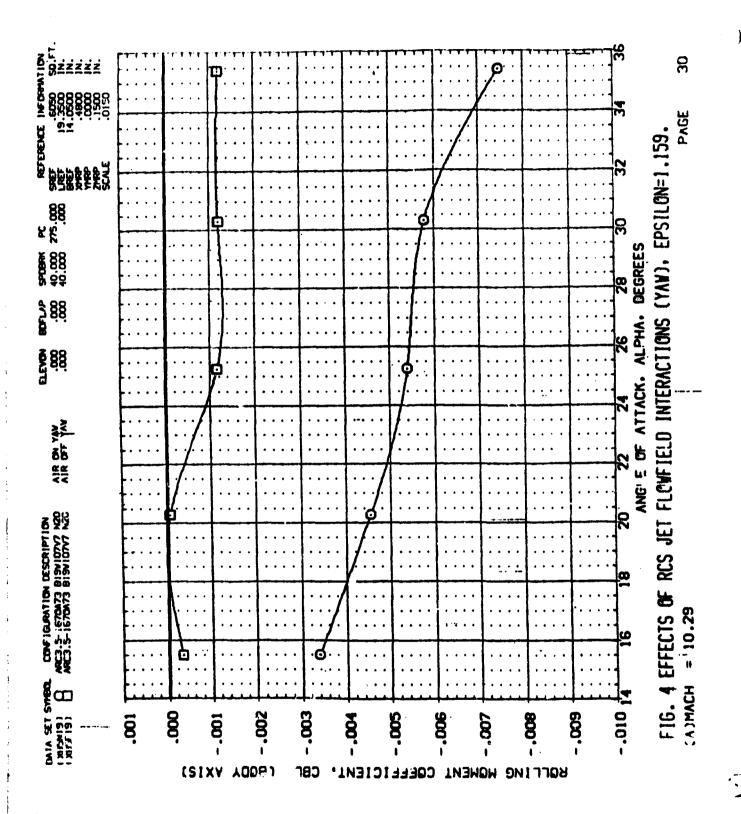
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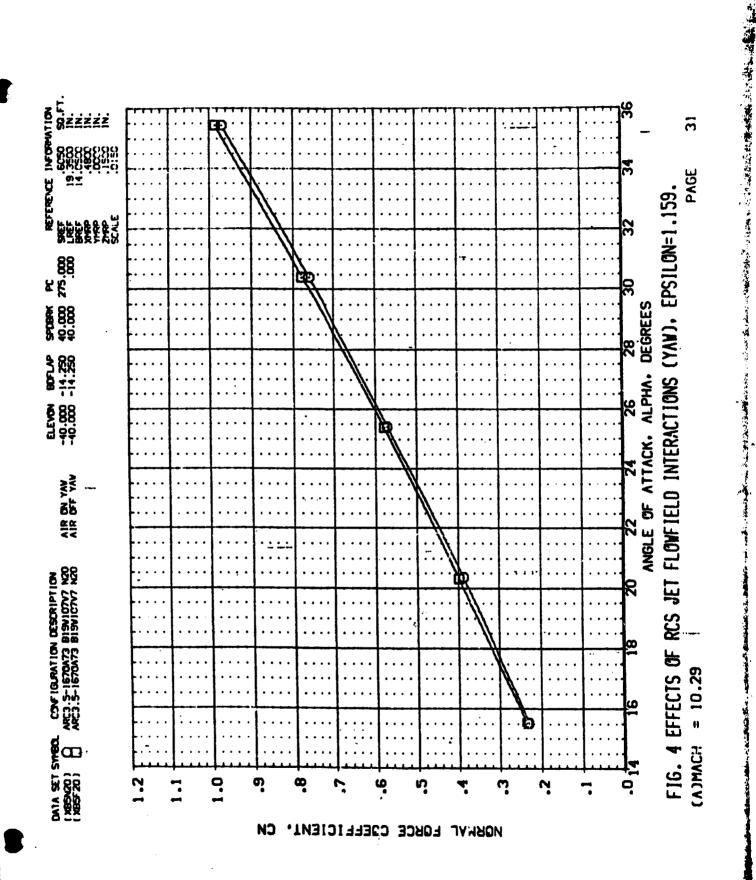




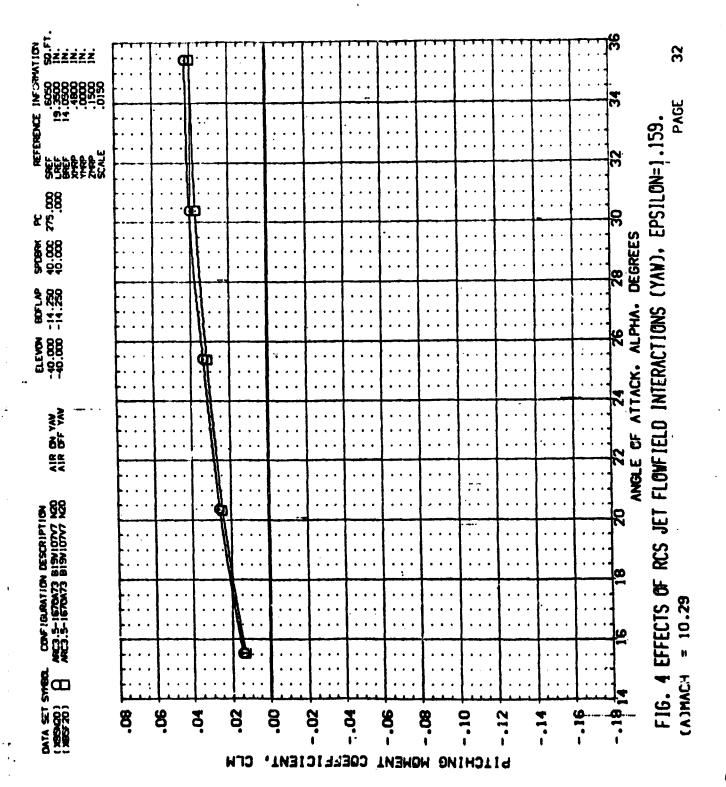




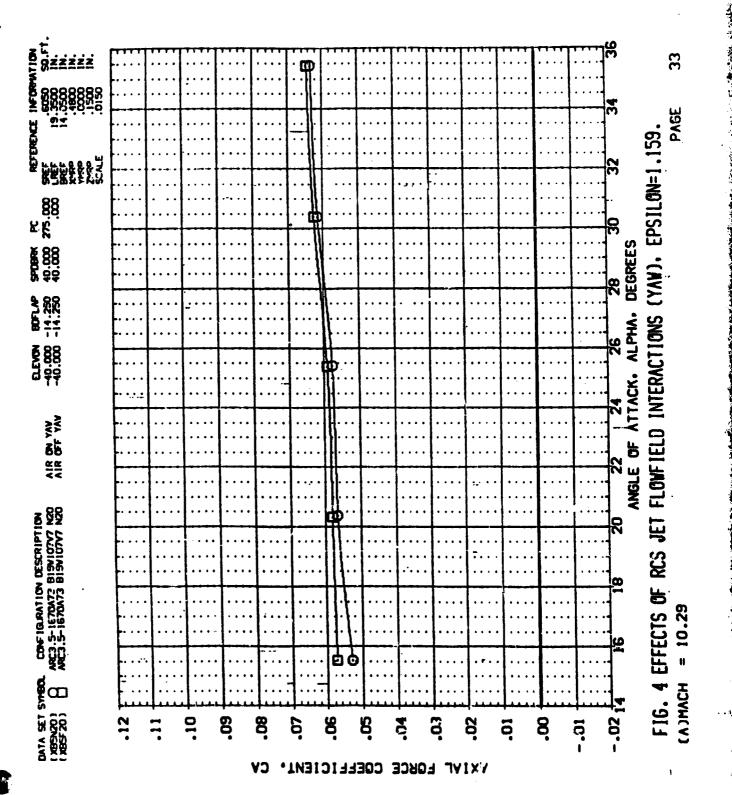


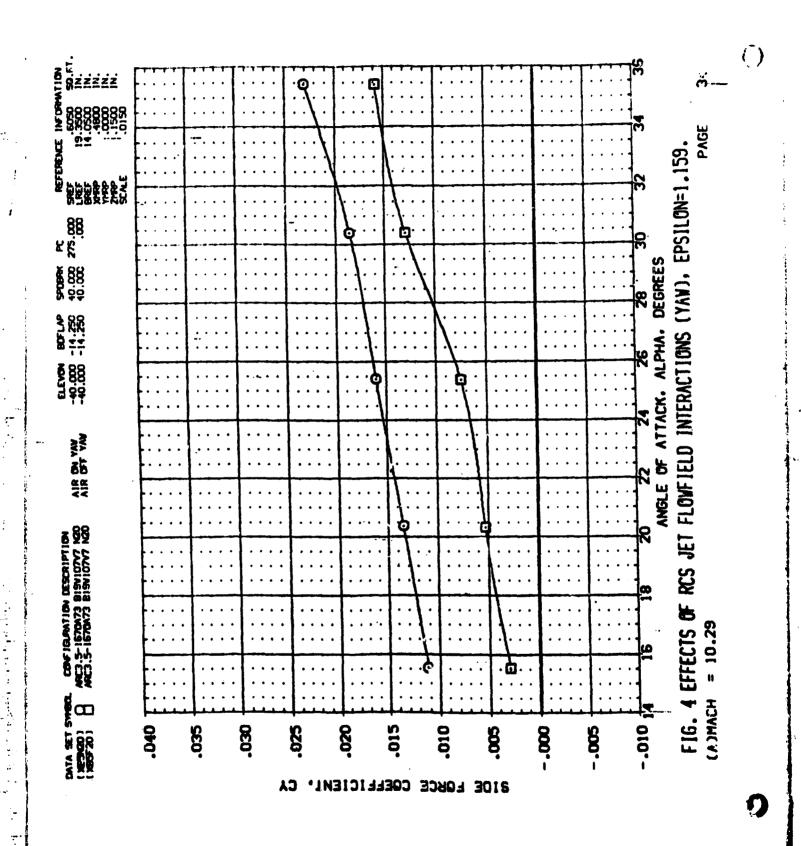


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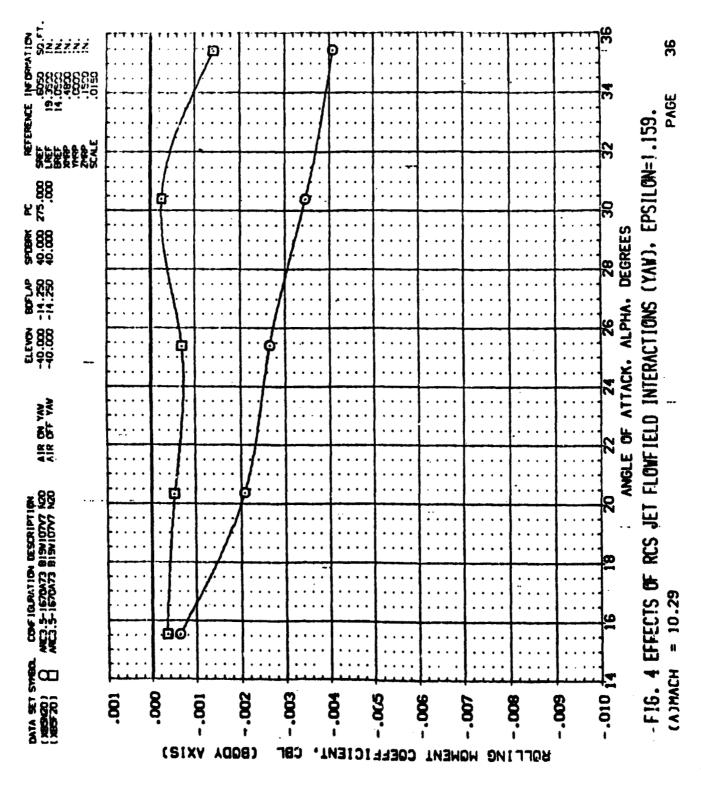


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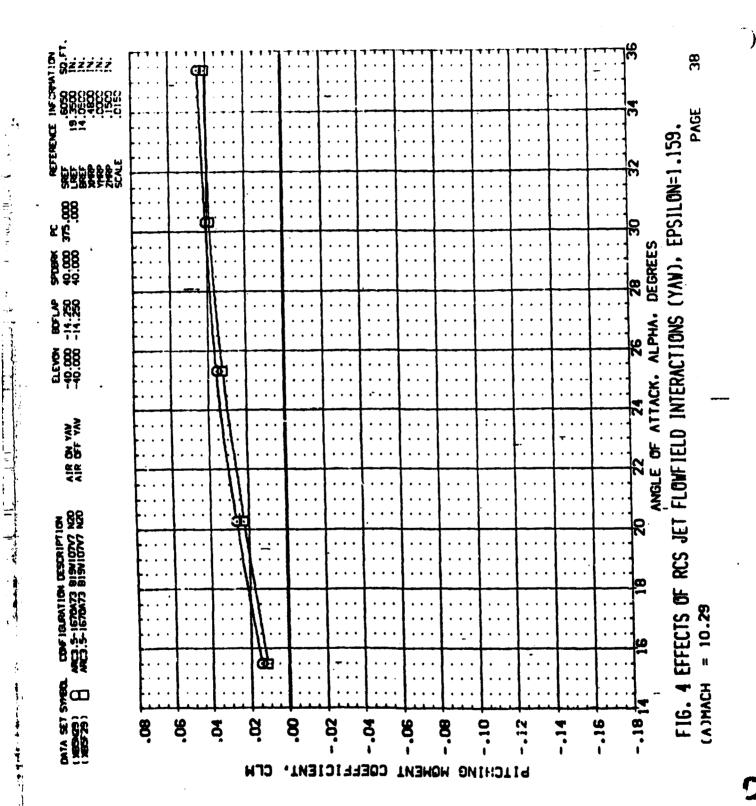
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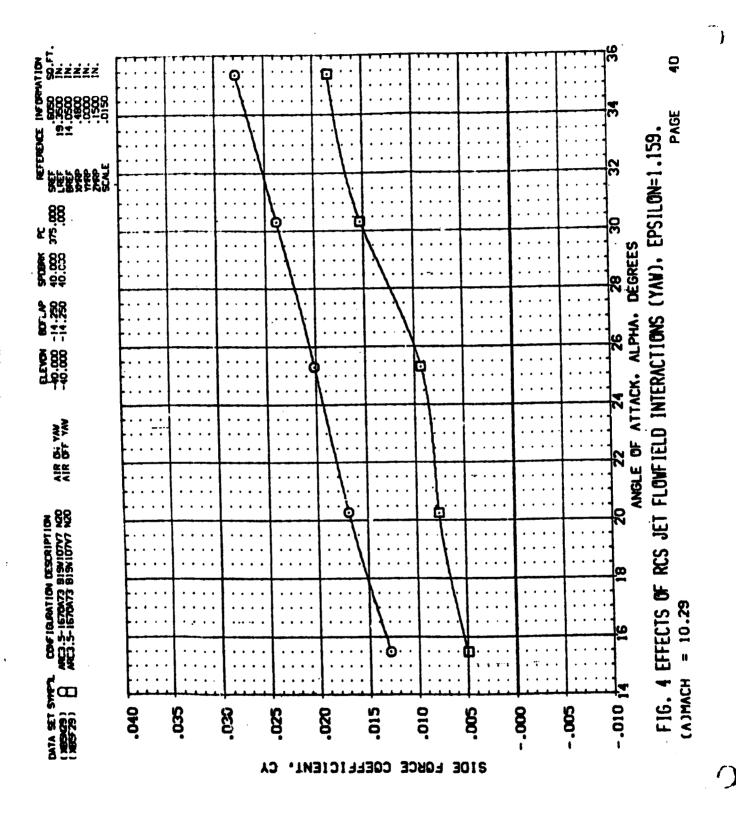


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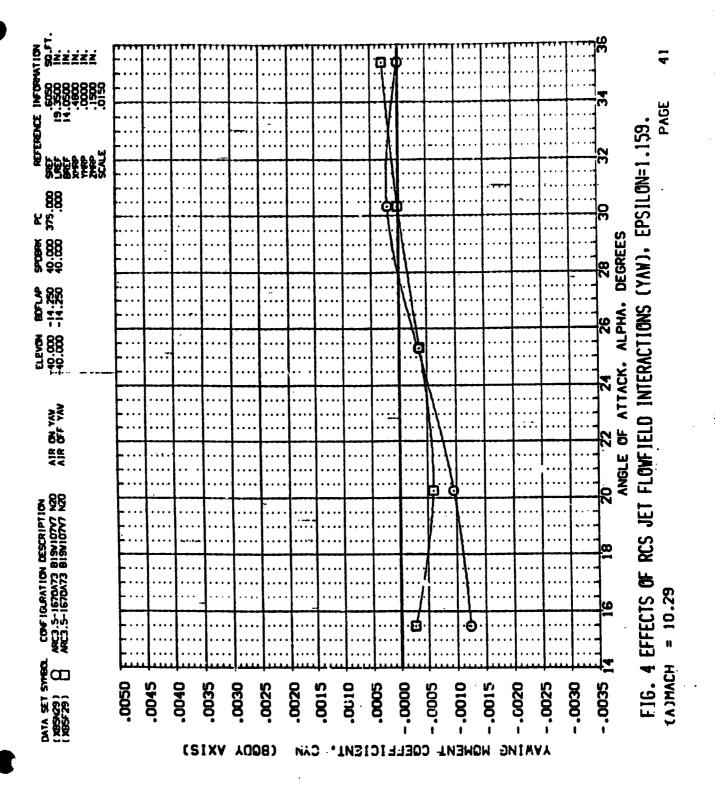
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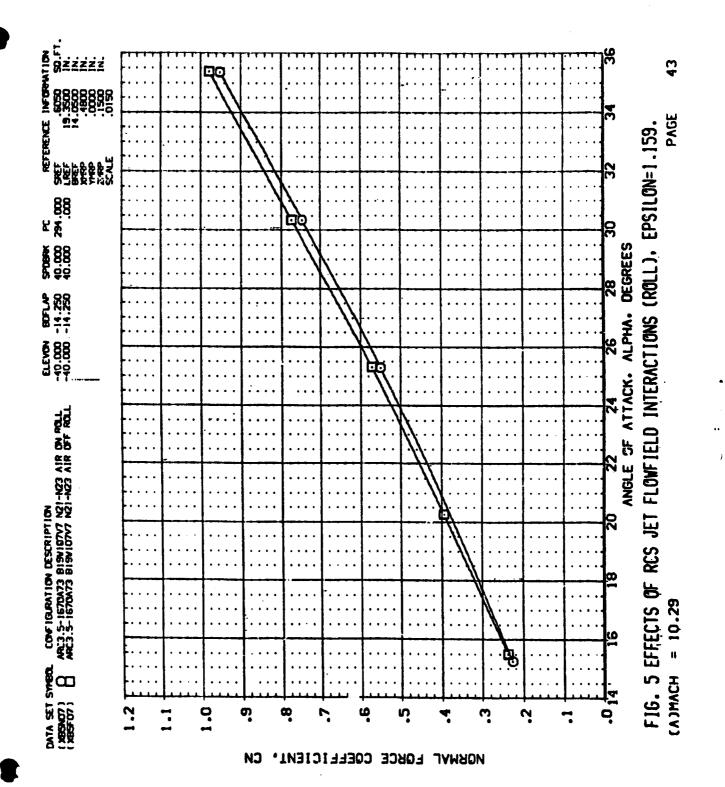
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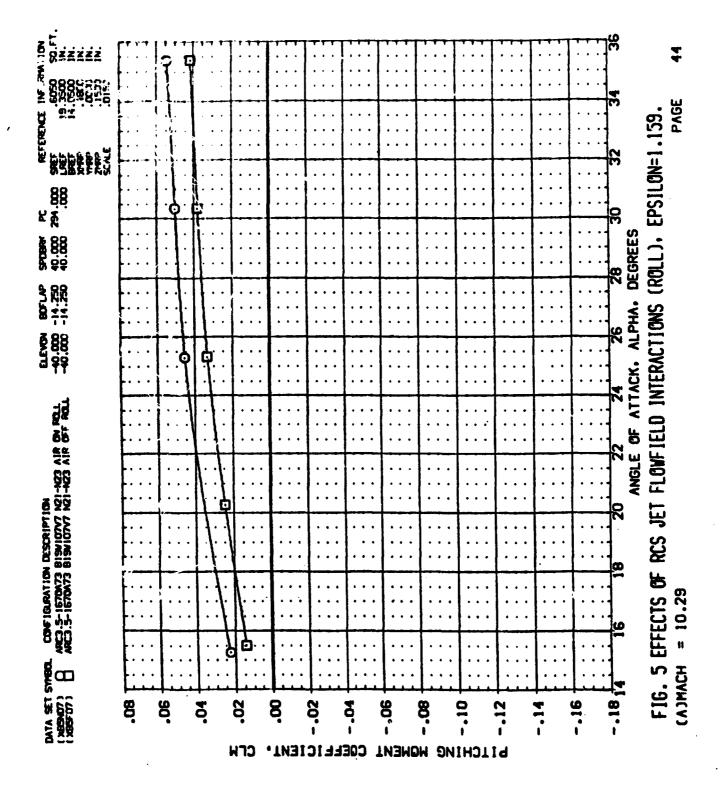


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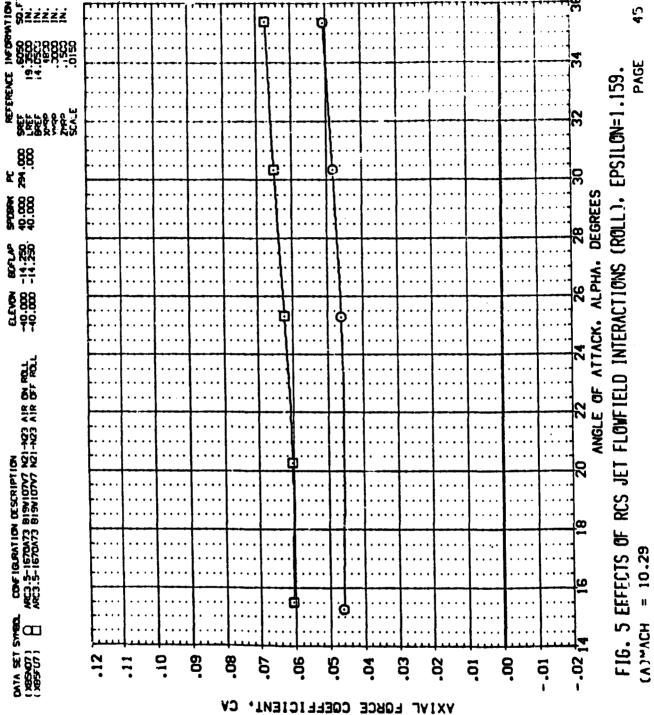
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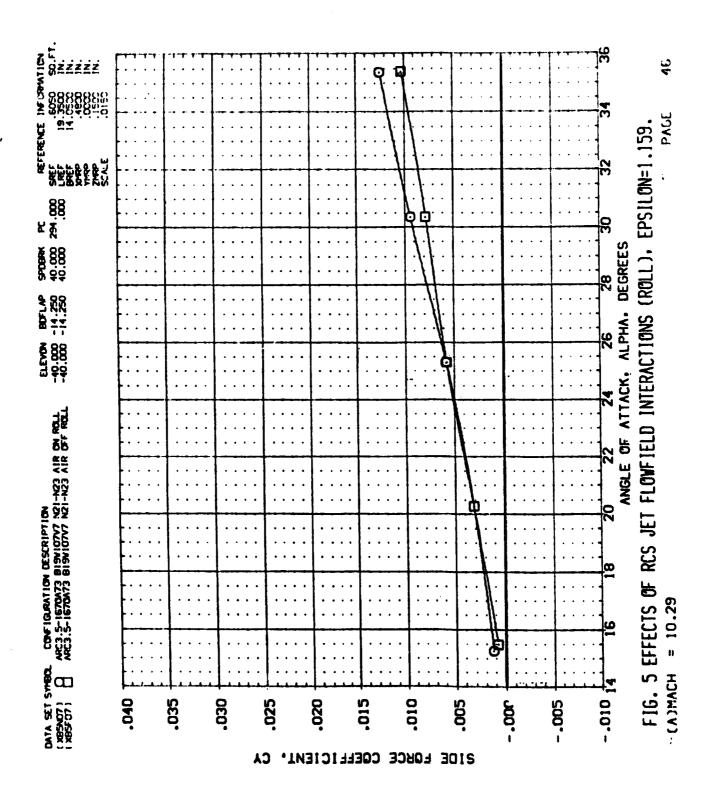


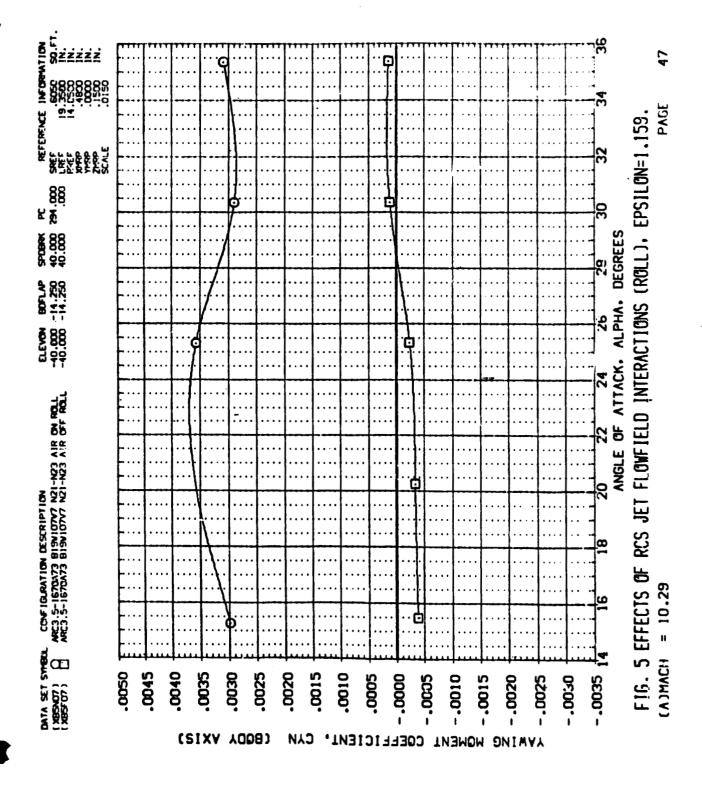


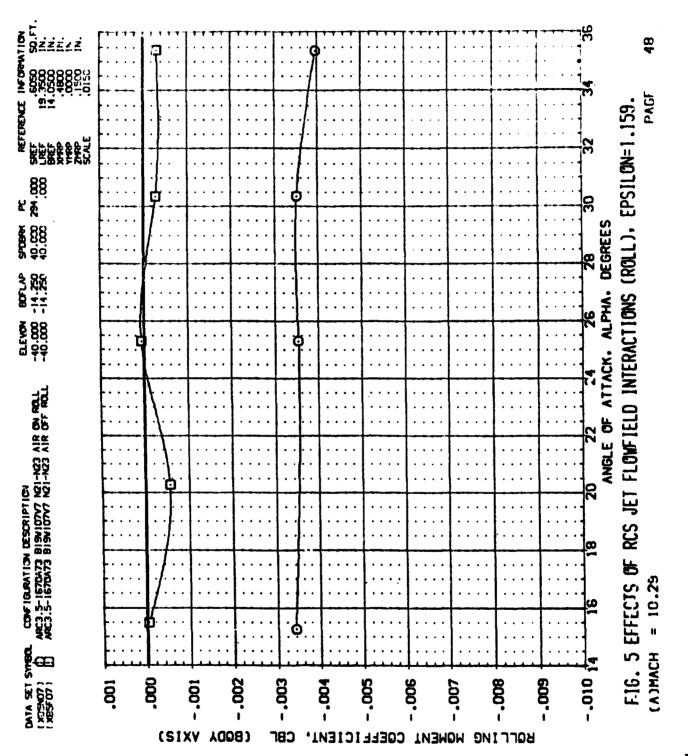
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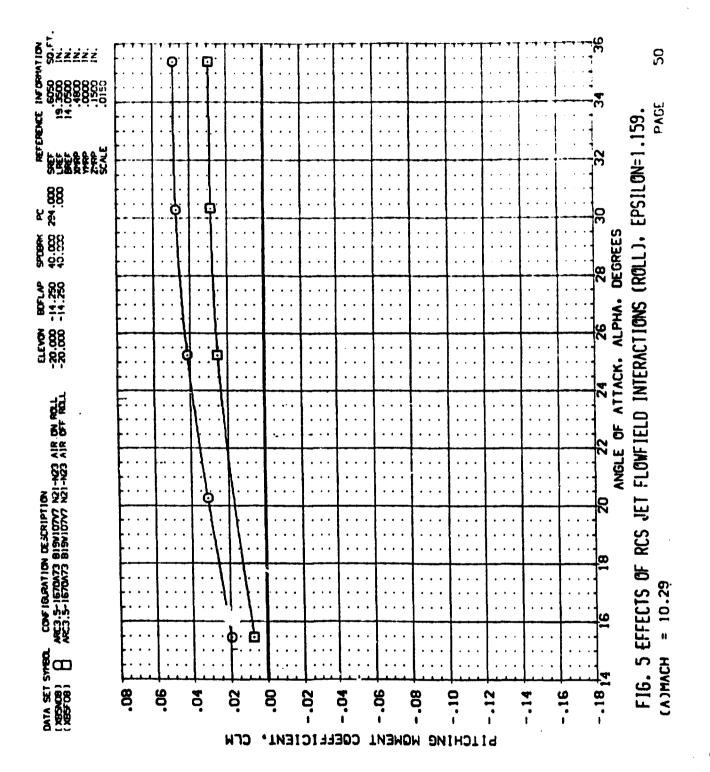




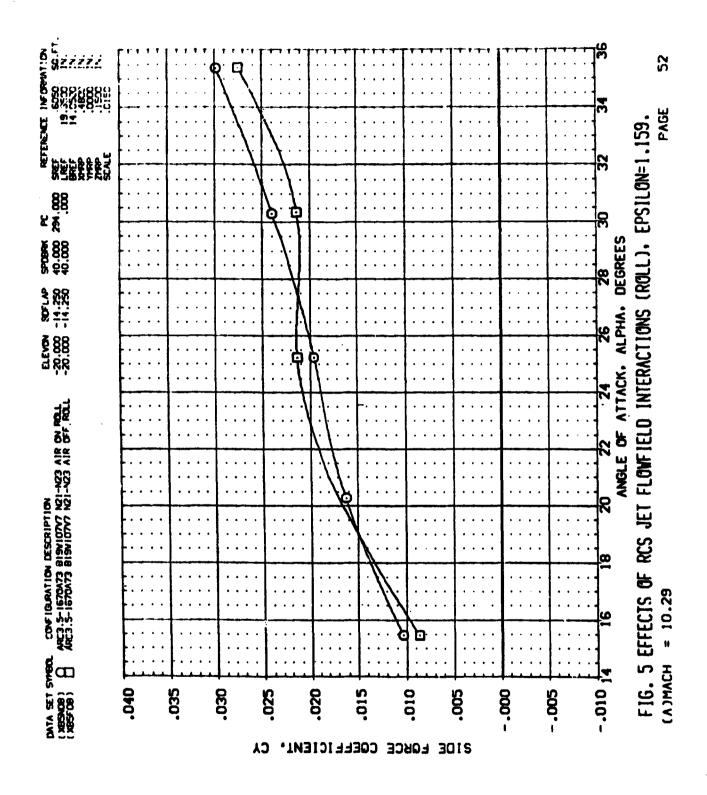


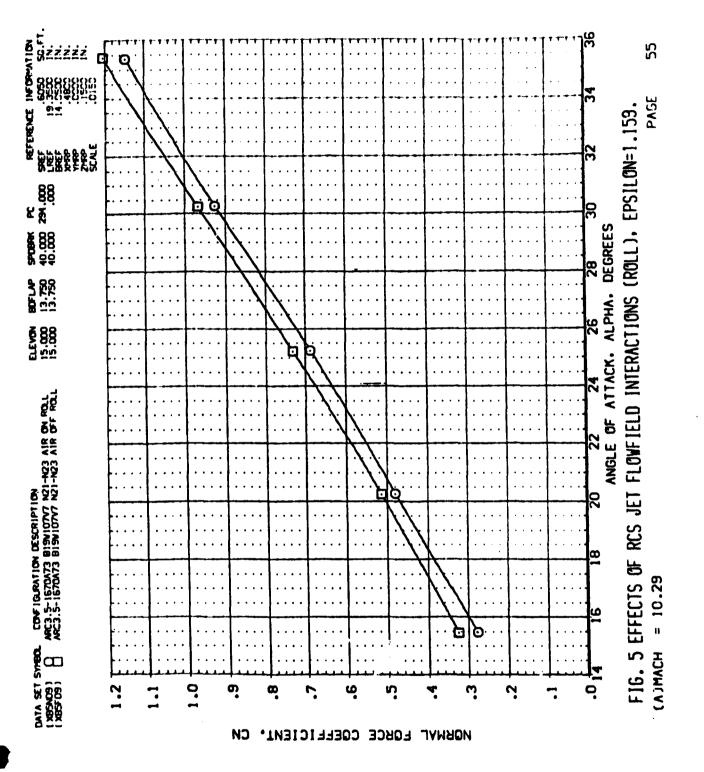
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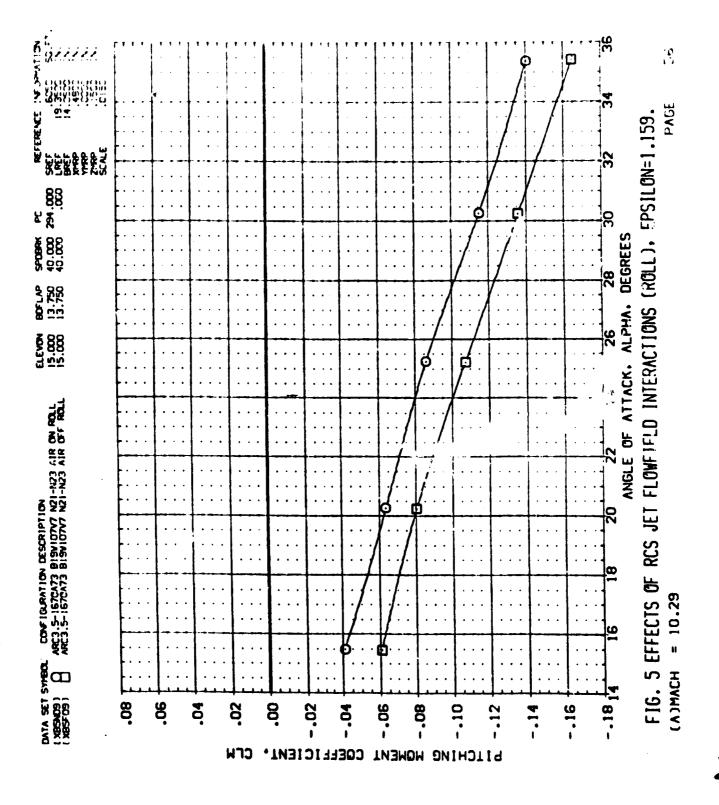
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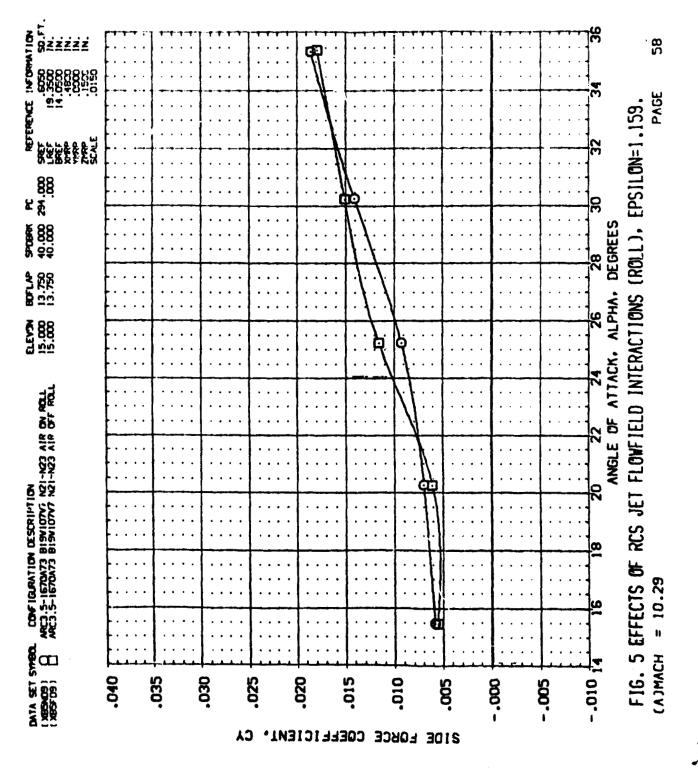




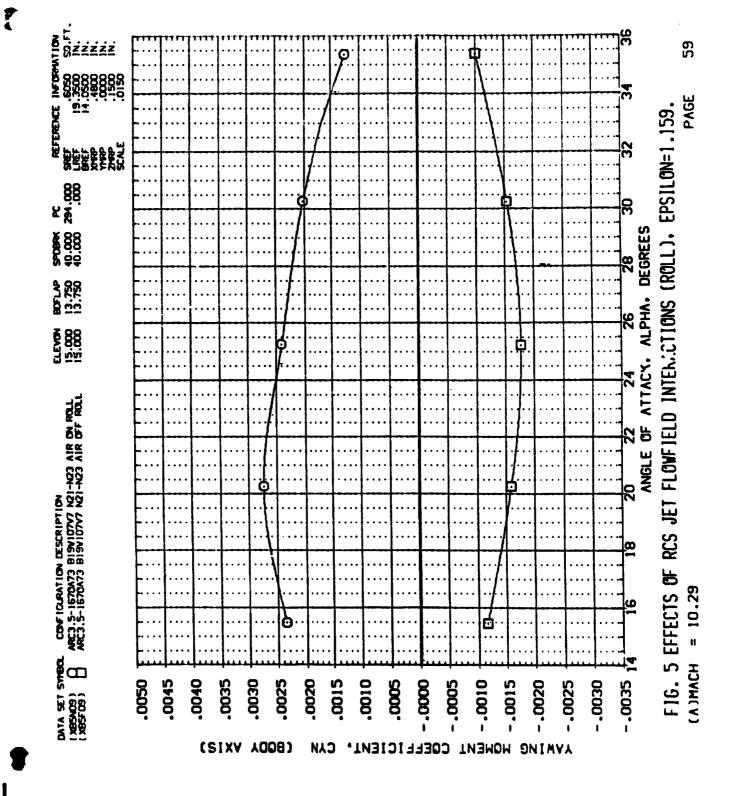


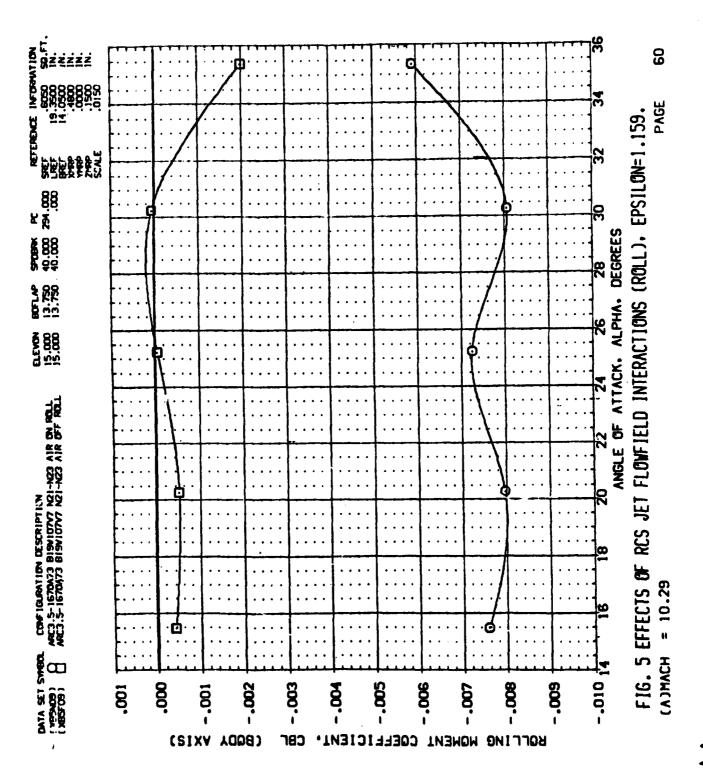
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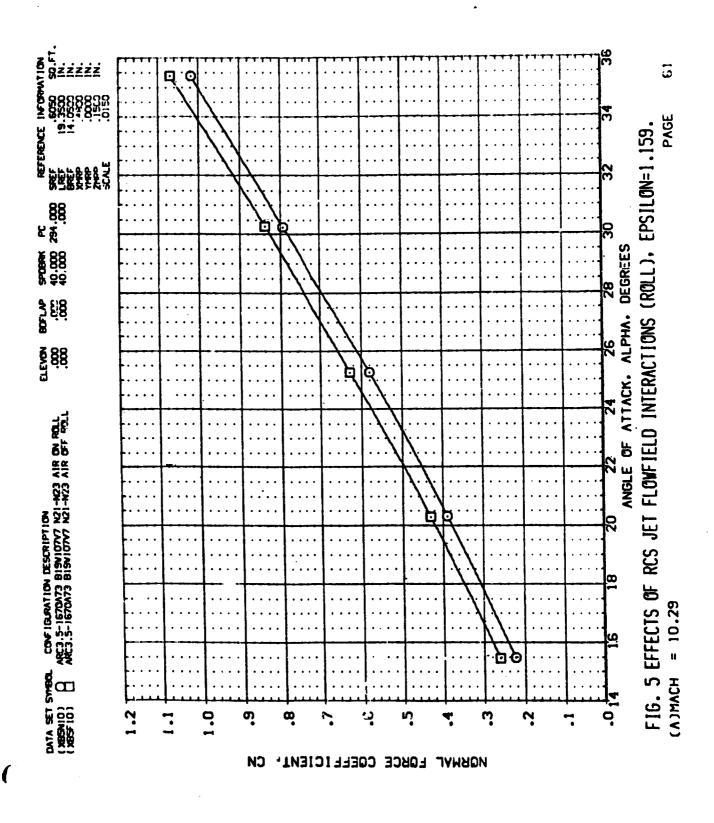


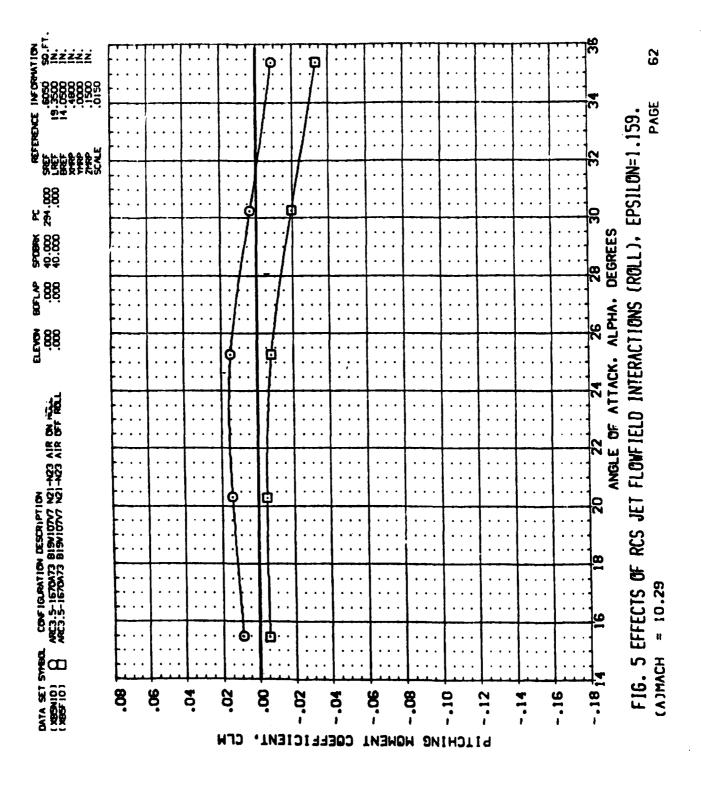
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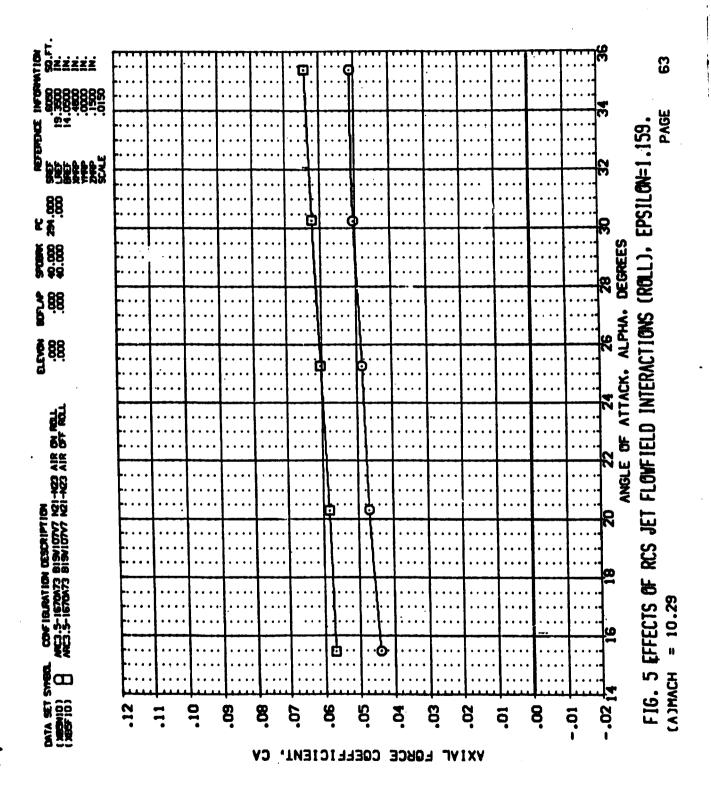


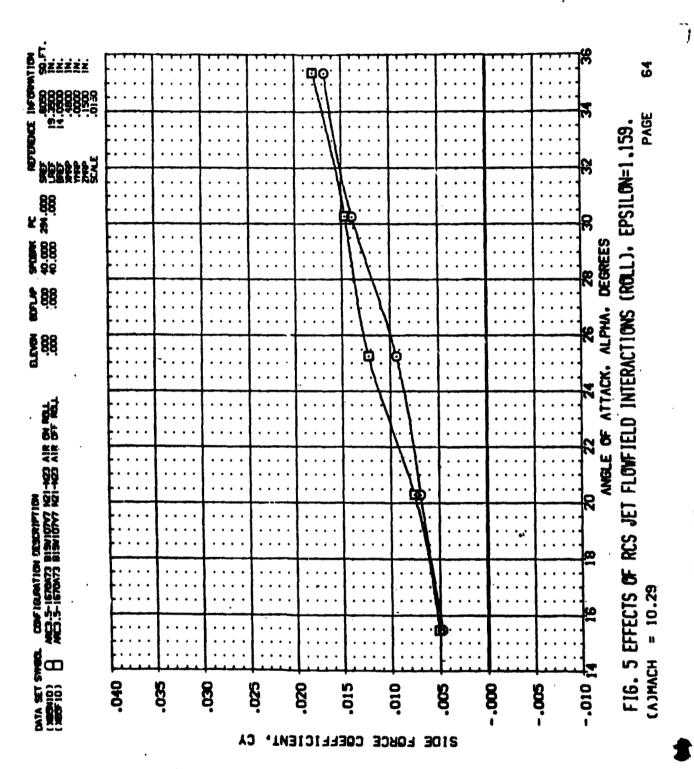


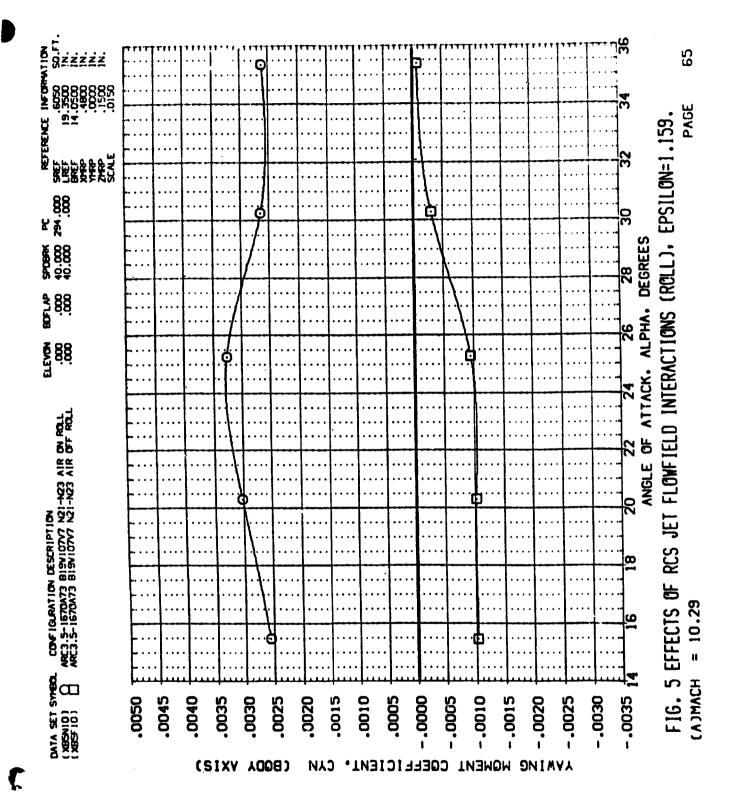
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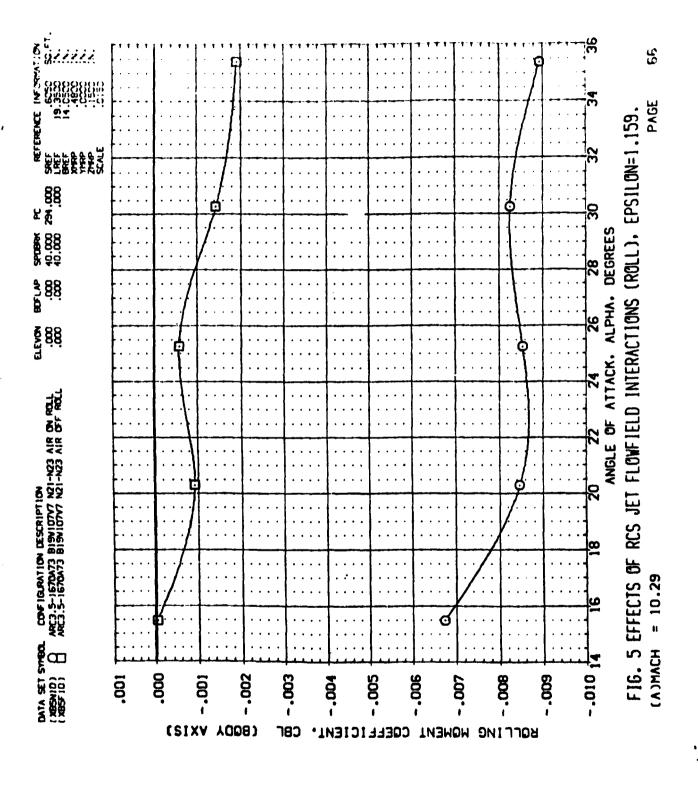


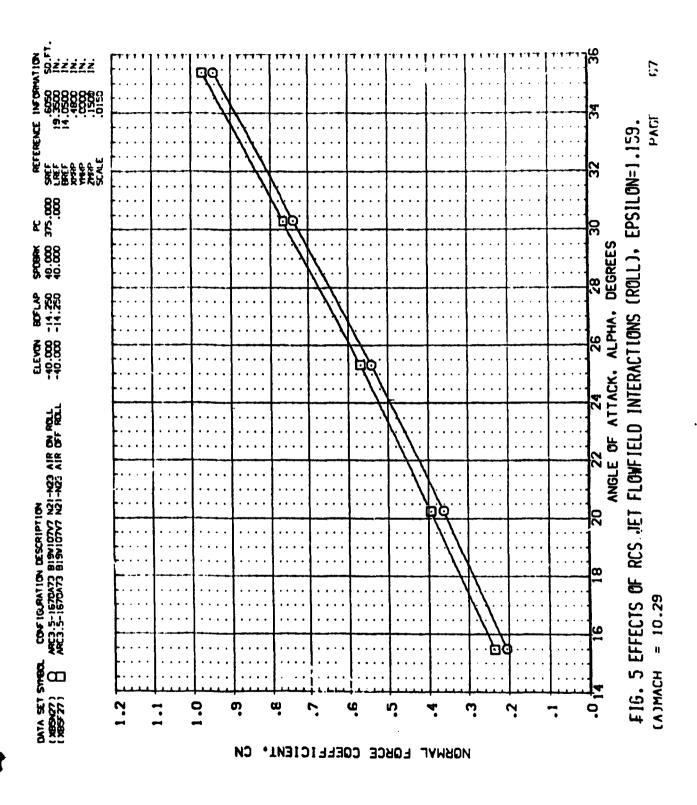






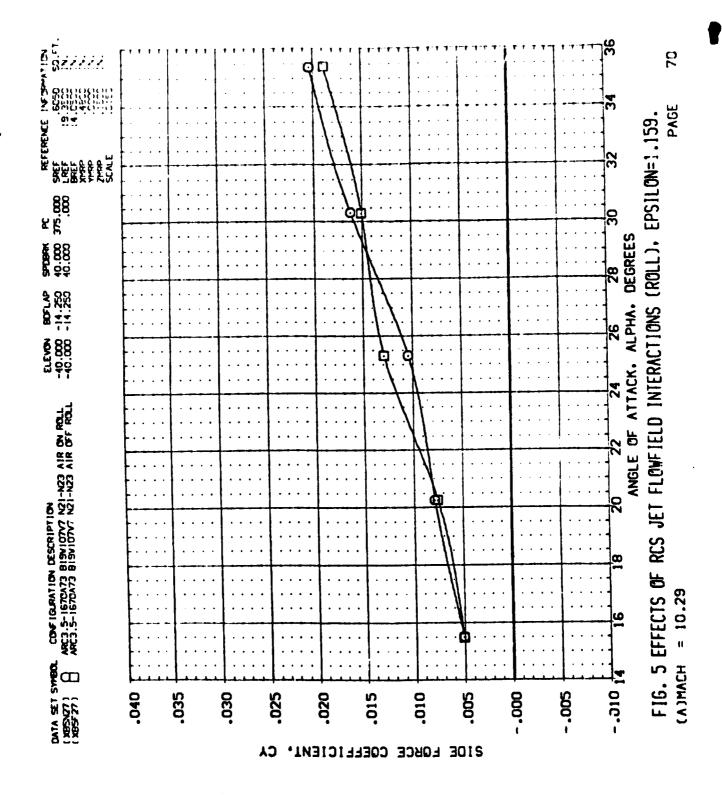


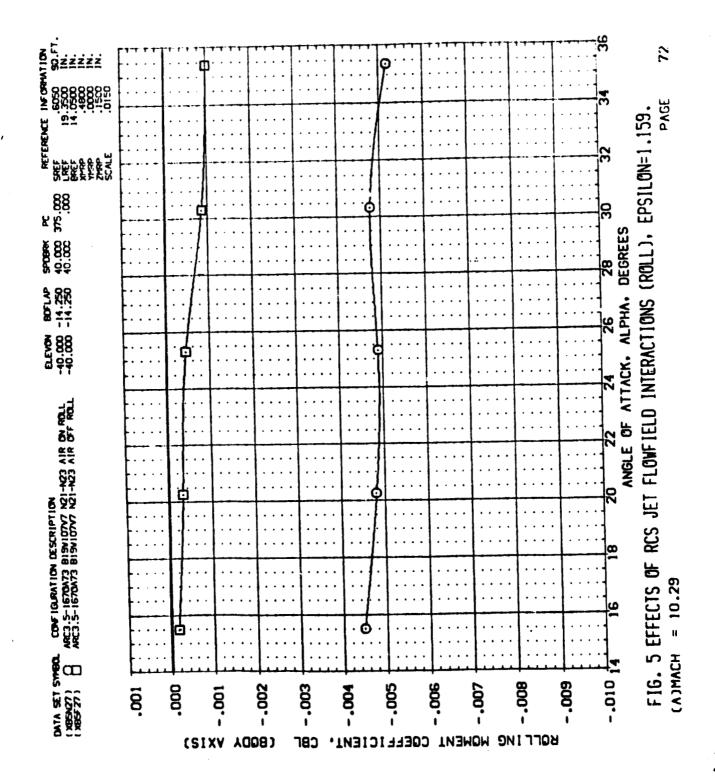




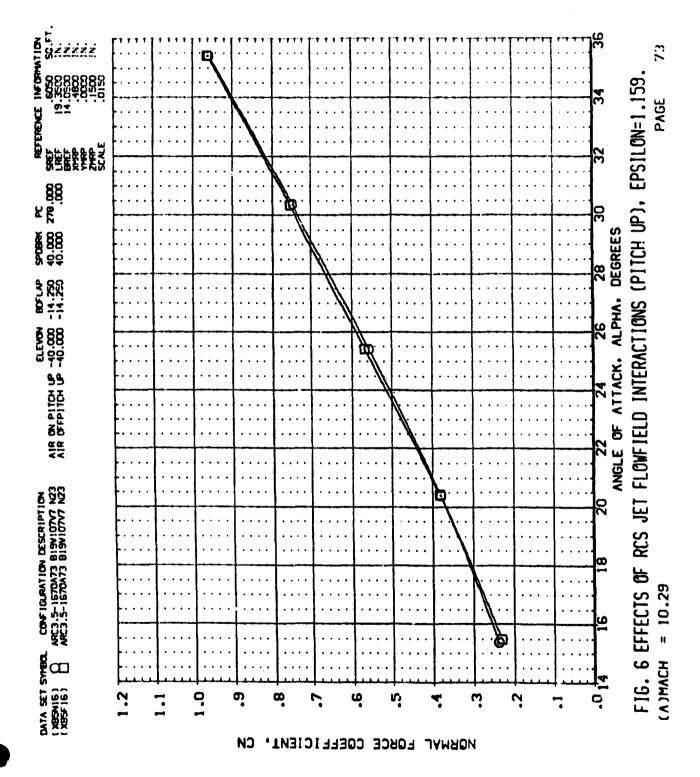
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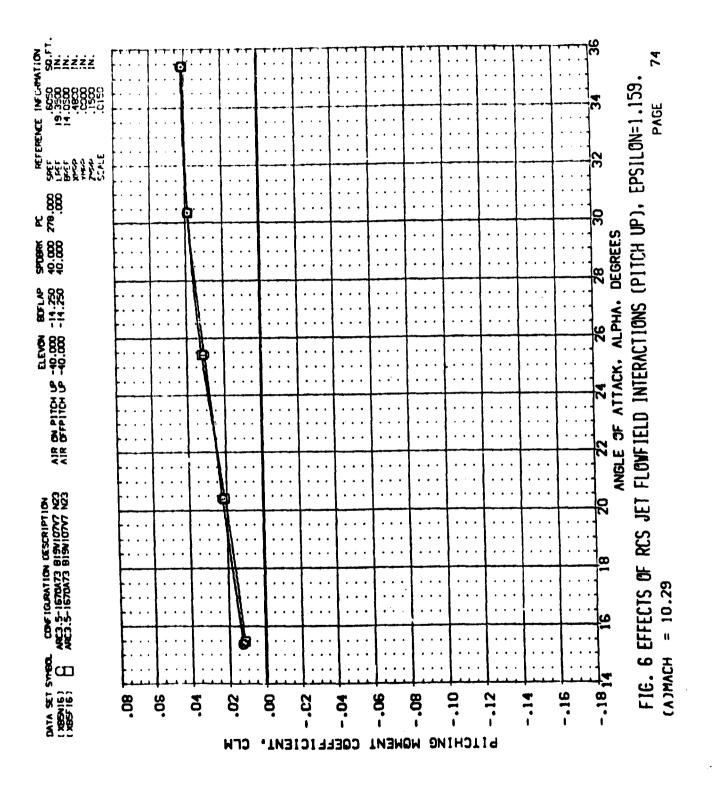
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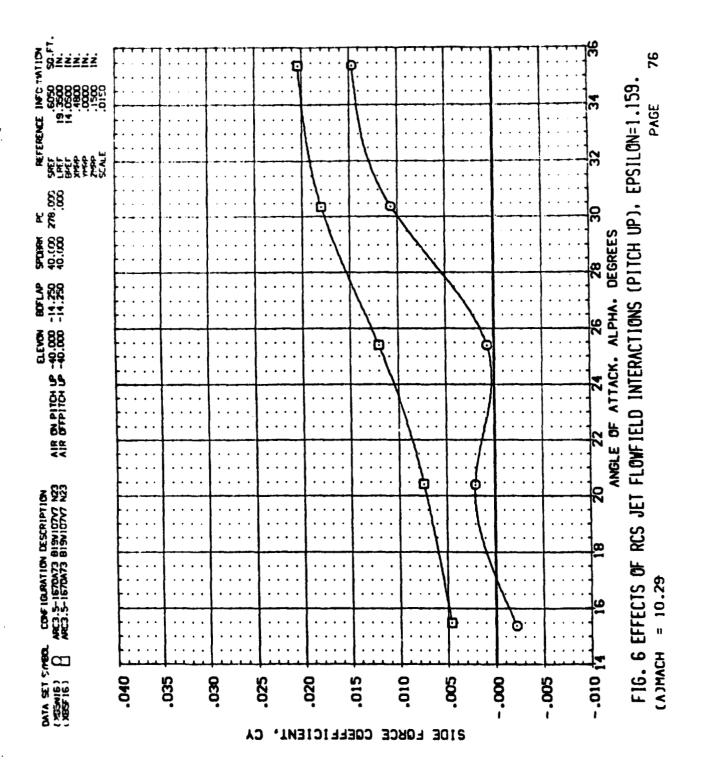
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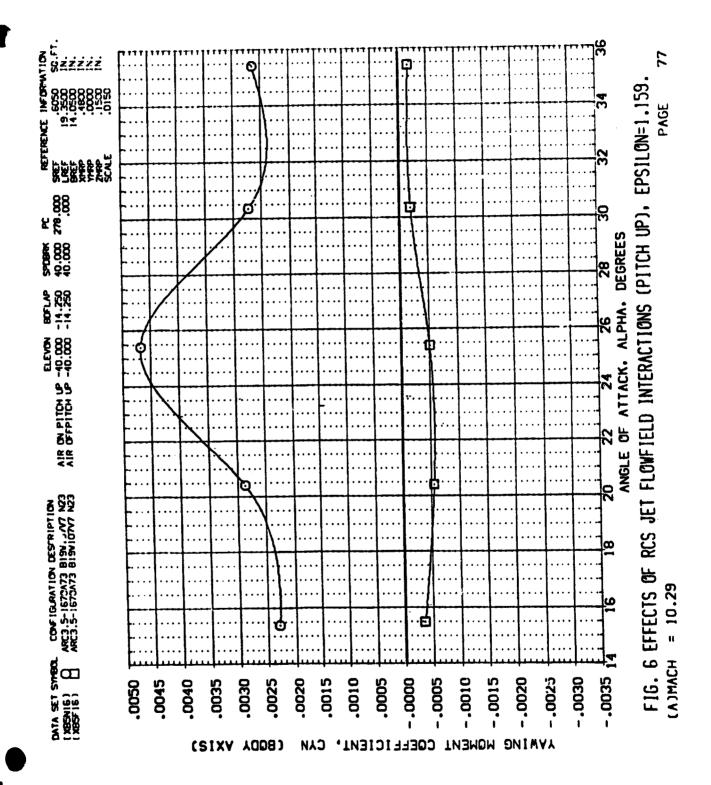


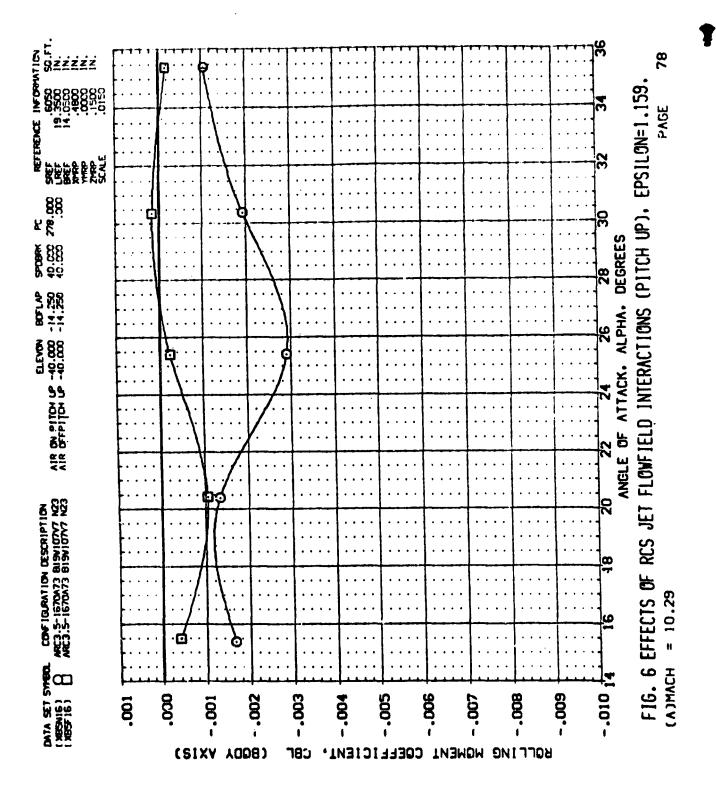
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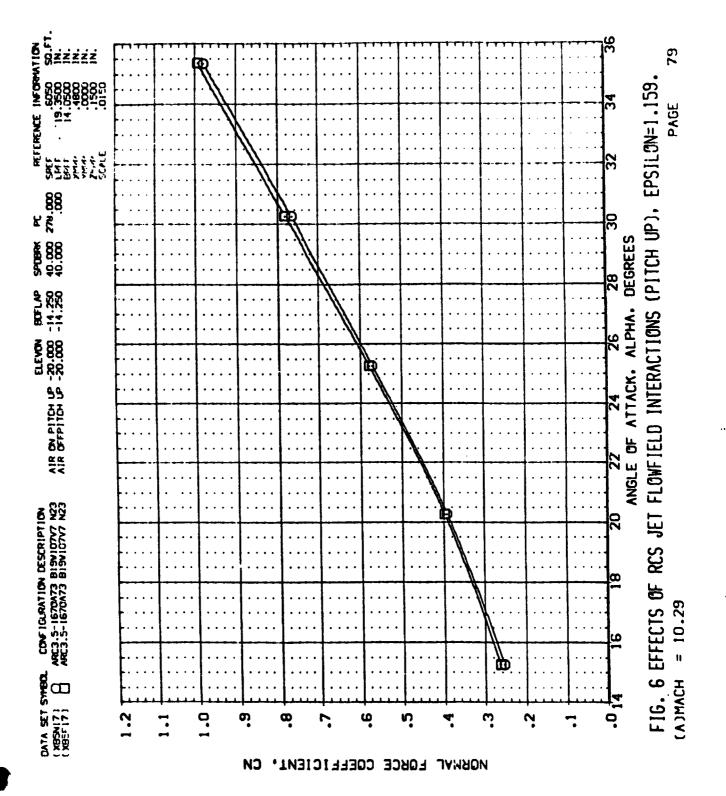
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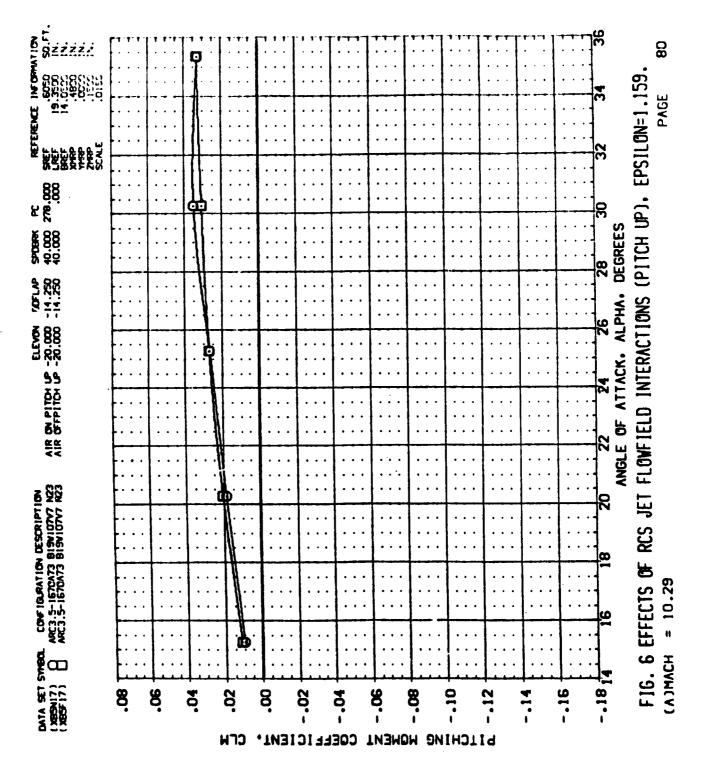


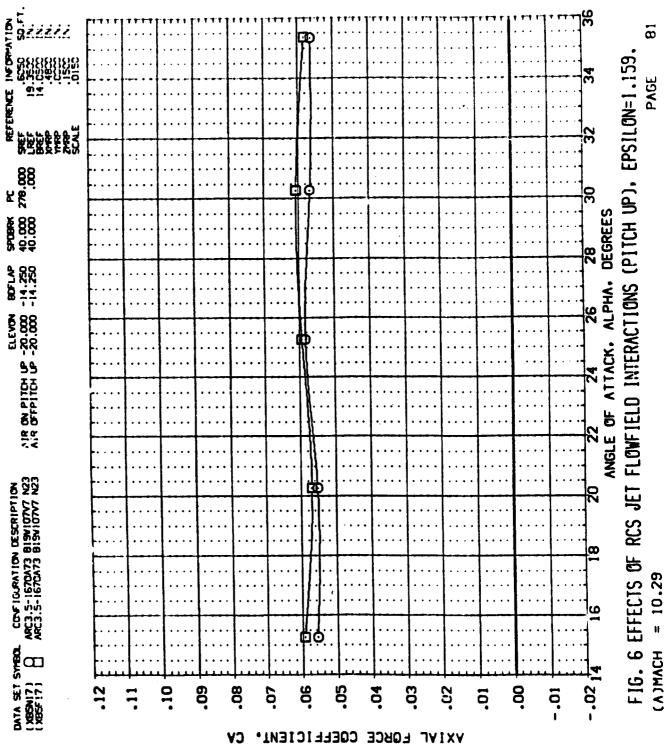
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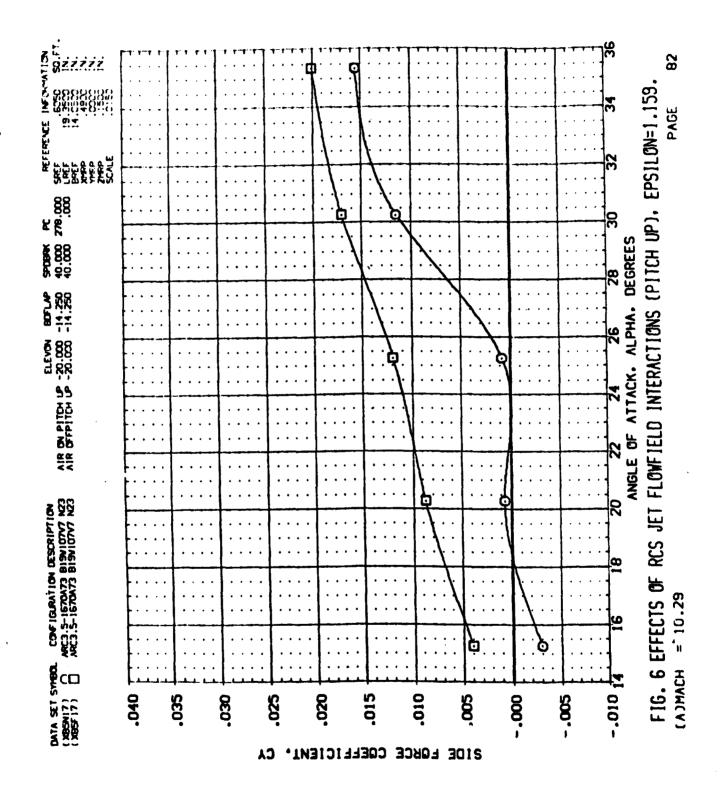


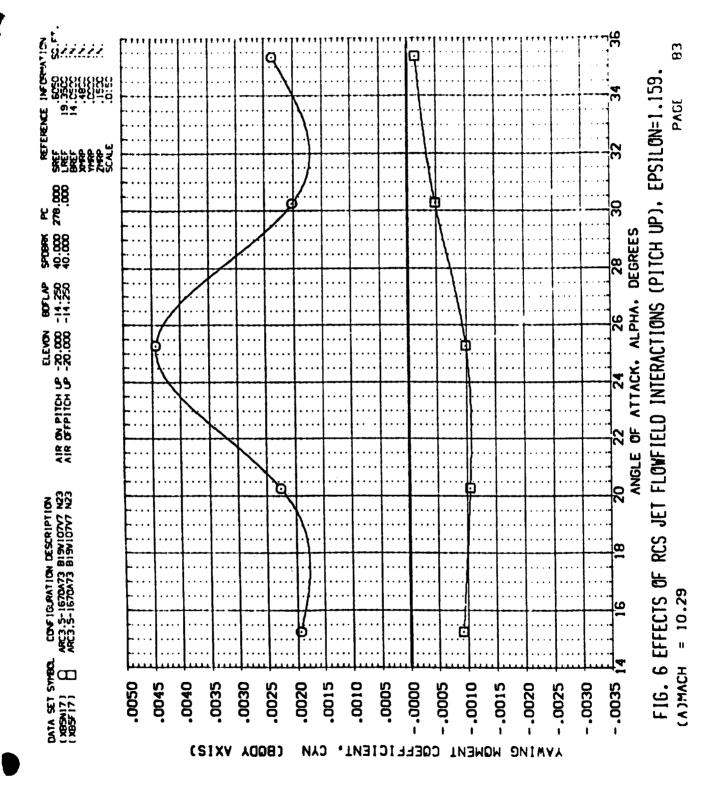


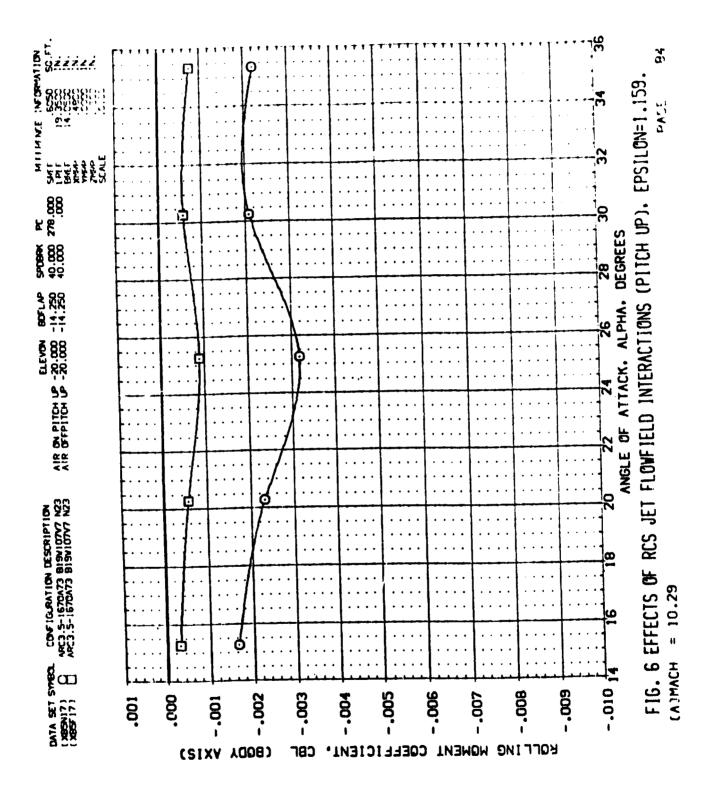


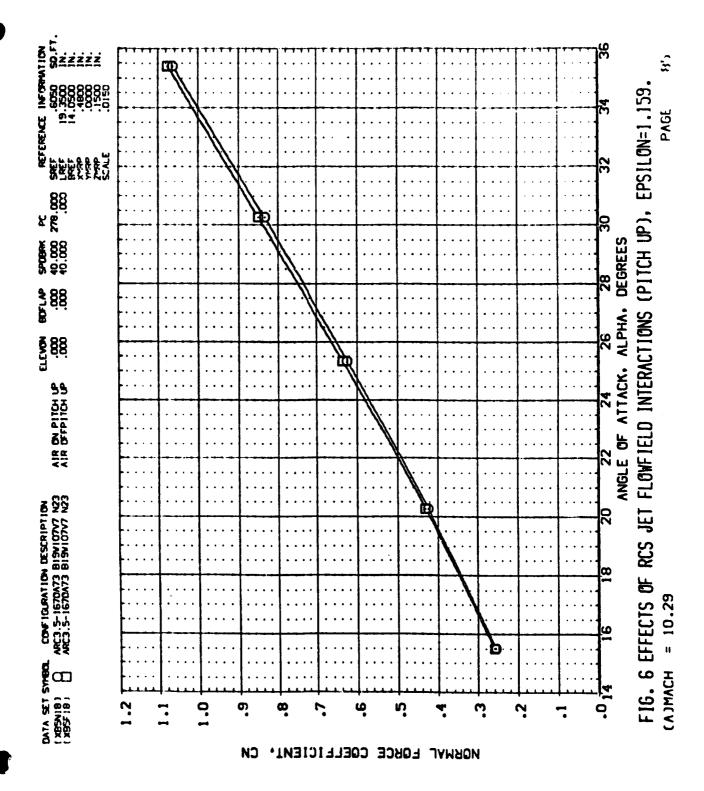


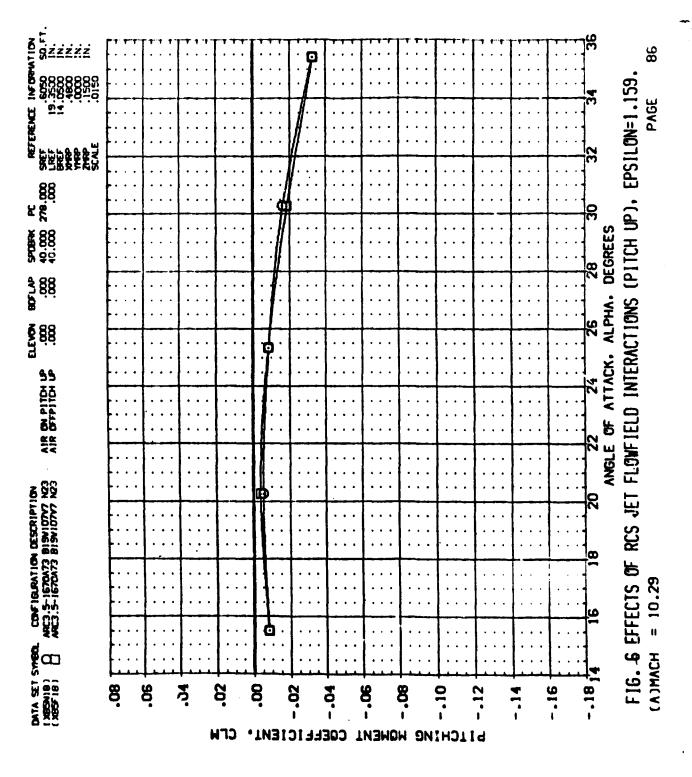


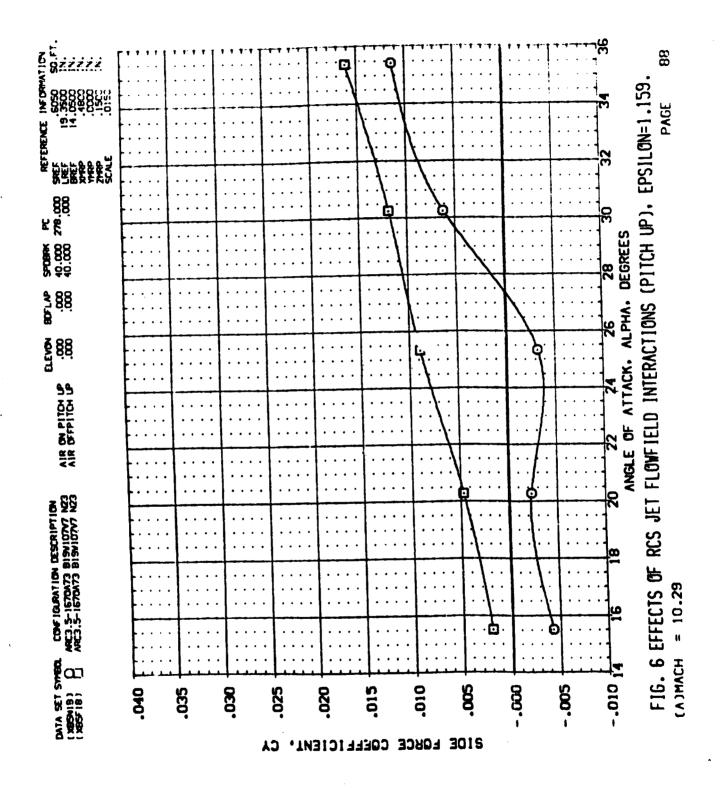


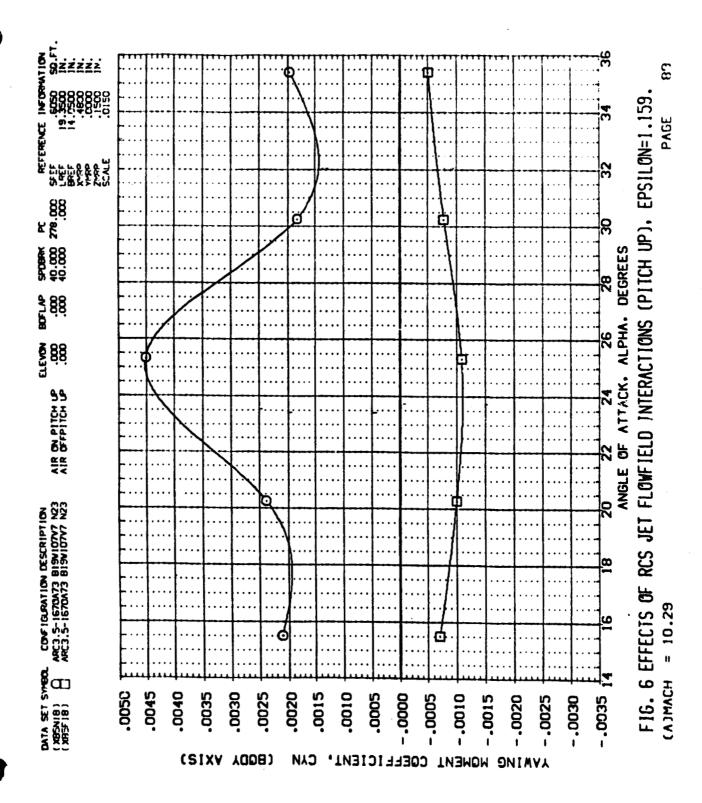


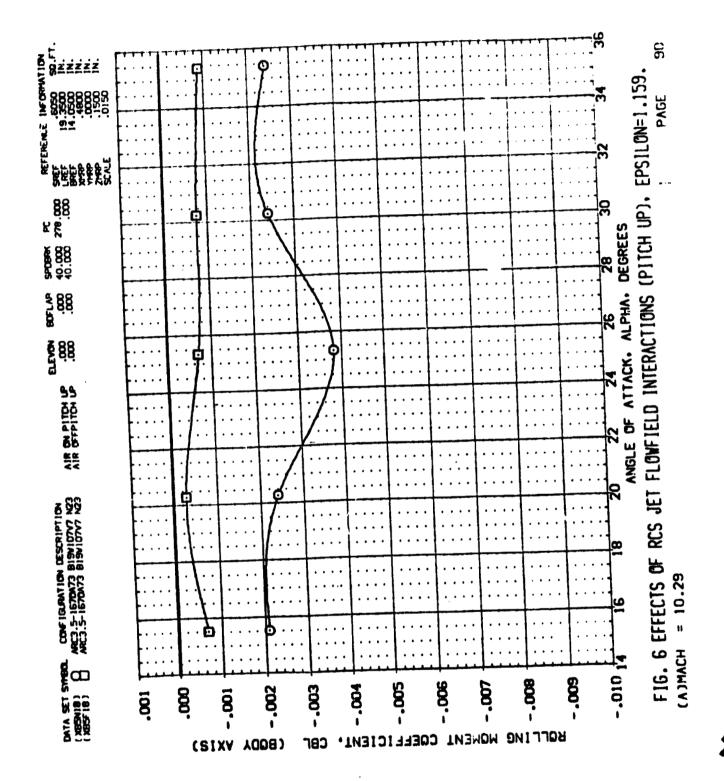




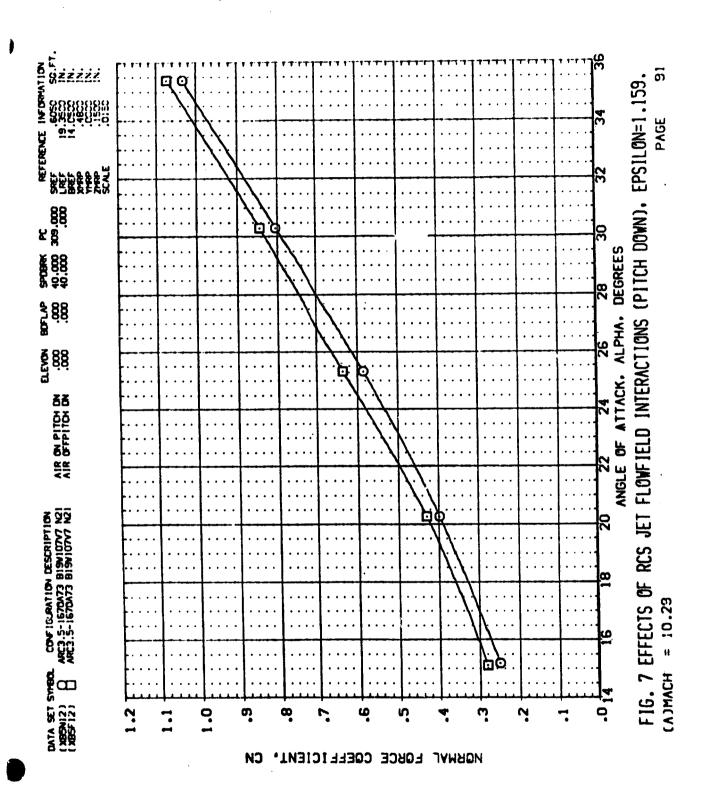








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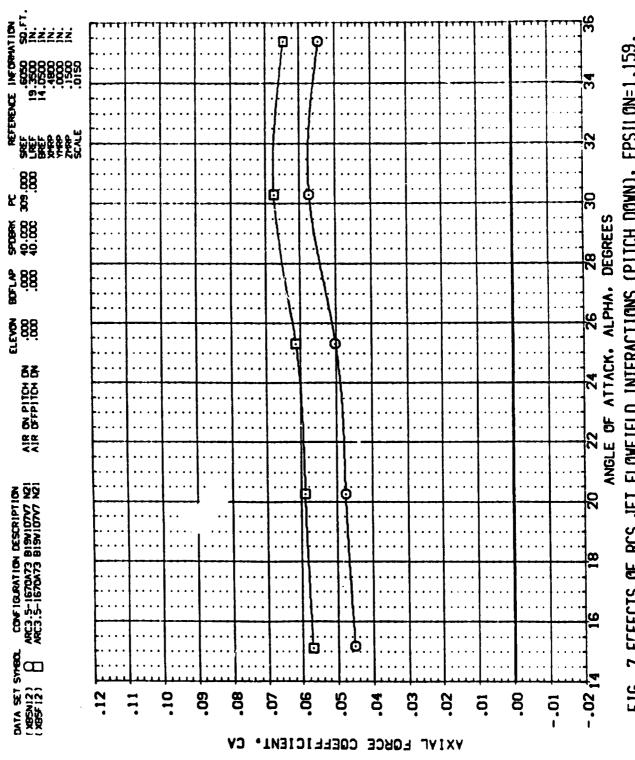
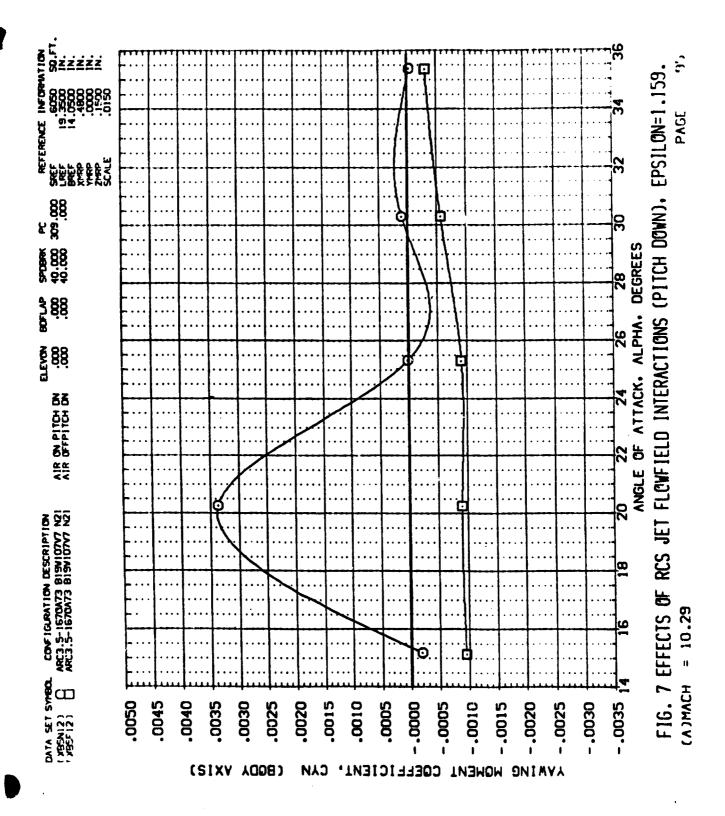
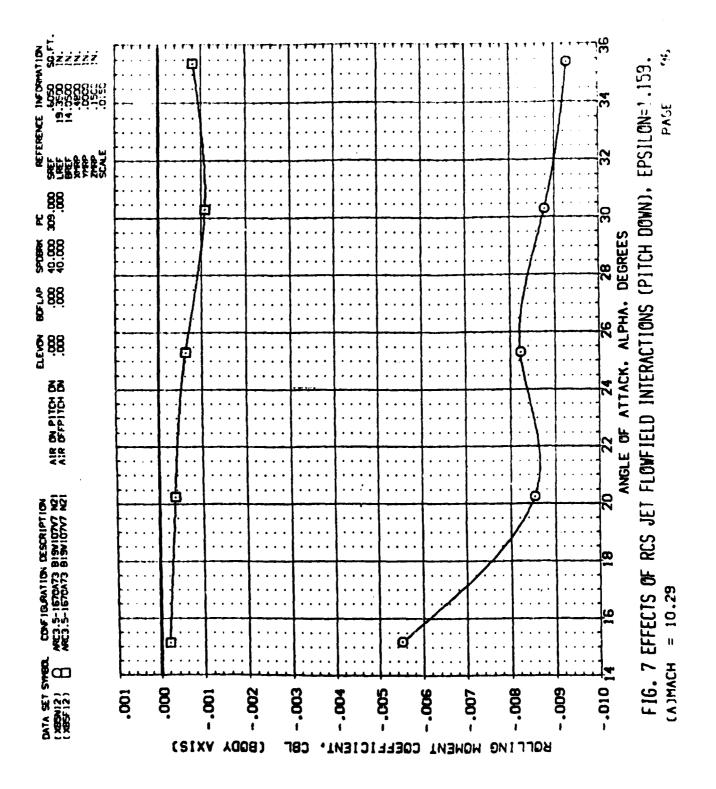
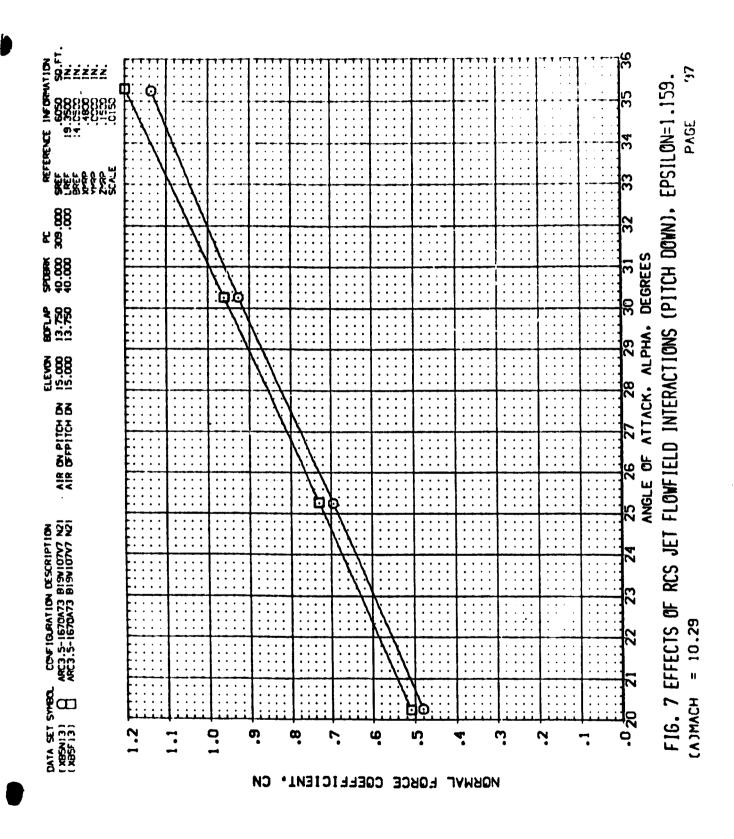


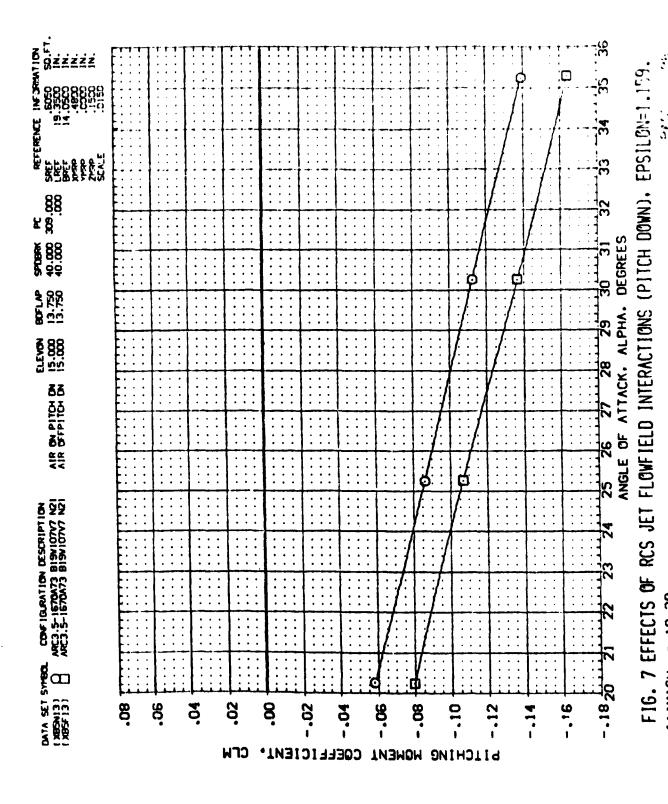
FIG. 7 EFFECTS OF RCS JET FLOWFIELD INTERACTIONS (PITCH DOWN). EPSILON=1.159. PAGE (A)MACH = 10.29



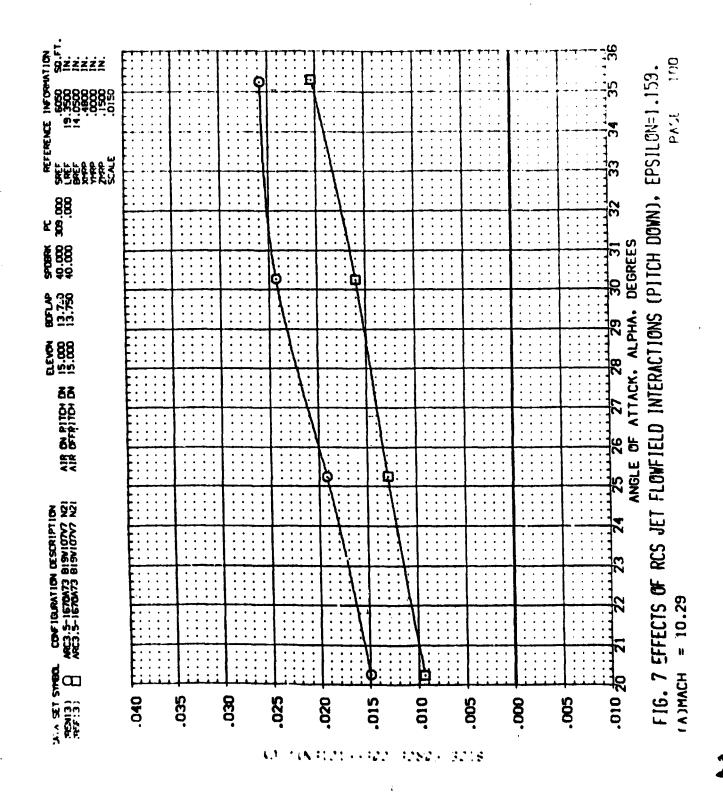


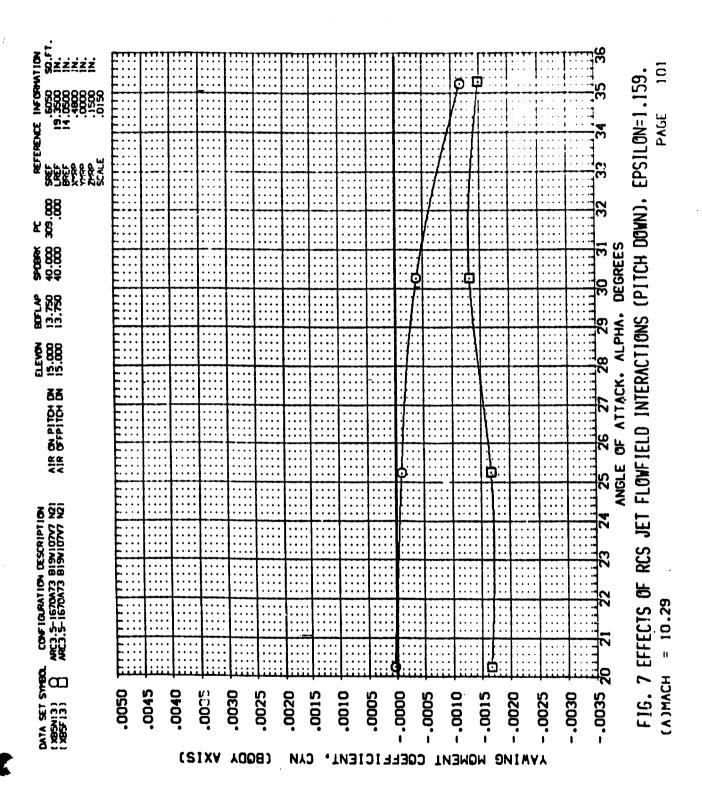


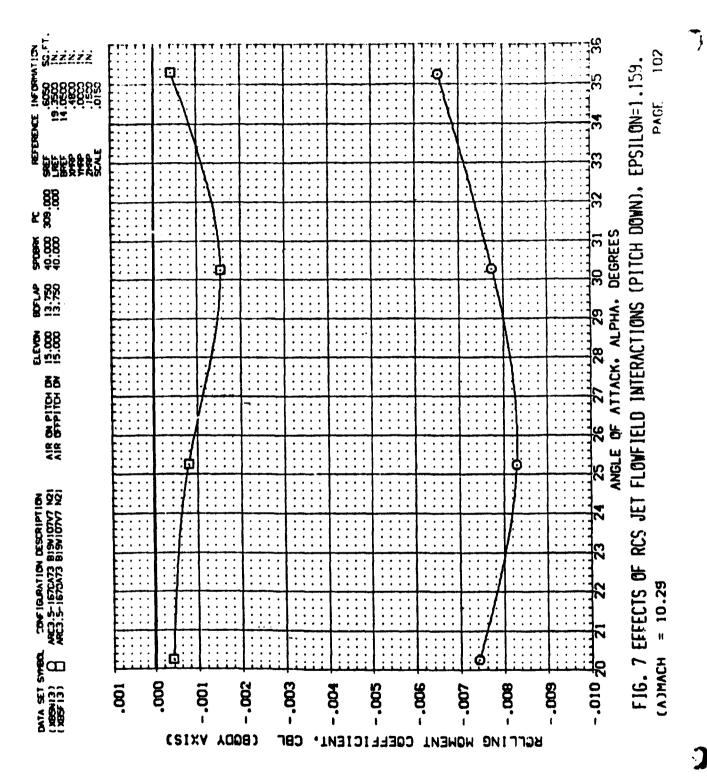
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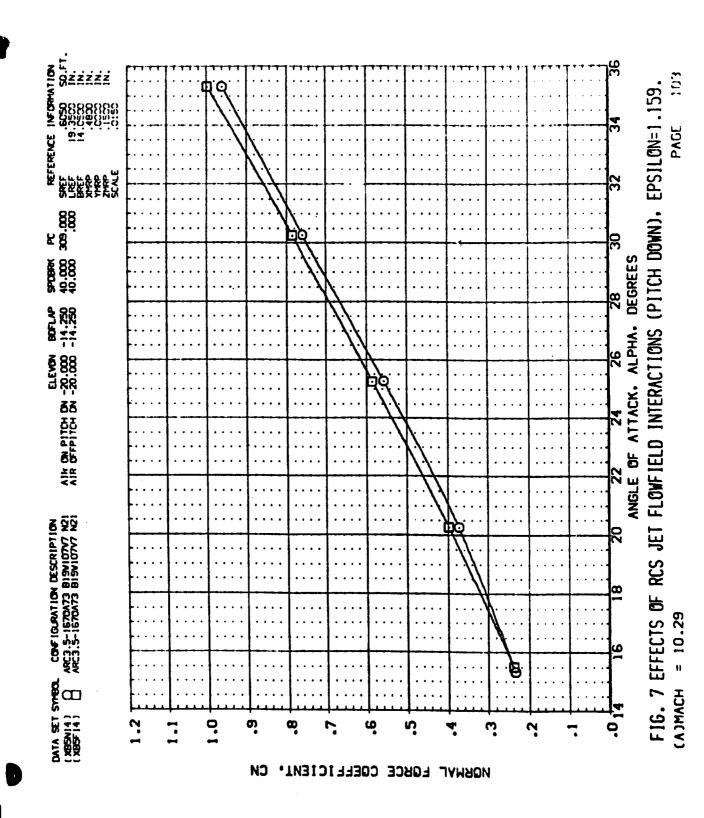


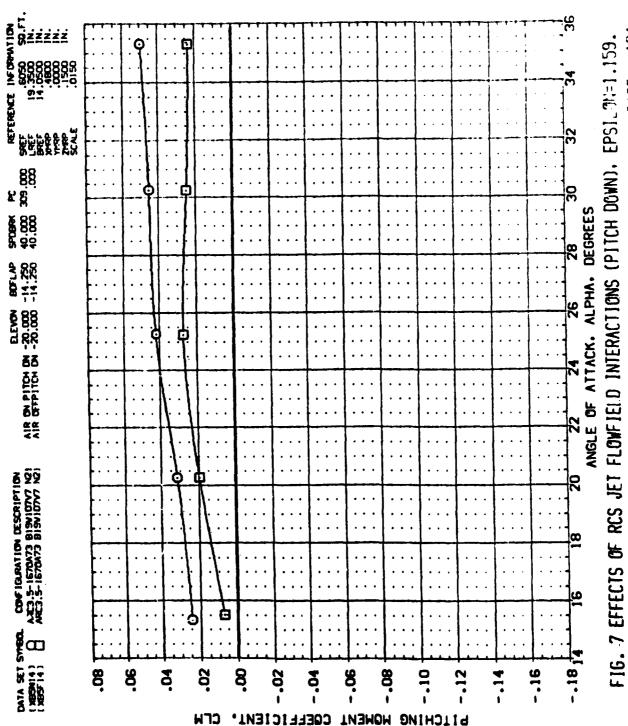
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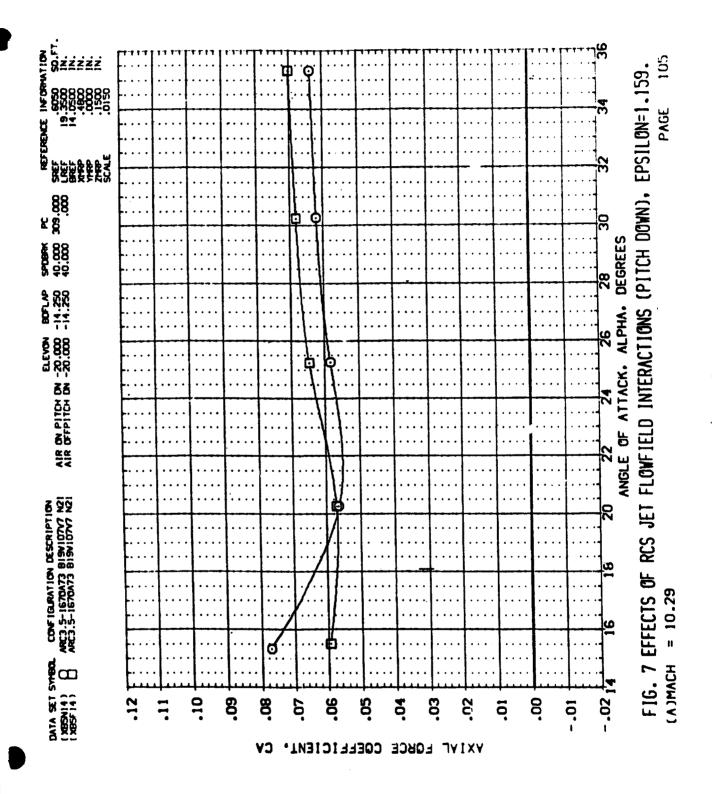


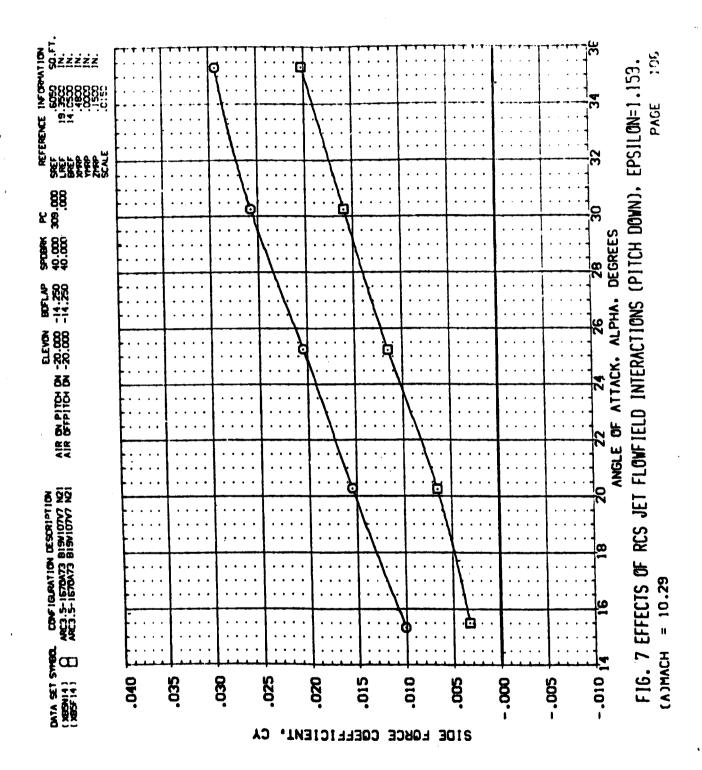


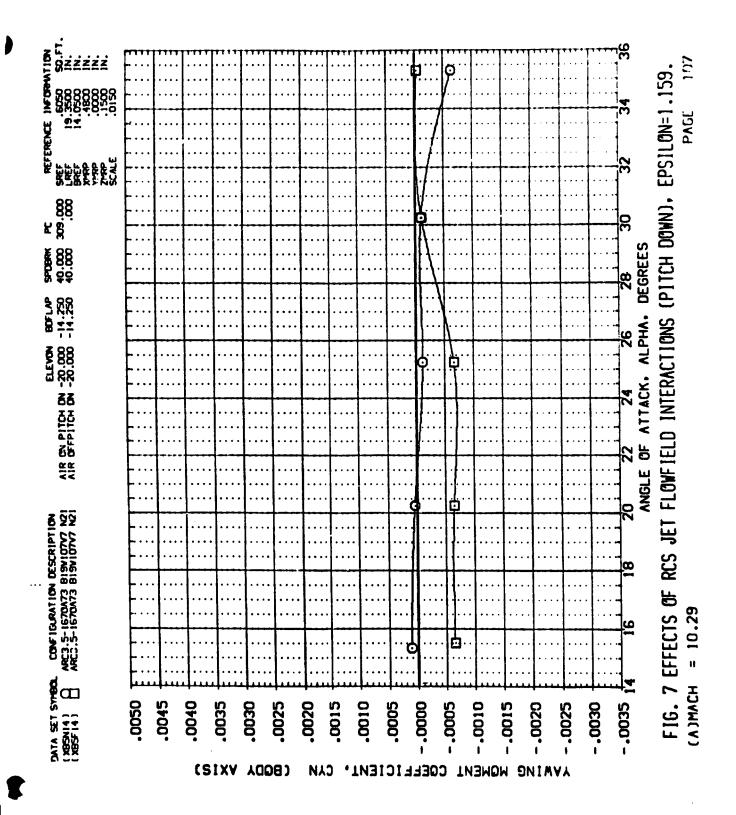


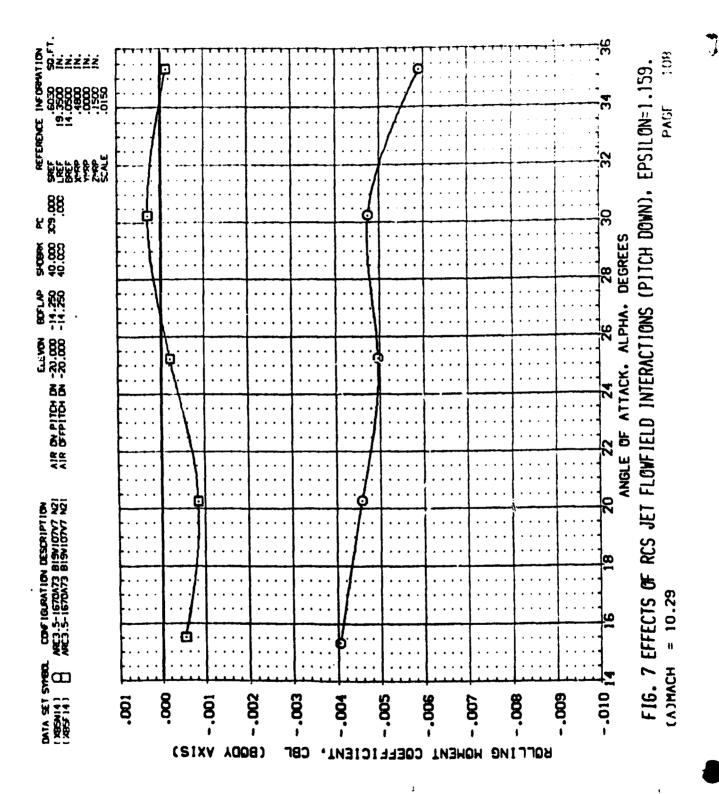
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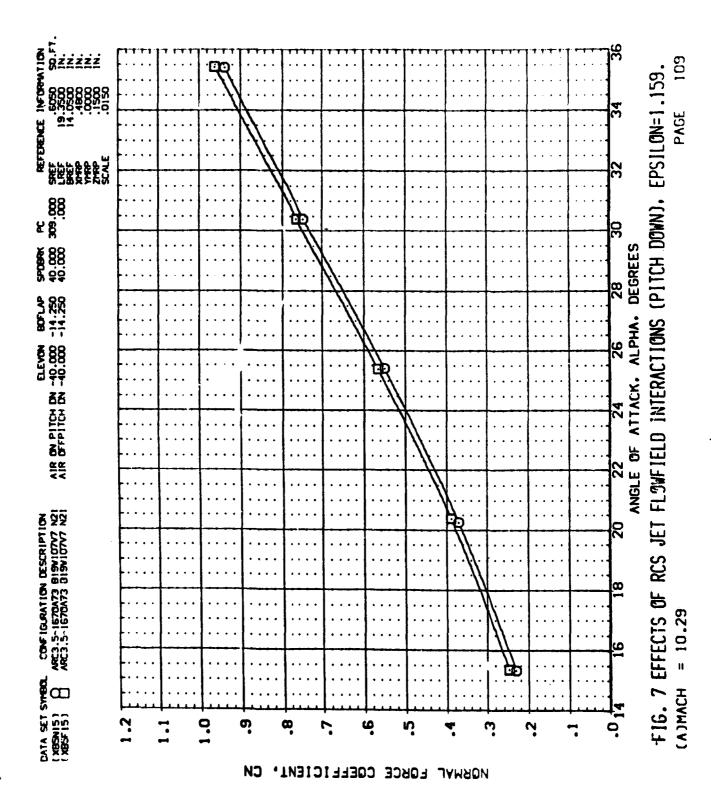
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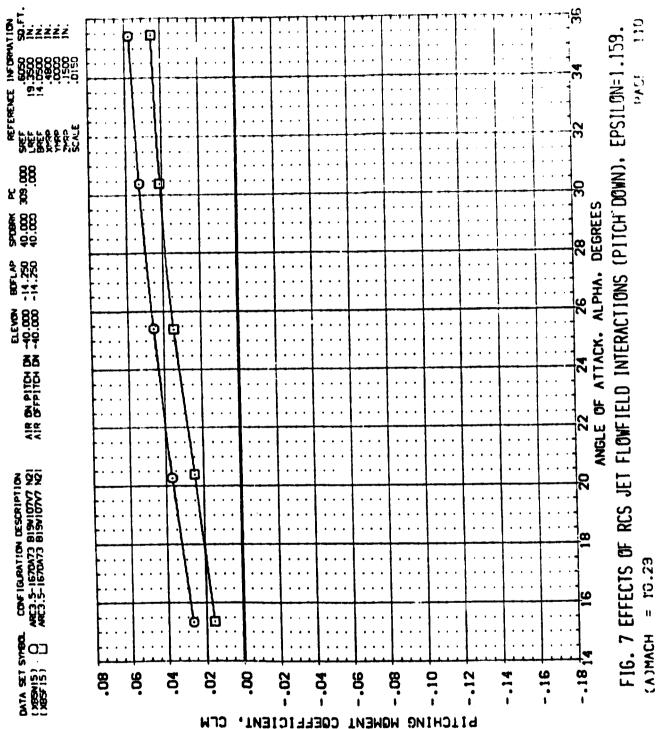




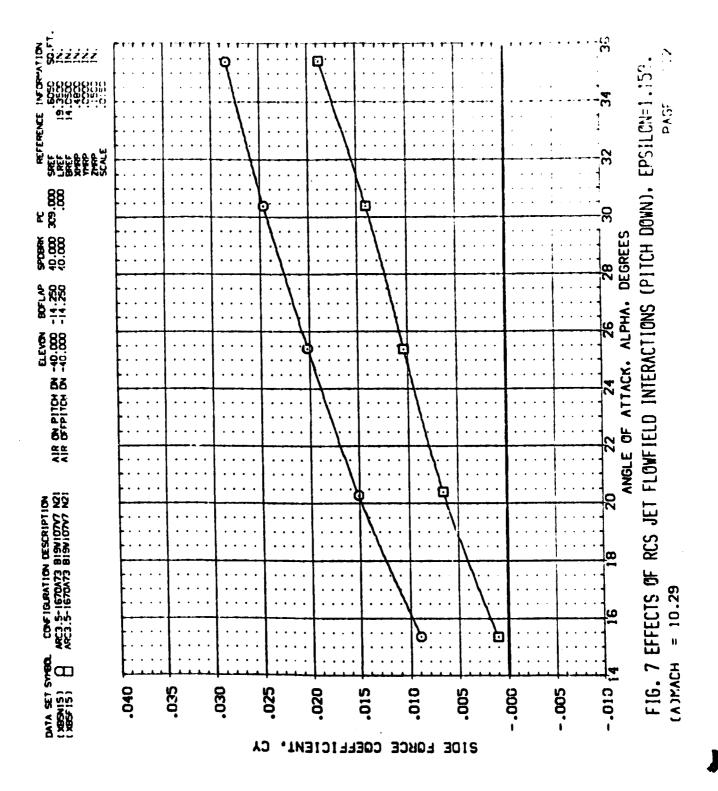


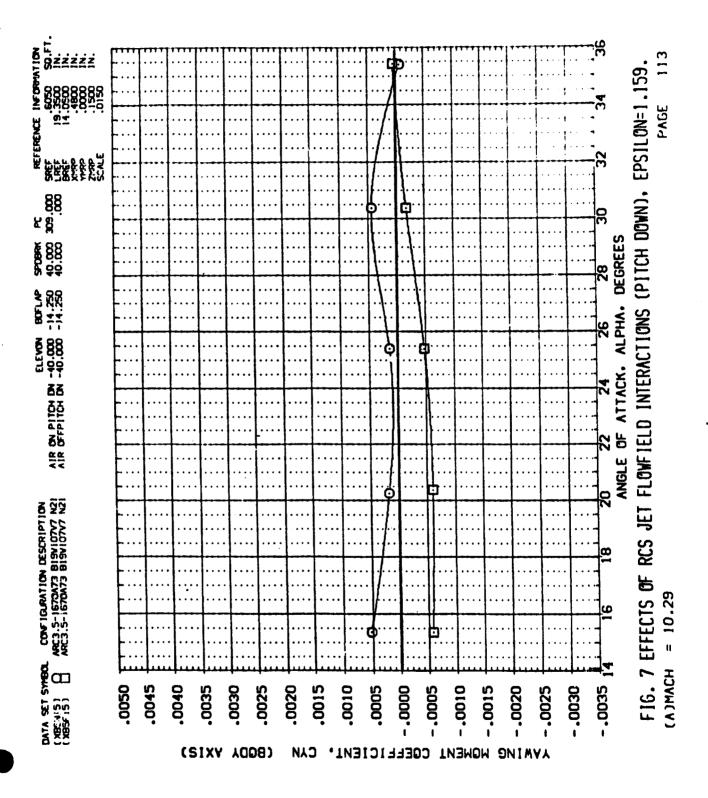


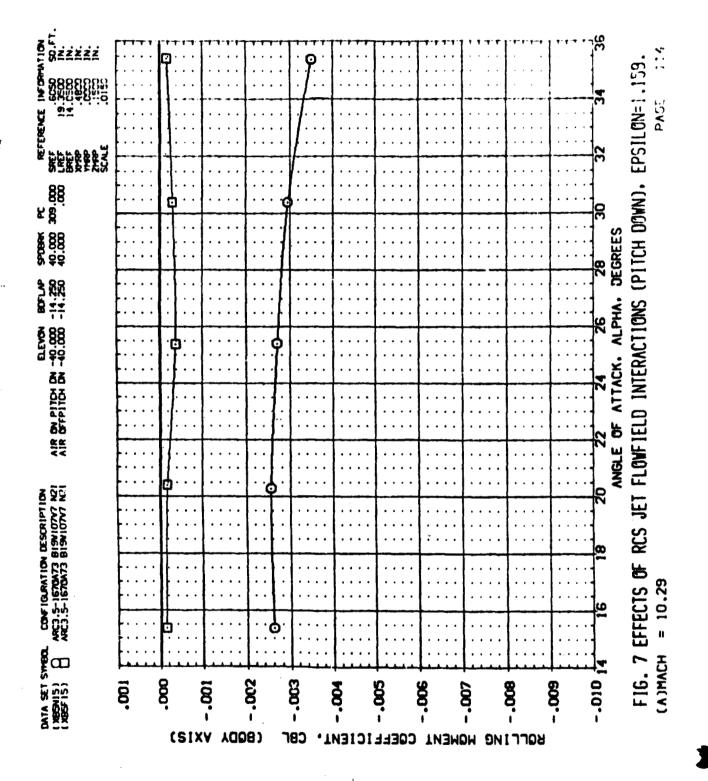




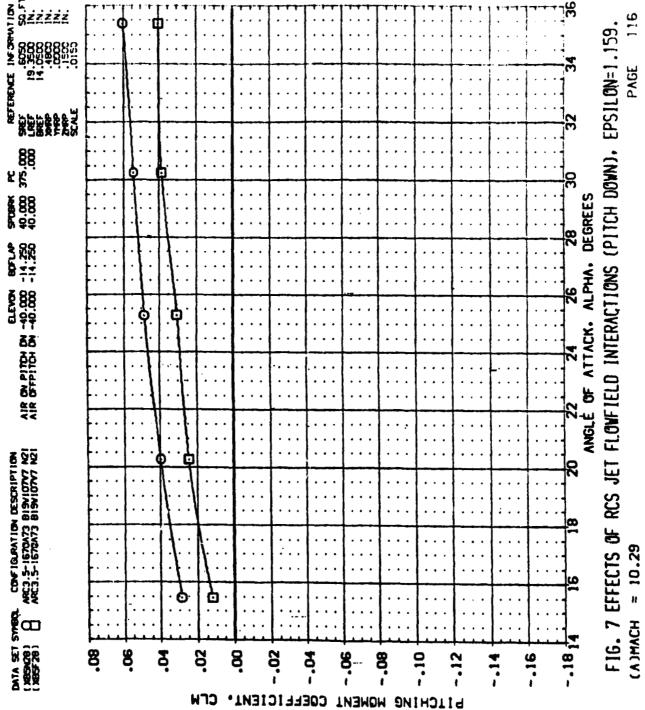
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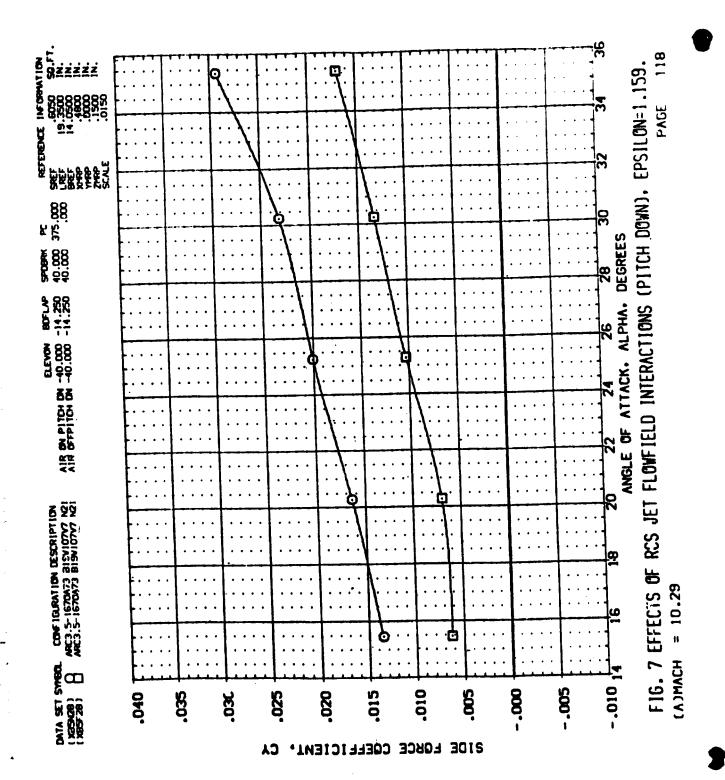






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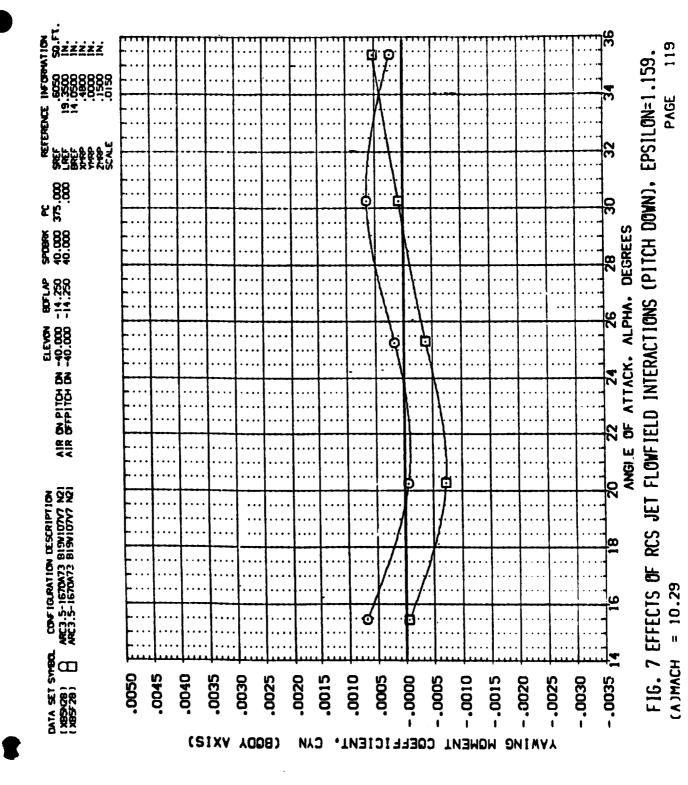




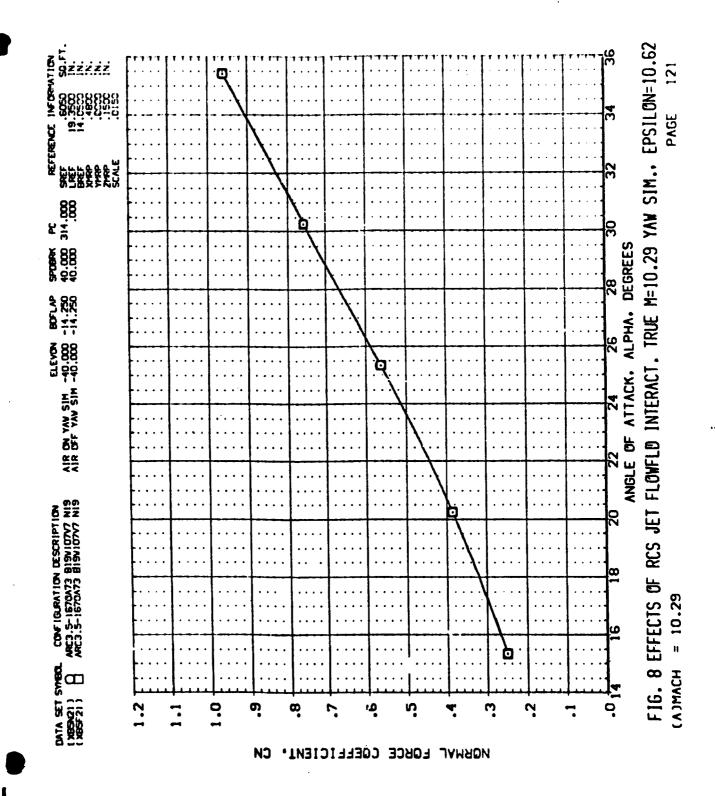
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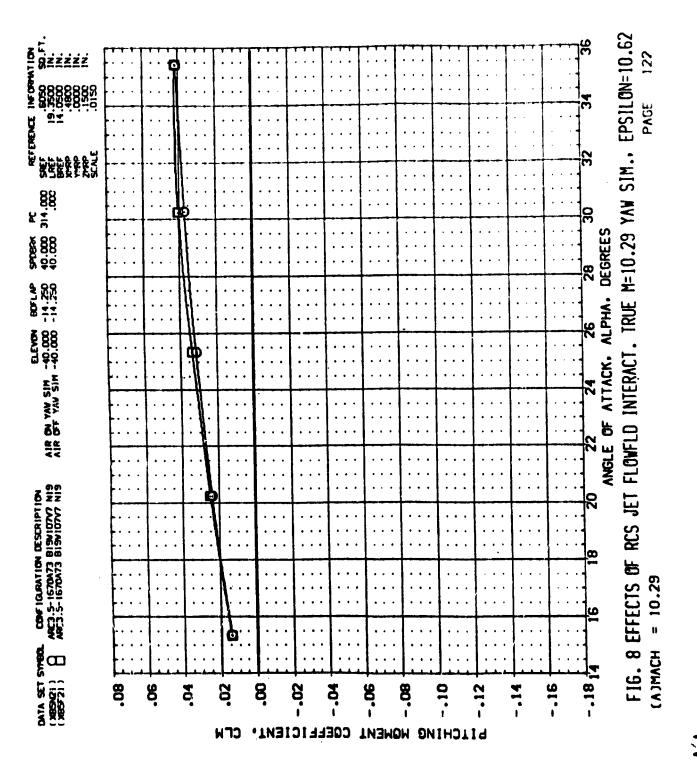
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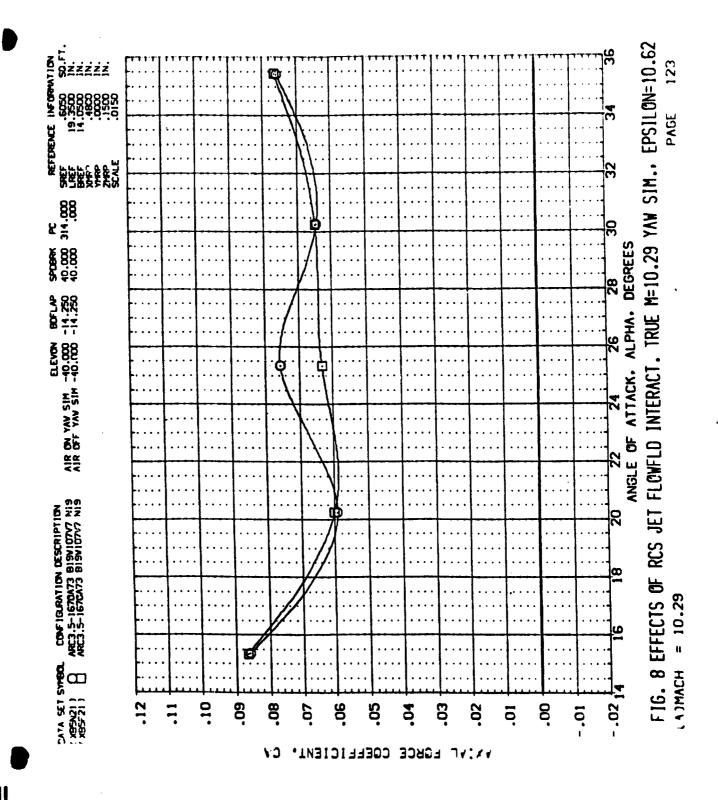


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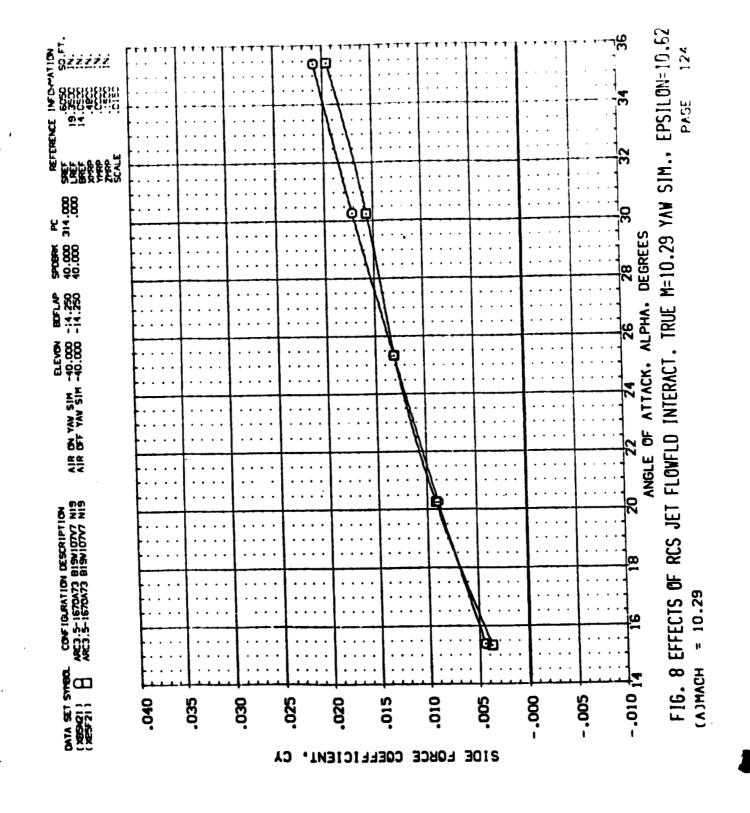


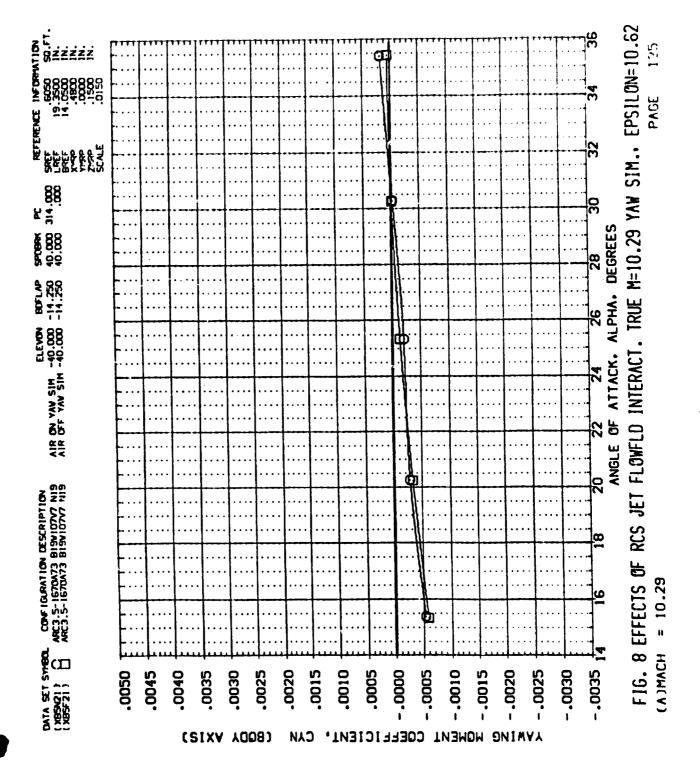
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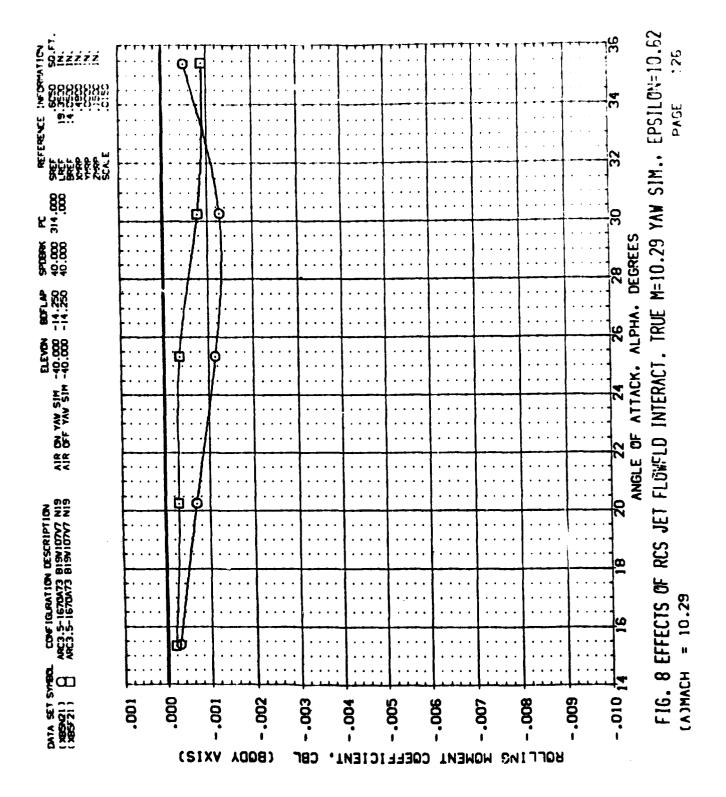


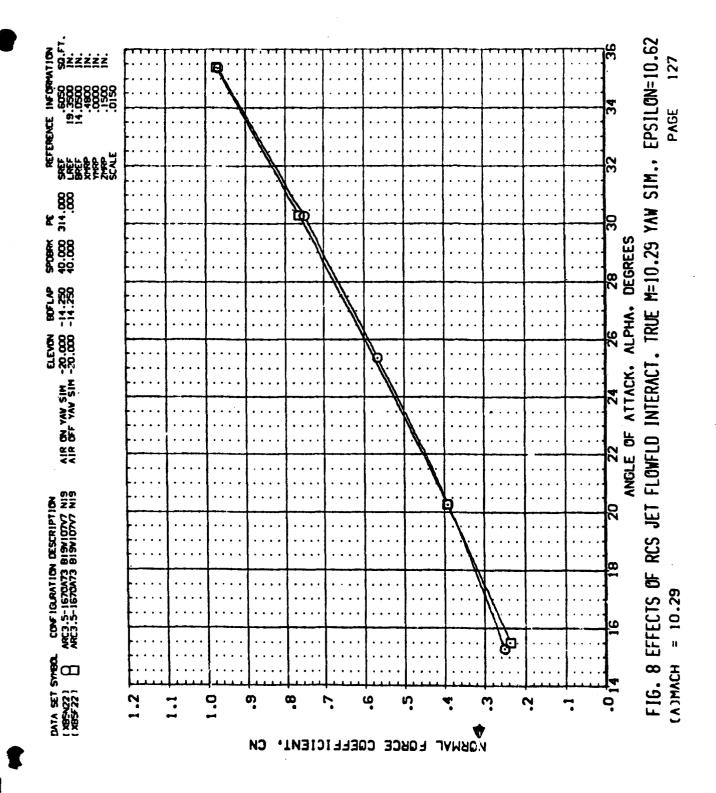
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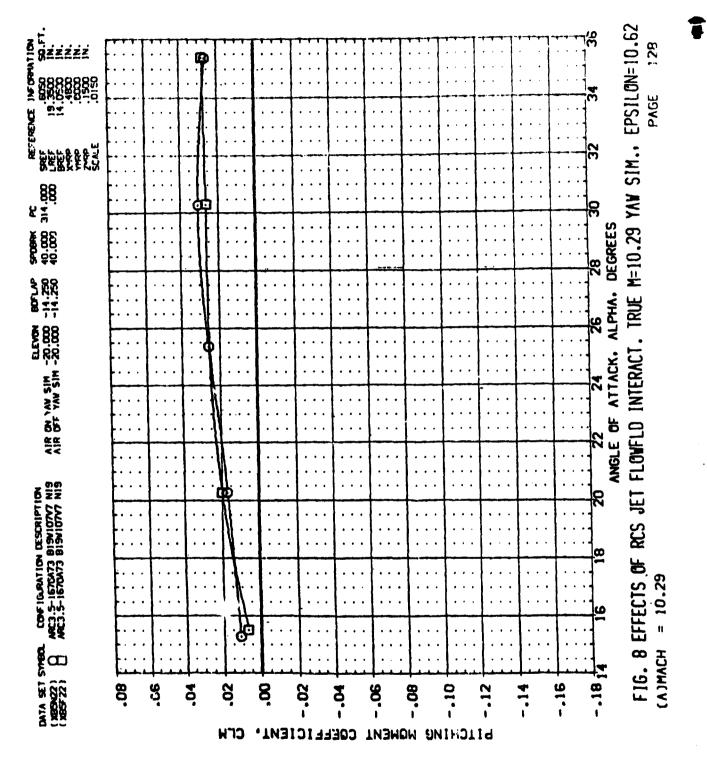
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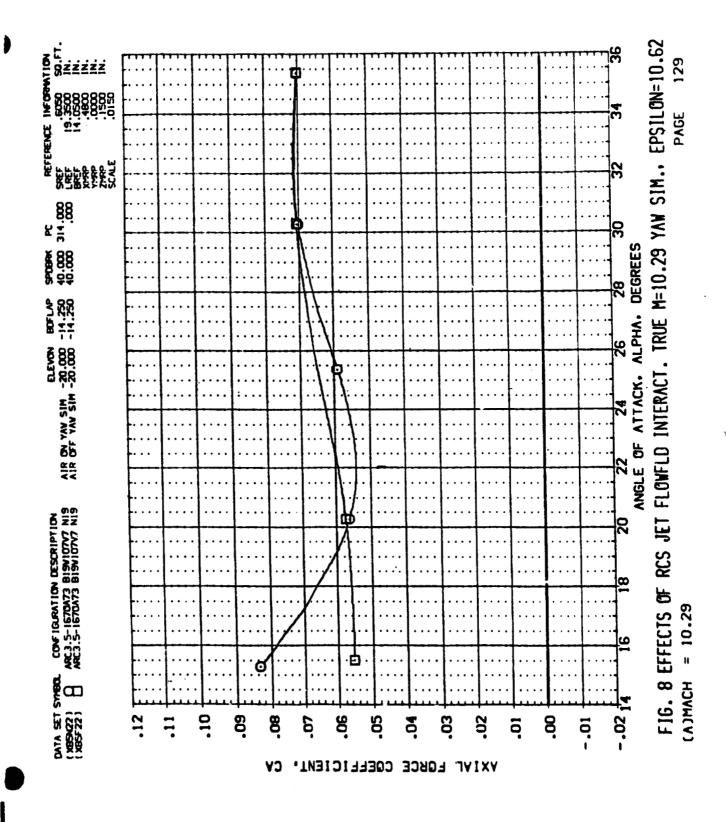




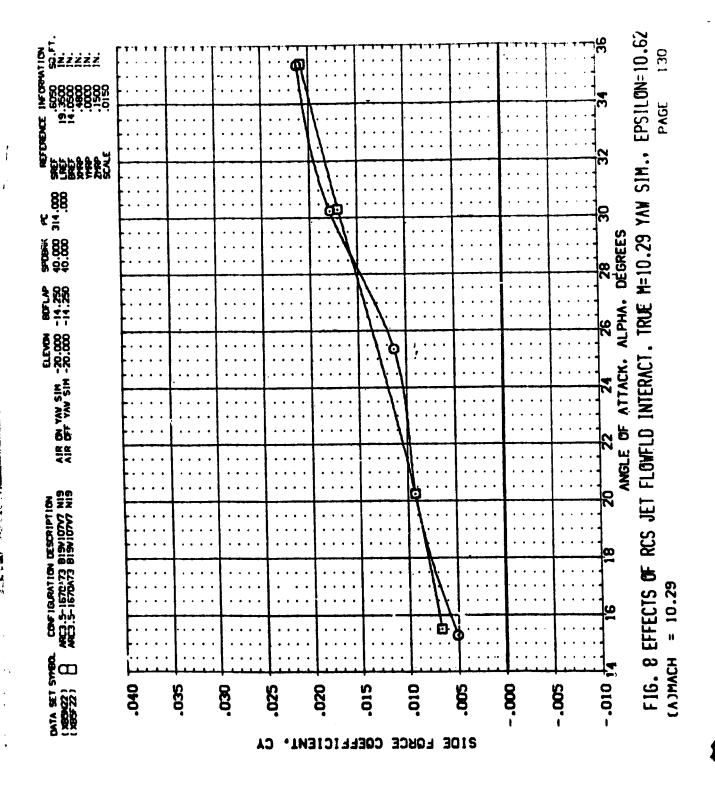


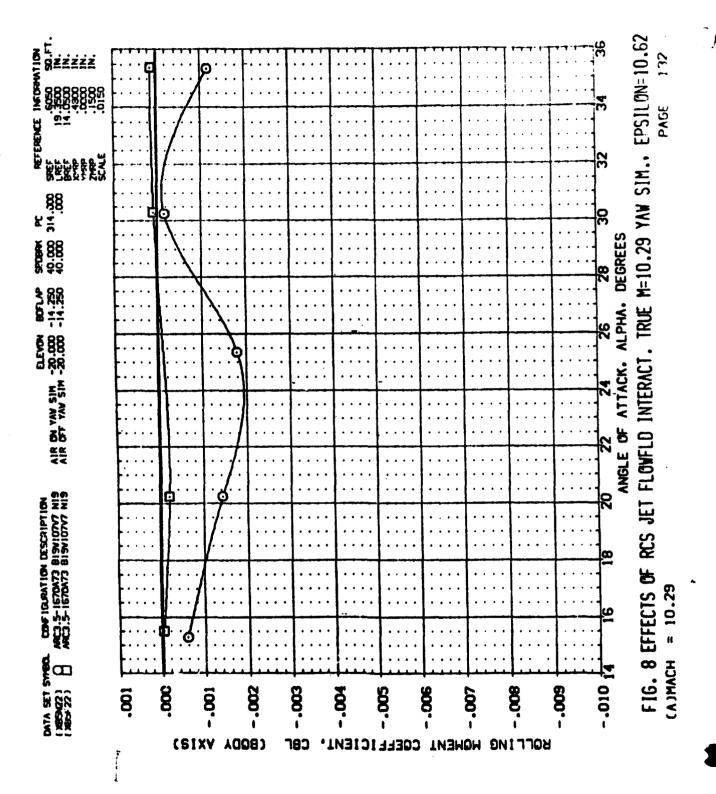


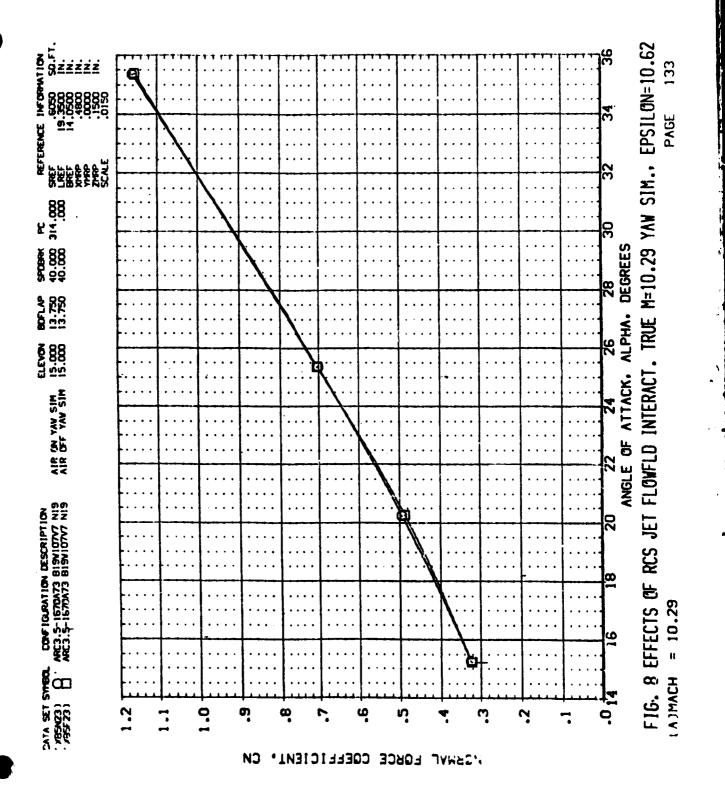


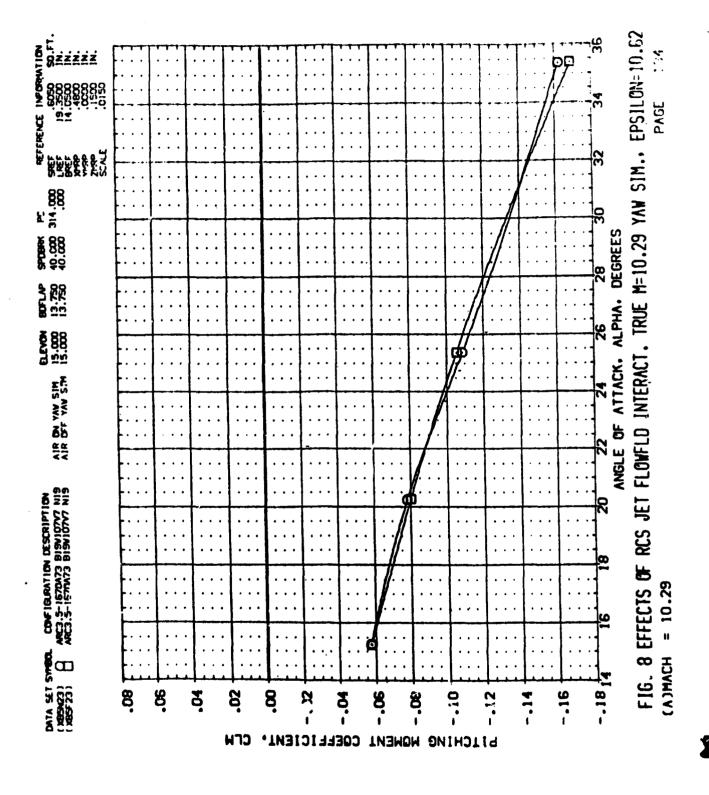


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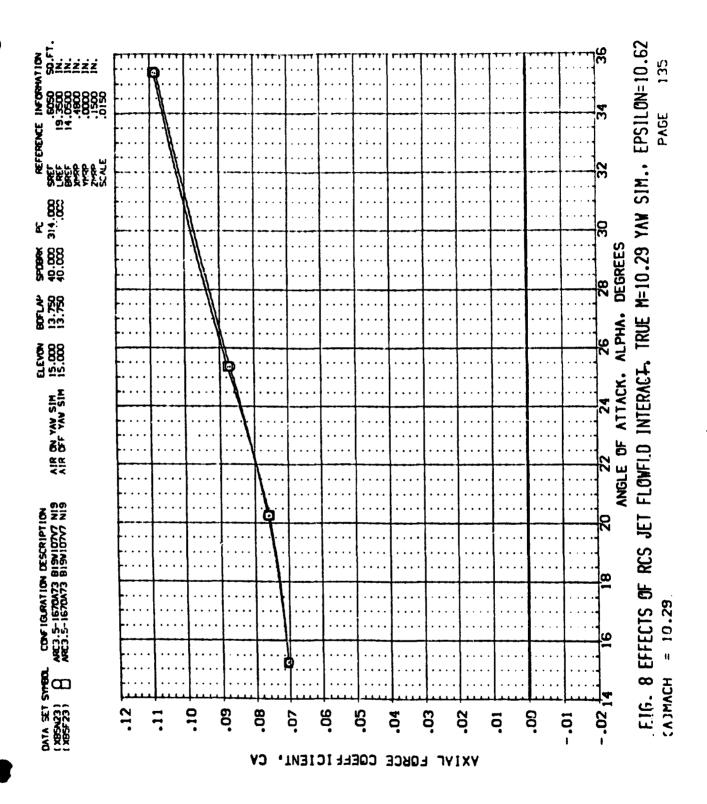


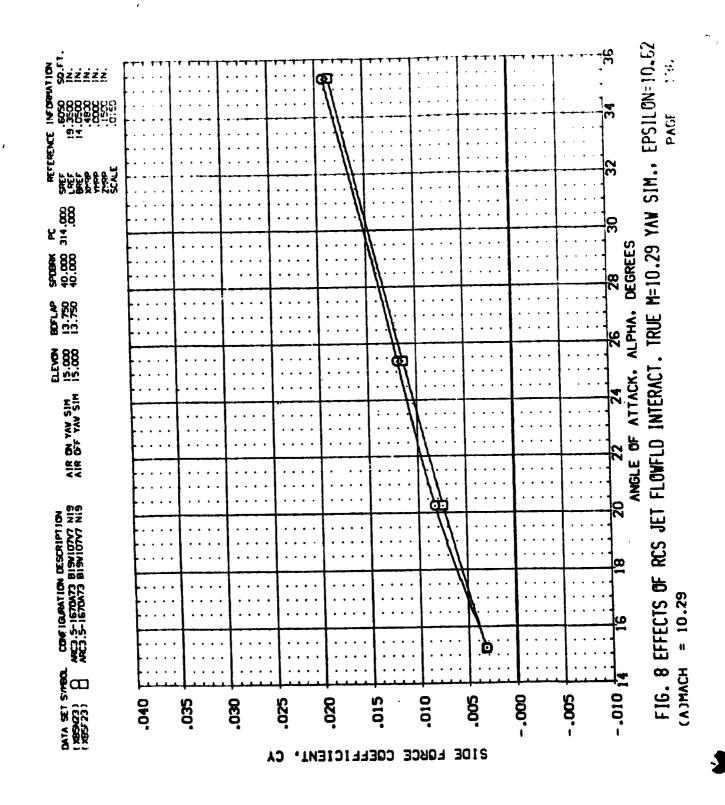




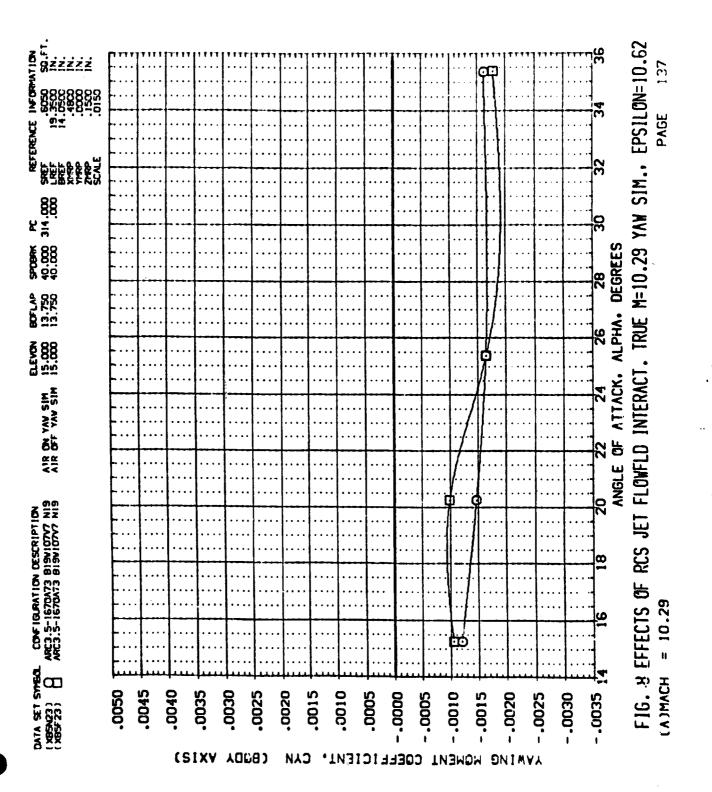


3 3 5

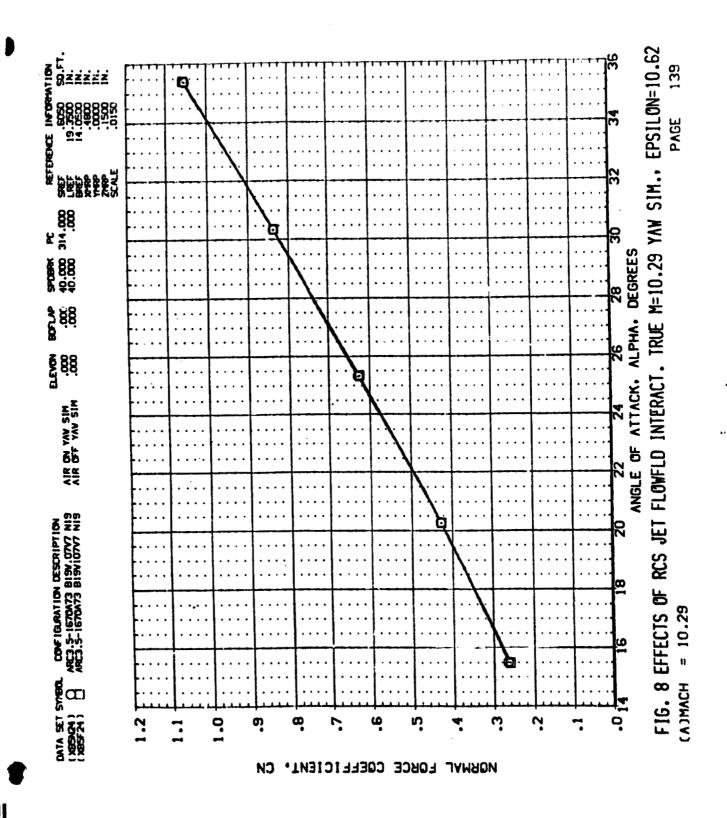


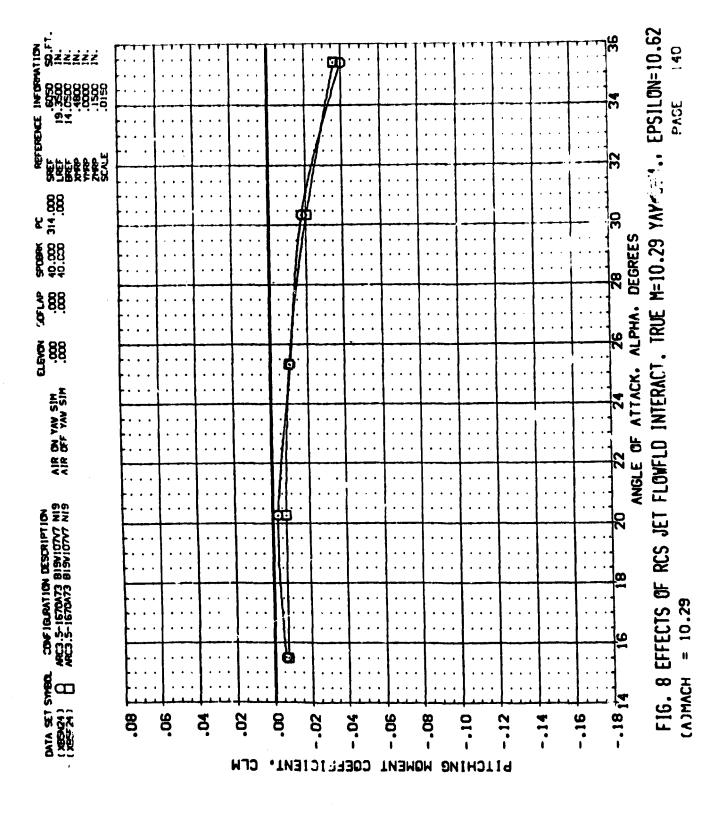


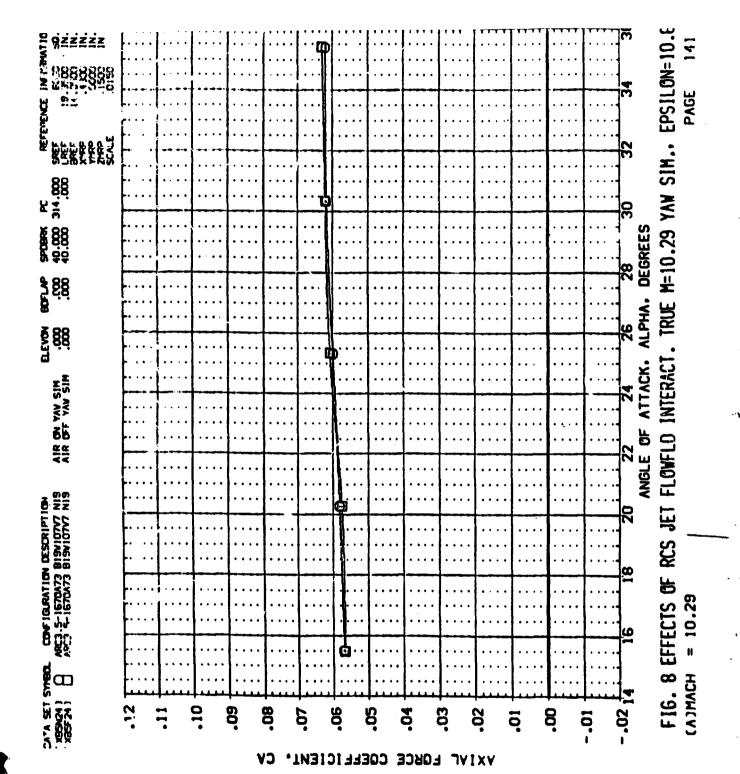
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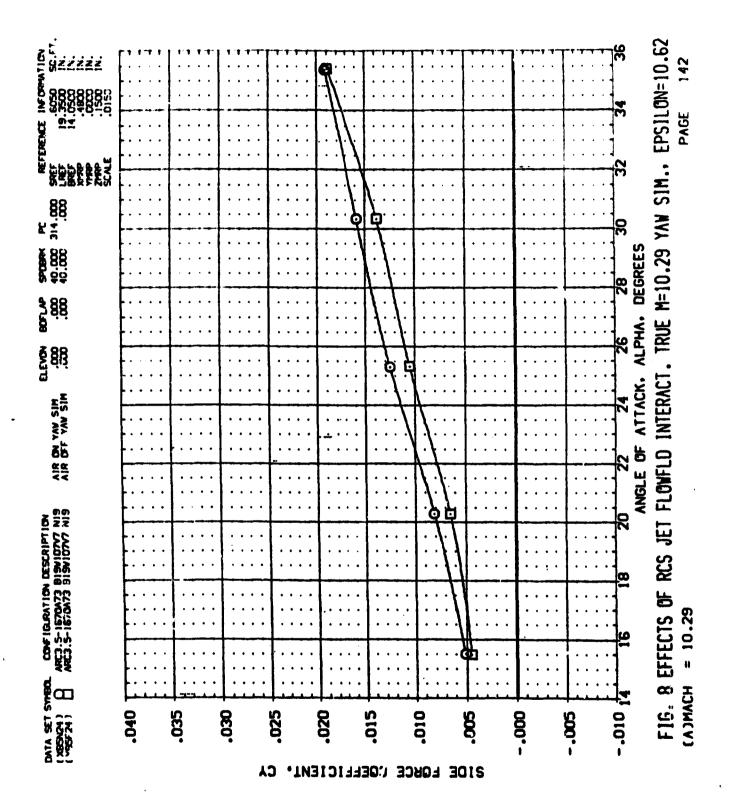
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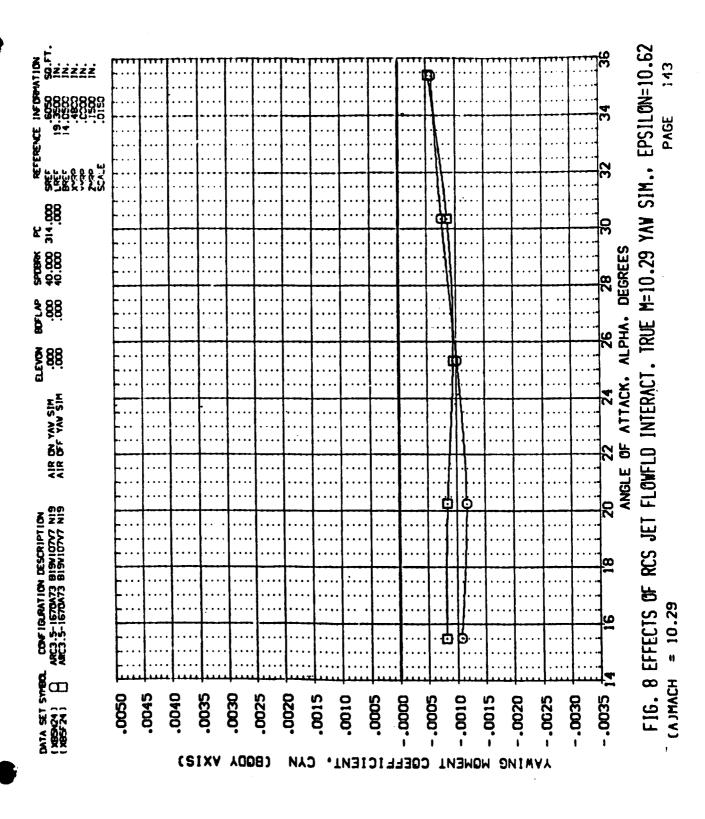


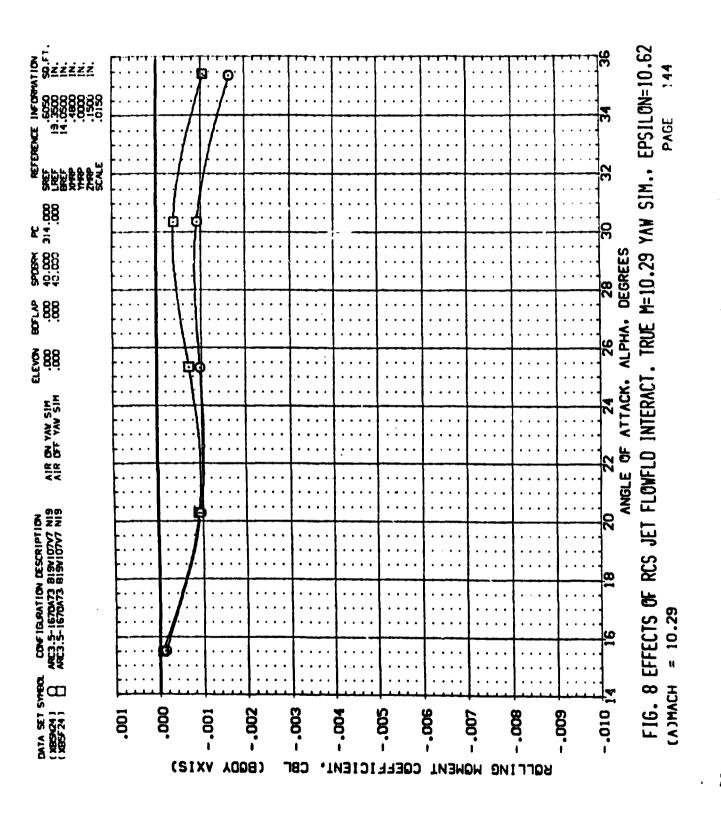


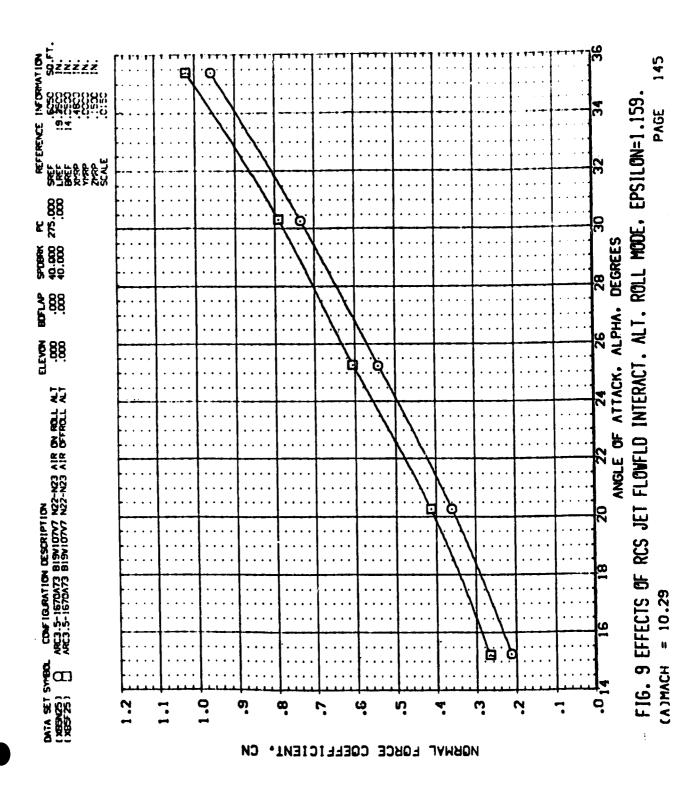
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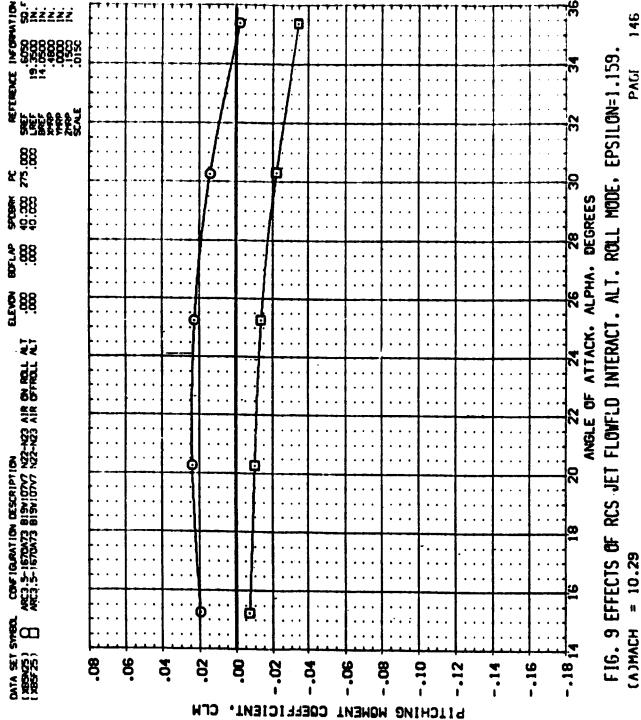
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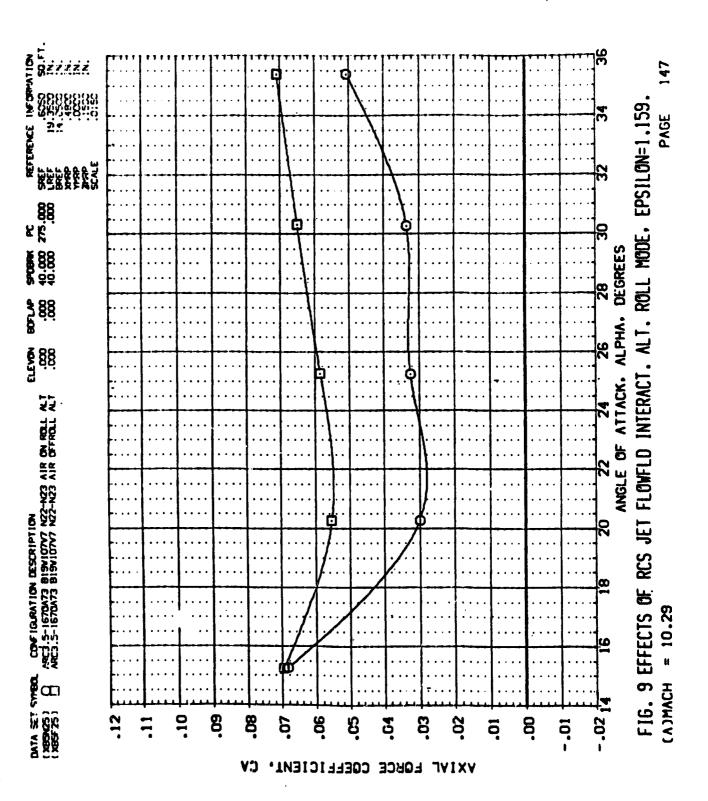


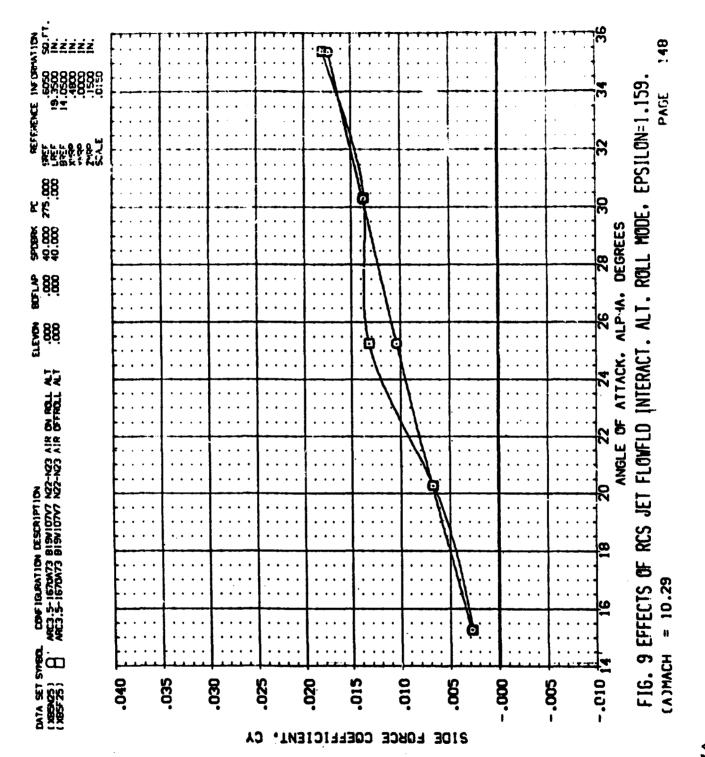


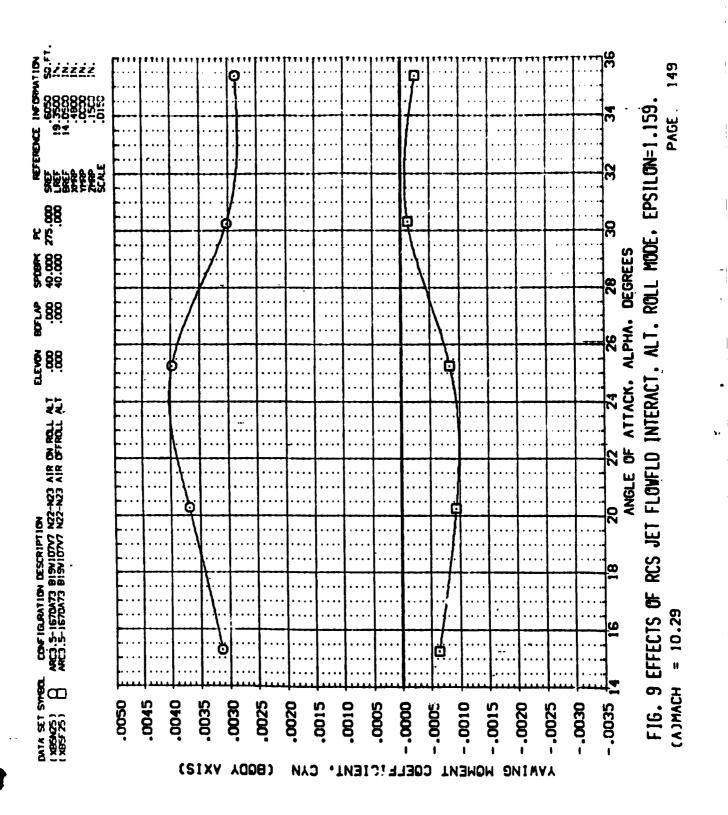
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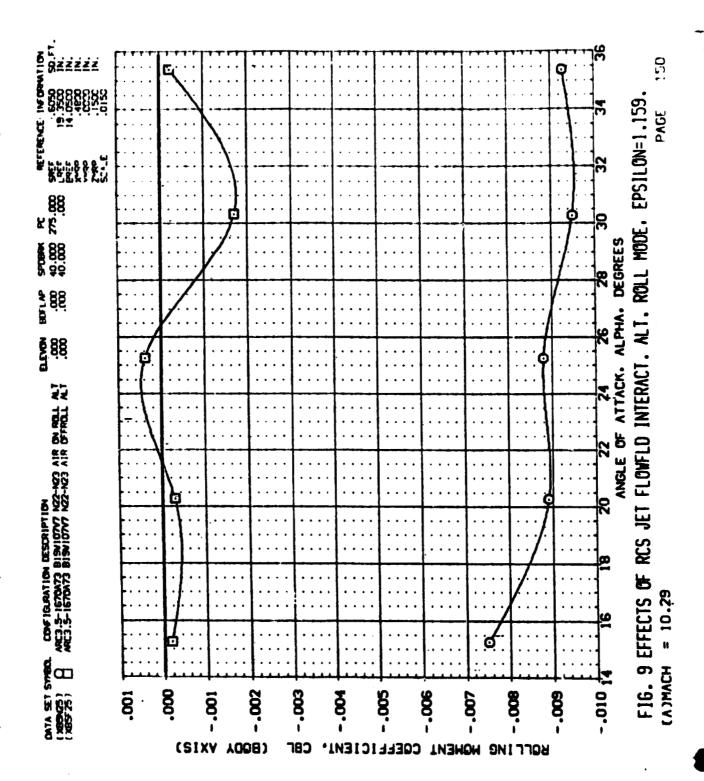
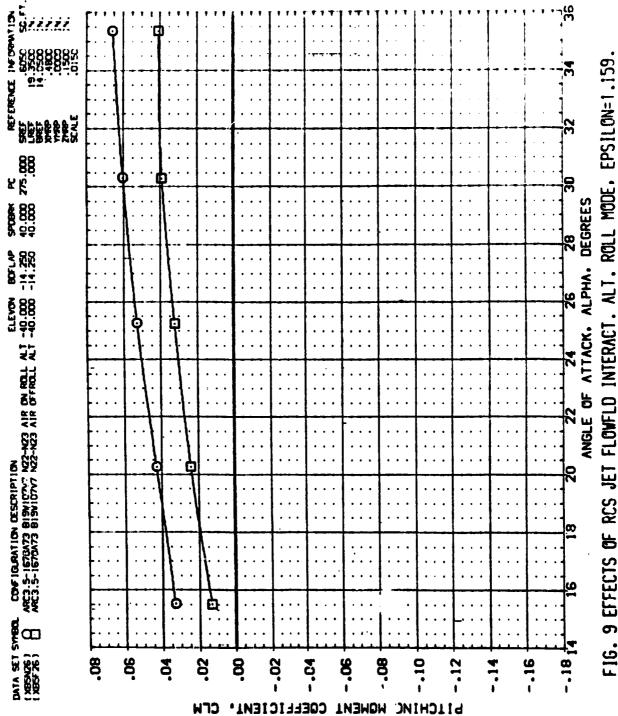


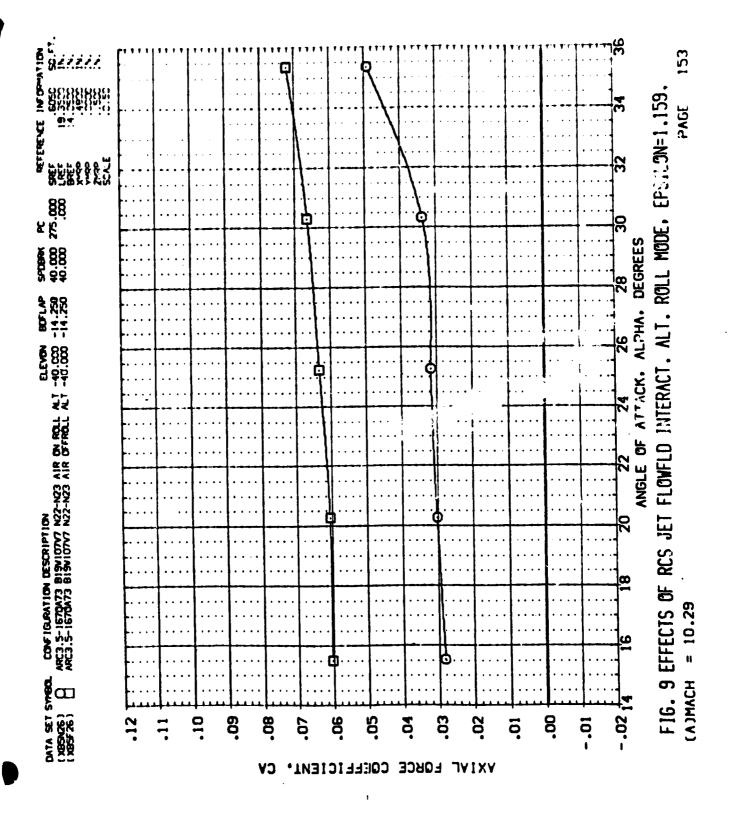
FIG. 9 EFFECTS OF RCS JET FLOWFLD INTERACT. ALT. ROLL MODE. EPSILON-1.159. PAGE = 10.29

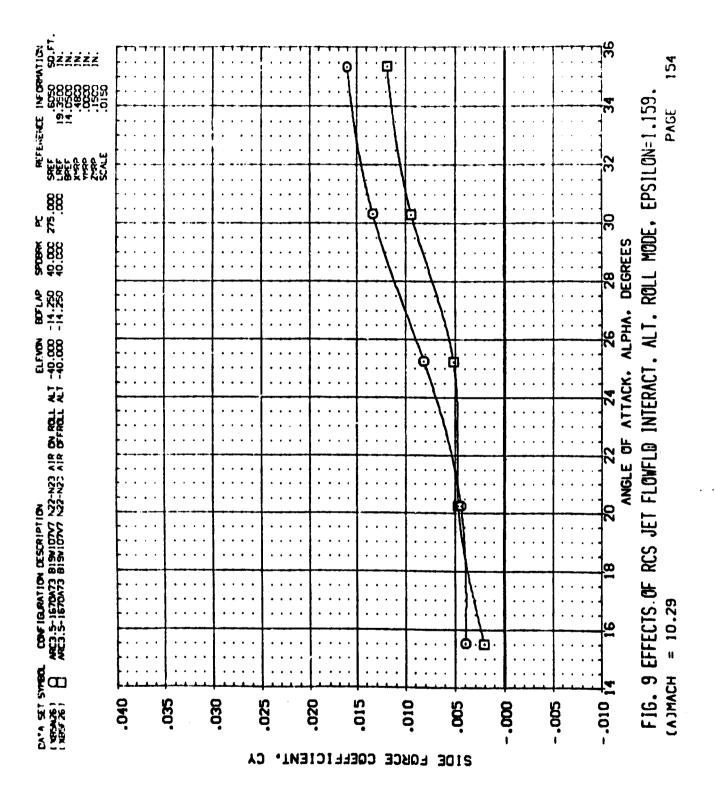


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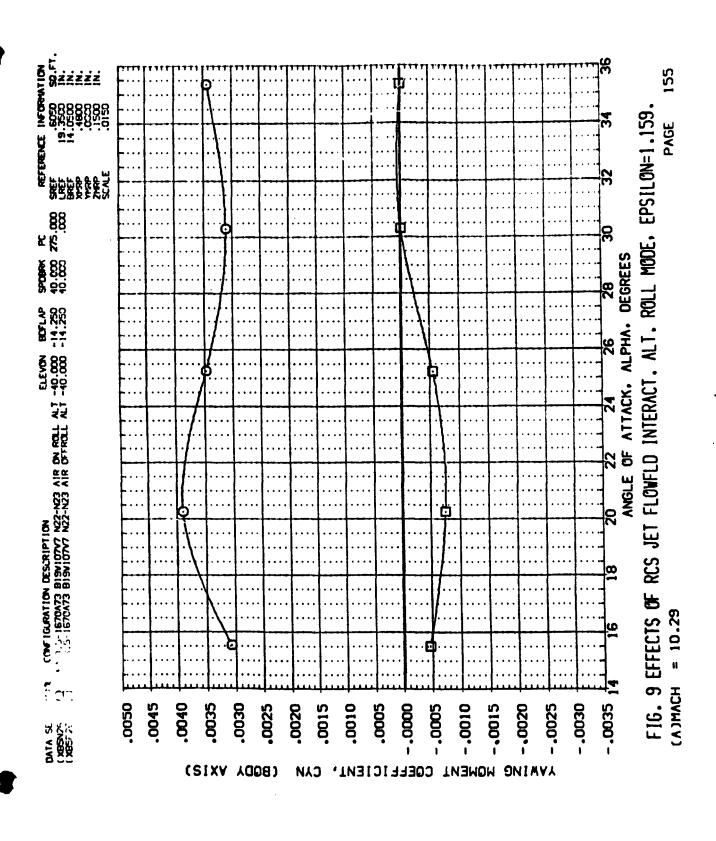
FIG. 9 EFFECTS OF RCS JET FLOWFLD INTERACT. ALT. ROLL MODE. EPSILON=1.159. PASE = 10.29CAJMACH

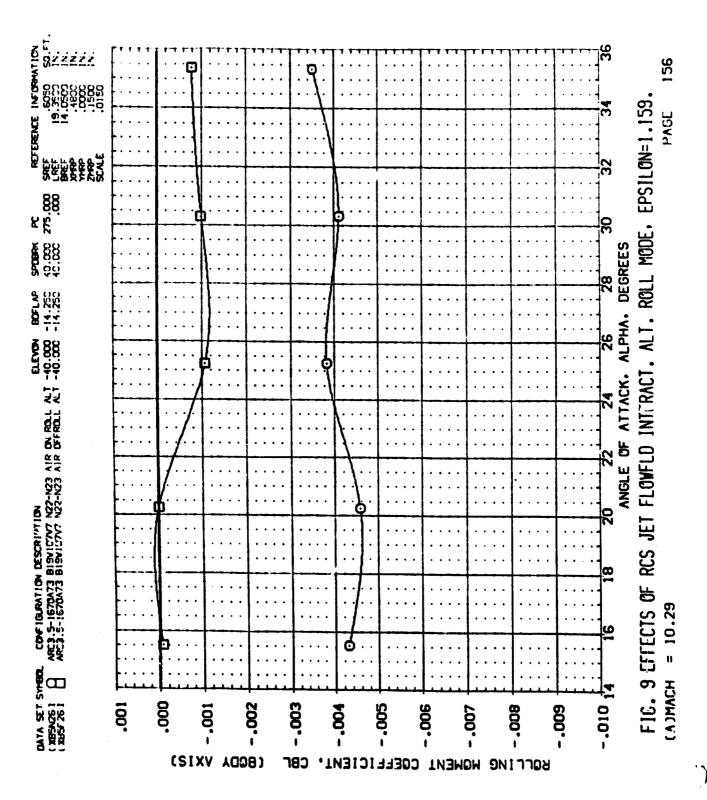
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APPENDIX TABULATED SOURCE DATA

Tabulation of plotted data are available on request from Data Management Services

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CALLEGE S No. 1 WANTED

.24357 .20690 .07319 .07319 .04354 .07686 .000 .000 .000 .000 .000 .1.19 .000. .000.0003 .000.0003 (85 007 73) (RBSF01) (05 OCT 73 PACE 1.07119 1.04024 1.04024 1.05405 1.43020 1.23044 1.92275 1.90117 1.0138 1.99907 1.37492 1.18318 BEFLAP = PSF = TT DEG = EPSLON = 4 PSF = 17 DE6 = EPSLON = PARAMETRIC DATA PARAMETRIC DATA (RB5FD2) CBL -.00038 -.00041 -.00047 -.00068 -.00019 .000 -20.000 40.000 1800.000 .000 18.000 40.000 1900.000 1.720 BETA = ELEVON = SPUBRK = PT PSI = REA. BETA = BLEVON = SPOBRK = PT PSI = REAL = REAL CON -.00072 -.00078 -.00083 -.00083 -.00003 1.94 CRADIENT INTERVAL = 20.00/ 30.00 GRADIENT INTERVAL = 20,00/ 30,00 .00715 .00534 .01344 .01780 .01780 -.04647 -.04647 -.04061 -.04042 -.112617 -.12483 -.00405 AIR OFF YAM AIR OFF YAM .00367 .00347 .00786 .01203 .01320 .00266 .01107 .00175 .01163 .01391 .01392 TABLEATED SOUNCE DATA - ARC 3.9-167 (OATS) ARCS.9-1670ATS BISMAUTY? NEO ARCS. 9-1670A75 B196407V7 NED 0.2732 .11280 .11280 .30763 .30643 .43643 .6260 .02630 0.13480 .13480 .14783 .35274 .8533 .85232 .85232 C. 22139 22139 33431 35431 65644 78443 GGG14 C. #8728 .43420 .43420 .60339 .79947 .1899.1 3 .4600 IN. .0000 IN. .1500 IN. .4600 IN. .0000 IN. .1500 IN. 2 0 3 20.007296 .007241 .00724 .009243 .009243 .009243 .00924 REN RO. 35 M 04 2.5531 2.5537 2.5967 2.5967 3.5059 3.5059 3.5059 3.5059 o. Seros .45024. .45024. .50202. .1.16763 REFERENCE DATA REFERENCE DATA . 6030 86.FT. 19.3500 IN. 14.0300 IN. 9.30 Pe.FT. 19.3900 Pr. 14.0500 IN. 15.220 15.220 20.234 20.337 30.378 85.358 20.323 30.323 33.274 15.492 20.231 DATE 13 NOV 73 10.230 10.230 10.230 10.230 10.230 10.230 16.130 16.130 16.130 16.130 16.130 SALC :: DAGC :: SCALE :: SCALE ::

PAGE 2	(NBSF03) (DS OCT 73)	PARAMETRIC DATA	֓֞֞֜֜֜֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓֓	מאראם	40,000 4 69 = 390,000	1800,000 T7 DE6 a 2000,000	EPSLON =			2	nnnze 1.76690 20184	1.74906	02708	1 M. W.			1.5190	-,000110356300552	6485FUT) (05 OCT 73)	PARACTRIC DATA		۲ "	-40,000 BDFLAF = -14.250		1000,000 TT 0C6 = 2000,000	1.720 EPSLON = 1.199		3	00010 1.7000243642	1.74563	_	-	1.41999	1.22090	P.0950	**************************************
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â										t	,				20110	0350	.010°	19000	H-NES AIR								oter teres	č	000		21600					. 00052
TABLEATED SOURCE DATA - ARC 3.5-16740A73)	ARCS, 9-16FOATS BIEMIGTY? NEG								1.91 GRAD	•	,	1500	11100	19107	Species.	.4479	.62139	.02237	ARCH.S-1670ATS BISHIUTY HEL-HES AIR OFF ROLL								1.83 GM	ε								.0. 21.20
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r		ATAS TARGET	Section 19		_	14.0500 1%.	0£ 10.				*	19.20	19.522	12:02 17:12	100.00	20.36	2	SADIENT		ATA TARREST)		_	_	14.0000 IN.	8.60				22.52	13.40	8.13	22.310	30,33	35.380	GRADIEM
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TABULATED SOUNCE DATA - ARC 3.9-167 (OATS) QATE 13 NO. 73

(MBSFDe) (05 OCT 73)

ARCS.5-1670A73 BISMAD7V7 NEI-NES AIR OFF ROLL

PARAPETRIC DATA	BETA = .000 PC = .000 ELEVON = -20.000 BOTLAP = -14.290 SPDBR = 40.000 @ PSF = 350.000 PT PSI = 1000.000 TT DEG = 2000.000 REAL = 1.720 EPBLON = 1.139
	,4600 IM. ,0000 IM. ,1900 IM.
	n n n
•	7 1 K
REFERENCE DATA	.4090 54.FT. 19.2300 1N. 14.0500 1N.
	SKAE :

RLN NO. 14/ D RU/L = 1.06 GRADIENT INTERVAL = 20.00/ 30.00

74 .23343 23084 23084 23084 23084
1.61336 1.65435 1.45717 1.23624
CBL 00033 .00022 00118 00123
.00733 .02520 .02520 .02520
200. 88130. 8130. 4130. 72730.
62711. 6200. 6200. 6200. 60000.
0. 20274 24.000 75734
2080. 6080. 8080. 7080. 7080.
9. 2023. 26482. 20000.
ALPNA 15,438 25,242 35,340 68,0104
10.290 10.290 10.290 10.290

PARAMETRIC DATA	.000 PC = .000 15,000 BDPLAP = 13,750 40,000 0 PSF = 390,000 00,000 TT DEG = 2000,000 1,720 EPSLON = 1,139
	BETA = .000 ELEVON = 15,000 SPTEMS = 40,000 PT P31 = 1600,000 REA = 1,720
	.4800 IN. .0000 IN. .1900 IN.
	11 11 11 12 12 12 12 12 12 12 12 12 12 1
REPERENCE DATA	.e050 38.FT. 14.3200 IN. 14.0500 IN.
	SECT.

GRADIENT INTERVAL = 20.00/ 30.00

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¥	- 3400	.22053	.1336	20.	2506	2506	-,0139
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ā	05636	-,06074	-,06039	10742	13600	1677	- 60449
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8	.16142	.15402	. 25210	19096.	.57695	.78928	20000
٥	.31187	36763	.45494	14530.	.78466	.91705	70.000
ð	.07401	06680	.07842	.0803e	10306	3211.	
5	STATE OF	32491	.91413	.73174	.96649	1.20472	
		15.450	83.02	25.230	30.254	35.396	
3		10.290	10.290	10.20	10.250	10.230	

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TABALATED SOURCE DATA - ARC 3,5-167 (DATS)

.5-1670A75 BISMIDTY? HEI-HES AIR OFF HOLL

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1			
1			

	MEI-MES AIR OF MALE	PARACERIC DATA	BETA # ,000 PC #	ELEVON # ,000 BOTLAP #	. COC. Cet = 754 9 000,000 a NOCOTE	PT P81 = 1800.000 TT DE6 =	NEA = 1,777 EP9LON =
* * *	ARCS, 9-1670A73 BISMOVY7 MEI-MES AIM OF MALL		.4600 114.	.0000 IN.	.1500 IN.		
107 ENCYC DATA 1003 ENCYC BA 171 18 13500 114 77 10150			*	× det			

BAN NO. 16/ 0 RM. = 1.01 GADIENT INTERVAL = 20.00/ 30.00

۲	29945	.27646	.11984	0203	17745	24728	02807
5	1.93013	1.89665	1.87828	1.64020	1.45331	1.24305	e7650
ŧ	00037	0000	00091	90096	00143	00192	20000
Ē	DDD66	00003	00102	-,00095	-, poor	-,0000	10000
ā	00549	00524	00447	00773	01951	03865	00065
5	· noene	,0000.	.00756	.01226	.01461	. Dt 805	56000
8	.13184	.12440	.ensee	. 32268	.47641	67779.	99530
đ	.25932	\$4623.	36290	.54219	.69237	.63556	20250
5	·ages	.0570	07860.	.Catto	.04532	.06419	66000
ð	.26115	-24D55	A2864	.0000	63613	1.07044	70000
1	15,190	19.44	20.735	13.13	30.279	33,364	CRADITOR
ğ	14:275	16.753	14.190	14.230	13.ENO	04.230	•

ARCS. 5-1670A75 B19A807V7 NEO AIR OFF YAV

NEPENDACE DATA

.e050 84.FT. 19.3850 B4. 14.0907 F4.

PARAETRIC DATA

(87 T30 80) (1178.00)

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\$	2.01532	1.93915	1.09490	1.67903	1.45135	1.24394	04254
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ĕ	00077	00072	00106	00110	00063	00036	-,00001
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t	.00336	1034	9990	.01360	10.	.01953	.00100
8	.12872	12221	.80306	36340	.47863	90249*	.08530
ರ	25942	10863	26690	.54300	.66933	.63600	.09095
5	.0366	01660-	.06744	.0000	.04219	.06430	06000
8	1044	. FREES	43636		. 64239	1.07071	.03874
*	15.184	19.430	20.136	25.330	30.34	35,335	CAADIEM
Ŏ.	10.250	10.730	10.230	30,290	10.290	30.890	

-		.000 -14.290 390.000 200.000 1.199	PC 3.78932 4.25031 4.31990 4.31175 4.05971 04171	- 2	.000, -14.2-30 -14.2-30 -1.1-39	4,00673 4,26082 4,26082 6,4,26982 7,26647 6,30337
(05 OCT 73	4	3 4 20 3	1,53798 1,78198 1,64999 1,441973 1,11909 -,04339	0065F15) (05 OCT 73 /	BOPLA" = 6 PSF " TT DEE "	E.08196 1.83484 1.7126 1.7126 1.26372 1.26386
(1009*14)	PARAVETRIC DATA	.000 PC -20,000 BOF -20,000 BOF -20,000 BOF -20,000 TT -20,000 TT -20,000 BPP	CBL -, 00034 -, 00087 -, 00087 -, 00081 -, 00012 -, 00012	OGGSF15) (,000 -40,000 40,000 1,720	
	2		Crit -,00071 -,00084 -,00084 -,00081 -,00004 -,00006	_	ECTA ELEVON :: SPEDBOX :: PT PS 1 :: REAL ::	CYN -,00057 -,00067 -,00067 -,0007 -,00017 -,00017
AIR OFFEITCH DM			CY CLOS (200.03 × LOURENT INTERVAL × E0.000 / 20.00 CY CLOS (200.03	AIR OFFFITON DR		CT CLH CYN CT CLH CYN THE CURSE OLISSECOUS THE CURSE OLISSE OLISSE THE COUSTCOUSE THE CURSE OLISSE THE CURSE OLIS
			CT			CT CT
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TABLEATED SOUNCE DATA - ARC 3.9-167 (DATS)	ARCS.9-1670A75 B1947UTV	4400 1M. .0000 1M. .1900 1M.	A	9-1670475	.0000 IN.	A 7523. 10. 10. 10. 10. 10. 10. 10. 10. 10. 10.
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### 19	0.000	.12143	76100.	00010	00070	00068	1.94568	.03545
### 25.399	78750	20316	79900	00377	-,00100	00027	1.66994	
### 190	.06132	.xm	99900	00631	00109	00069	1.67460	
### 1.04039 .0003 #################################	.08641	.48945	.01106	-,01669	00078	-,00079	1.44233	•
######################################	.06578	.67731	.01614	03306	00051	-,0006	1.2327	22629
##TUTUTOCE DATA	.000	.02456	29000	-,00090	-,00002	-, 0000	04245	02753
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85.300. C8168. 669.08 021.00. C8470.1 198.28	.00057	32526	.0173	-,00963	00105	00119	1.66314	
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	06490	.67796	16710.	03464	00033	00120	_	•
eecon. soorn. malayas	66000	.02419	.00063	00079	70000	00022	04227	00349

PAGE 9 (RBSFE0) (05 OCT 75)	###	CBL 00026 1 00033 1 00033 1 00073 1 00073 1 00043 1	(20.00 CTM COL L/O PC -,00036 -,00316 1,46233 -,09301 -,00033 -,00333 1,61793 1,5141 -,00013 -,00574 1,4206 ,52291 -,00004 -,00574 1,4306 ,20271 -,00004 -,00590 1,19329 -,01727
AIR OFF TAN	BETA ELEVON SPOBRY PT PS3	> 11111	AIR OFF YAA! SIM BETA = ELEVON = STESSE = FT PSI = REAL =	CY CLM CYM CY CLM CYM 66 .00072 .014850003 66 .01904 .030460003 66 .01904 .030460000 66 .01953 .04186 .0000 66 .01953 .00186
TABULATED SOURCE DATA - ARC 3,5-167 (OAT3) ARC3,5-1670A73 B19A107V7 NEO	.40cm IN. .00cm IN. .1870 IN.	FOV. = 1.55 GRADIEN GL CD C 22619 .14607 . 21226 .11940 . 33273 .19254 . 49747 .30156 . 53647 .44723 . 75615 .62416 .	ARC.,9-1670A73 B19A107V7 N19 .460C ?N000D IN150J PF.	A 1.65 GADIB C. C. C. C. 2.1003 1.4906 1.4916 1.4919 1.4
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DATE 13 NOV ?	SAGT = .0030 54.T. LAGT = 19.3500 16. SAGT = 19.3500 16. SAGT = .0130	MACH ALPHA 10.200 15.205 10.200 20.305 10.200 20.306 10.200 30.406 10.200 35.421	######################################	M.CH ALPHA 10.270 15.330 10.270 25.330 10.270 25.330 10.270 35.438 10.270 35.438

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			MO.	-1eroars s	ARCS.S-1670A75 BISMIGTV7 HIS		AIR OFF YAM SIN	_	(465F22)	2) (05 001 73	, tr 13
	REPORTE DATA	E DATA							PARAMETRIC DATA	DATA	
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		AUN NO.	o. 26 / 0	# 7A	1.55 CRADI	CRADIENT INTERVAL = 20.00/ 30.00	W. = 20.0	00.00 /0			
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10.290	15.20	.25521	.04352	.22223	.14732	.00361	.00942	-,00065	00027	1.90845	-1976
10.290	15.495	14953.	:0550:	SEATH.	.11633	9969. 7267.	9690G.	-,00047	02000	1.63520	2.66m3
10.290	20.249	. 33507	8 /ca-	69469	90477	96910	02625	£1000 -	.0000	1.39724	14225
10.290	8. S.		2000	75216	95029	09020	.02756	CNODE4	*1000.	1.21203	1.06960
10.00	CANDIDA	00000	00000	00000	00000	00000	00000	00000	GGGGG.	.00000	00005
			ARC3.5	-1670A73 B	ARCS.9-1670A73 B194107V7 M9		AIR OFF YAU SIM	_	(Rb5F23)	3) (05 OCT 73	Б
	NEFERENCE DATA	E DATA							PARAMETRIC DATA	: DATA	
	. FO 50 54.FT.	Page .		.48CD 1H.				BETA :	ODU:	" "	80.
95	19.3900 IN.	-	200.	OCCUS IN.				PEON =	19.000	# WE	13.750
SCALE ::	14,0900 IN.	À	n K	1960 JW.				P1 P3 L4	1900.000	11 056 =	200.000
							:	1	1.720	- NOTEG	10.00
		RE IO	26 0	. 78	1.18 SAND	CRADIENT INTERVAL = 20.00/ 30.00	¥.	8.8. 8.00			
9	*	8	5	đ	8	5	ā	£	ŧ	2	ĭ
10.20	15.234	.32257	.07067	99262	.15295	.00321	-,05759	-,00105	CO:192	1.91348	45266
10.290	15.450	07440	99610.	.06642	, 03696	.0278	.00669	00521	0002	1.70461	.26472
10.80	20.231	.48653	.07556	.43016	19263	.00743	07997	0008	00117	1.75464	1.65737
062.01	29.372	70480	.08771	199824	.301E4	.01133	10548	00165	90000	1.57100	02162.
00.00	35.374	1.19997	.10095	.67950	, 798DA	.01057	16049	00179	-,(10233	1.16022	
	CHADIENT	08290	.00230	.00302	.02789	.000.	00496	00013	22 000.	-, D4356	27894

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DATE 15 N	<u>;</u>	TABLE	ATED SOLA	ICE BATA - AC	TABULATED SOURCE DATA - A4C 3.9-1674CA731	131				ť	PAGE 11
			*	103.5-16704PS	A9(3,5-1670A73 B19/J107V7 M19		AIR OFF TAN SIN	*	(#B9F2+)		(05 007 75)
	REFERENCE CATA	ב כי.י							PARAMETRIC CATA	C CATA	
-	19 55 65		•1	ABDS IN.				B£1A =	8	יי ע	.930
	N1 0000 01	:		C PUBLISH.				z	000	COSTAB =	900
	14.0500 IN.		,	.NI OCS1.				SPEER :	40.000	254 4	380,000
4.4	0510.							PT PSI =	1000,000	17 CE6 =	2000,000
								۳ ۲	1.72	EPSLON n	15.623
		RUN NO.	6 . 35/ 0	a Type o	1.48 GRA	DIENT INTER	CAADIENT INTERVAL # 20,00/ 30,00	00' 30'00			
2	V (244	5	3	ಕ	8	Š	ā	Š	턴	6	ž
10.230	15.273	27826	27660.	.25209	13094	.00333	-,00900	06000*-	00021	1.92962	69731
10.290	15.478	.29901	.05675		.12362	.00447	00714	190001-	GDDD6	1.69372	
10.290	20.25	42617	.95753	•	.20132	67900	-,00724	00064	16000	1.88516	.07559
10,290	25.334	.62503	.06785	•	.32279	.01053	01013	-,00095	00072	1.67167	01224
10,290	30,359	.63576	.06224	•	.47608	.01384	02D42	-,00086	60030	1.44680	.00533
10.290	35.406	1.05790	.06313	•	.66439	66810.	03616	00084	00105	1.24274	8 CC91.
	GRADIENT	.03931	.00065		.0236	.00079	-,00057	30000*-	£0000°	04203	01729
			ŧ	:3.5-1670A73	ARC3.9-1670A73 B1964U7V7 NEE-NE3 AIR CFFRCIL ALT	E-NZ3 AIR	OFFROLL ALT	_	(RB5F25)	25) (05 OCT 73	, E
	REPERENCE DATA	E DATA							PARAMETRIC DATA	: DATA	
-	.eoso se.rī.	\$.		.46CD IN.				e VEIA	8		000*
	19.3500 IN.	A		.0000 IN.				ELEVON ::	00.	BOFLAP =	900
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SCALE :	0610.							PT PS] =	1.720	11 DEG = 1	1.159
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204	46.4	5	ð	đ	8	5	*	Ē	Ē	5	۲
10.230	15.237	66993	.06965	23930	15757	.00265	00702	-,00063	00017	1.74194	60612
10.290	15.462	.24470	.05413	.22141	.11741	.00477	00936	00072	£1000.	1.86576	.17923
10.290	863°02	.41242	.05536	.36775	.19472	.00666	,00991	+6000*-	00027	1.68657	.09471
10.290	25.240	67409.	.05849	. 92200	.31097	.01322	01357	-,00064	.00039	1.67860	22217
10.290	30.327	. 79053	.06512	.64948	.45537	.01370	02205	00012	00171	1.42627	29198
10.290	35,363	1.02096	.07097	. 79130	.64903	.01606	03444	00026	00023	1.21921	60612
	CHADIENT	.03944	.00062	.03062	.02323	.00131	00073	-0000	.00013	04196	-,04534

CA CASO 14. CA CASO 14.	21 79Vd	FFROLL ALT (1895'26) (05 OCT 73)	PARANETRIC DATA	-	, ,	# # #	GRADIENT INTERVAL = \$0.00/ 30.00	CYN OBL 1.47872	0005000006 1.73977	1.60276	90100 - 9000	-,00001 -,00099 1,41633	1.20644	.0000400021	(RBSTE7) (DB CCT 75)		PARAMETRIC DATA	-	N = -40.000 BOFLAP =	11	1 70 100	15 ED, UDV 30, UD		1,6009	.0142700046 1.74612	.01232 00036 30310.	.0044700076	THETT DODA? DODA	00025 - 00000 - 1.42360	The state of the s	20801.
TABALATED SO TO SEE : 2000 :		9-167 (OA73)	MOVE HE						5715						•	BISMUTYT NEI-NES AIR													29945		.44109 .01505
		SOURCE DATA - ARE S.	ARCS. 9-1670573 B19		.4800 IN.	.1900 IN.	NO. II ROVL B	1	-							CAOCA 1670A73			AT CASE	. N. 0000	NI 0061.		28 0		•	-					. Della.
		TABULATED		IDECE DATA		:	•							•				DESICE DATA					1								25.312 .57005

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				ARC3.	9-1670A73	ARCS.9-1670A73 BISHIDTV7 NEI		AIR OFFEITCH DN		(RB5F28)		(05 001 73)
	REFERENCE DATA	E DATA								PARAMETRIC DATA	DATA	
CALE ::	.6090 56.FT. 10.3500 IN. 14.0500 IN.	7. X 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4 4	11 H H	94. 700.	.4600 IN. .0000 IN. .1500 IN.				BETA ELEVON S SPUBBK S PT PSI S REAL S	.000 -40.000 40.000 1600.000	PC 2 BOFLAP :: Q PSF :: TT DEG :: EPSLON ::	.000. 14.250 350.000 800.000
		RUN NO.		%	. 7	1.56 GRA	DIENT INTE	GRADIENT INTERVAL = 20,00/ 30,00	00' 30'00			
104	ALPHA	5	5		d	8	5	3	N. C	8	۲/٥	
10.290	19,189	25744	66290	£ £	20103.	1202	06600.	.01224	-,0000	11000	1.011036	-,01333
10,230	20.50	.39291	8	06035	34762	19287	26600	.02476	00072	-,00047	1,60233	06570
10.290	25.295	.57335	.06354	25	.49124	.30241	.0100	.03D42	90000	00137	1.62444	39735
10,290	30.262	.77622	Ş.	06390	22.	.44010		.0396Z		CZION'-	1.42610	40717
10.250	35.387 GRADIENT	.03567	.0000.	. 8	. 76115	.02177	.00067	21100.	.0000	00018	03536	-,06592
				ARCS.9	-1670A73	ARCS.5-1670A75 B194807V7 NZD		AIR OFF YAU		(RBSF2	(RBSF29) (05 OCT 73)	ב ב
	RETERBACE DATA	E DATA								PARMETRIC DATA	DATA	
	.F.98 98.FT.	988 .F		4.	.4600 IN.				ETA ::	000	۲ ۲	600.
5	19.3500 IN.			9	.0000 1M.				ELEVON =	40.000	BOFLAP =	-14.250
SCALE ::	0510	}	1						FT PSI ::	1.720	TT DEG =	1.159
		RUN NO.		35/ 0	# Y#	1.50 GRADIENT INTERVAL = E0.00/ 30.00	ler tyre	VAL = 20.0	20.00			
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10.250	19.84	.esme	.06261	5	.23152	.12796	.00423	.01356	-,00033	00032	1.60907	62367
10.230	15.462	12962	.09976	Ę	.21173	12051.	\$64QD.	.01145	00026	00011	1.75613	.11922
10.210	20.25	.39565	.06023	2	33032	.19353	.00766	.02251	-,0009	-,00033	1.01116	16515
10.230	23.321	. 57453	.06336	96	.492£4	.30300	.00932	.03227	-,00036	-,00068	1.62453	29964
10.290	30,331	. 77007	.06581	5	.63143	.44968	.01528	.03672	toppo.	76000	1.41679	33476
10.230	39.364	.97306	.06710	10	.75447	.61617	.01848	04075	.00026	-,00094	1.22049	40461
			-								1	

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Manual Color Col	DATE 13 NOV 75	2	TABILA	ITD SOUNCE	DATA - ARC	TABALATED SOUNCE DATA - ARC 3,5-167 (DATS)	=				2	PAGE 11
1				MC3.	9-16TOATS	SISMOTY? HED		M VAL		(A6 9M)		- R D
1,000 0a, 17, 0app 1,000 1h, 0app 1,000 0a, 17, 0app 1,000 0ap		BEDEROKE .	E DATA							PARAMETR 1C	DATA	
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14.0000 N. 14.000 N.		- 00 OE OE	-	4.	100 IN.				H 4	oca.		9000
1	} !	**			10° 10°.					-20.000	BEN'LAP :	3
	•	10.000		•	2				SPEEKK :	40.200	# PSF &	350,000
		14.0500 EM.		•					PT P\$1 #	1000.000	11 CE6 =	2000.000
State Stat		0610.							REA =	1.720	DISTON =	1.139
1.0 1.0			RCS0 N				ाठम असक्त	W. = 20.(00° 00° /00			
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13.441 129503 129503 129504 129504 139504 1	Ş	15	5	, į	, 1	27.43	0000	ecoto.	00129	00223	1.87878	274.06700
1.0	10.E80	13.61		200	12021			10007	00162	-,00178	1.85611	276.80200
Second S	10.290	19.481	2287	0.000.	.21267	. 1140		20.00	- 00103	00313	1.00079	273.78400
### ### ### ### #### #### #### #### ####	10.230	20.240	Series.	.05462	.54513					- PRINT	1.66155	277.30100
### 90.371	10.210	25.310	96060	.05619	10004		TOB ET		P. Commission		44779	
### 13.322	10.250	20.37	78560	79090	.64722	.44857			, coo			
### ### ### ### ### ### ### ### ### ##		20.00	07906	.06164	ret.	.62166	.02522	69030	00097	~,00396	1.23832	273.33400
### ##################################			P. DE-PLU	07000	69000	. OE273	66000	5.000.	90000	-,00000	03924	19269
### ##################################												
Section Sect				WEG.	.9-1670A75	BISMOTY? NE		PWA NO		7689A		
## 18-3500 IN. 1989		DOGG	E DATA							PARACTRIC	: DATA	
# 14.0000 IN. 1400				1	2				BETA =	980		273.000
# 19,3500 IN, Tree Library Libra			•						E 500 =	15,000	# OV-1408	000.
1		19.3500 IM.	-	n :					Section	40,000	n 759 0	390,000
### .0130 #### ###############################		14.0800 TN.	•	# #	. M. OO				1	1800.000	11 DE6 =	2000,000
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DA73)				LADIENT INT	.00627		.01667		.02445	NEI-NES AIR			MOIENT INTE	5	22100.	.00941	.01250	00000
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	DATE 13 HOV 73	2	TABLEA	TED SOUR	IE DATA - AR	TABLEATED SOURCE DATA - ARC 5.5-167 (DATS)	ê				Ē	PAGE 16
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DATE 13 MOV 73	7484	1,4750 9	OM .	14TA - 48C	TABLLATED SOLR "ATA - ARC 3,5-1670A73) ARC3,5-1670A73 BISMIDTV7 M21-M23 AIR ON MOLL) - HE S AIR (TOL M		(789410)	2	PAGE 17
	REFERENCE DATA								PARAVETRIC DATA	c DATA	
.e030 34. 14.0300 1N. 14.0300 1N.	.6030 34.FT. 208P 19.3500 IN. 198F 14.0300 IN. 208F	# # #	4.00 08.1.	.4800 IN. .0000 IN. .1500 IN.				ELEYON :: SPOBRK :: PT P81 :: REAL ::	.000 .000 40.000 1800.000	PC BOTLAP II TO BE II EPSLON II EPSLON II	294.000 .000 390.000 2000.000
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	CALLES.	Caged 5	ñ	90000	. 29131	10800	.01477	.00327	00654	1.72963	
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PAGE 10	M . (1899H12) (04 OCT 75)	PARAMETRIC DATA	•	.000 BOFLN" = 36 40,000 @ PSF = 35 1600,000 TT DEC = 200 1,720 BPSLOM =		형	0001900551 2.10442	1,12569 1,0367 1,05648	20036 - County 1, 71764	.00001	1.26192 322	21490 00000 04715	20000	HIDN (TRESNES) (DA OCT 75)	ATMO SIEME	COLUMN TO THE TAXABLE PARTY OF THE PARTY OF	۲ ا ا	25.000 burns		NEA = 1.720 Daton	20°00/ 30°00	5	CTN	THE 00742 1.90112	213050 1.60215	1.42604	1.0694 1.22343 32	00016	
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	TABLEATED BOUNCE DATA - ARC 3.9-167 (DATS)	ARCS. 5-167Chr3 esement		,4600 1M. ,0000 1M. ,1900 1M.	,	18/ 0 MML =		T2632.		13960	26106				ARCS.9-1670A73 B19A8D7V7 NET			s ,4600 IN.	0000 JW.		1	10 mm.	•	96769	1000				- 1010°
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325.16100 351.30300 359.34900 -.nomi **325.75300** 326.13500 332.01900 352.01900 325.73600 350.000 2000.000 1.159 326.76400 (NBSM14) (04 OCT 75) -14.290 (MB3N15) (D4 OCT 73) L/D 1.50862 1.61506 1.64628 1.43180 1.23099 E.28197 E.07786 1.78900 1.51269 1.28159 -.05690 PARMETRIC DATA PARMETRIC DATA ...00406 -..00438 -..00438 -..00473 -..00673 .000 -20.000 40.000 1800.000 -.00234 -.00234 -.00234 -.00232 -.00332 40,000 1600,000 ELEVON :: SPORK :: PT PSI :: REAL :: .00012 .00004 ..00012 ..00010 ..00010 ELEVON = SPESSIK = PT PSI = NEA. = .00092 .00014 .00014 .00001-1.52 GRADIENT INTERVAL = 20.00/ 30.00 20.00 yo.03 AIR ON PITCH DN AIR ON PITCH DN 60.00. 60.00. 60.00. 60.00. 60.00. 60.00. .02707 .02694 .04564 .05736 .05736 CRADIENT INTERVAL .01004 .01004 .02043 .02077 .02050 TABULATED SOURCE DATA - ARC 3.5-167 (DATS) AACS, 9-1670A73 BISMIDTY7 NEL ARCS. 9-1670ATS BISHADTV7 NEI .13660 .18603 .25061 .43574 .63574 .06171 .1 5125 .1 5125 .1 5125 .1 1127 .57634 .02163 15X7 a .33039 .47900 .62390 .74670 **1** a. .20607 .21278 .33508 .46108 .62213 .74120 .4600 1N. .0000 1N. .1900 1N. .0000 IN. .1900 IN. ğ ر تار ه .09639 .08944 .06190 .06357 .03363 .03707 .04708 .04208 <u>5</u> **35** 50. O. 252424. 25728. 257.92. 250408. 250408. . 95011 . 95011 . 95011 . 74471 . 99869 REPENDICE SATA REPERENCE DATA . 19.3500 1N. 19.3500 1N. 14.0500 .N. . 6050 58.FT. 19.3500 78. 14.0500 JW. 25.257 30.264 35.300 20.236 DATE 15 NAV 7.1 10.230 10.230 10.230 10.230 10.230

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BATE 13 HOV 73	Ş	TABLE	ATED BOUR	ICE DATA - A	TABILATED BOUNCE DATA - ANC 3.9-167 (DATS)	(C)				•	PAGE 21
			\$	IC3.9-1670A7	ARCS.S-16TOATS BISMIDTUT NES		AIR ON PITCH UP	ı.	(0 tag@u)		ו מו מנו מו
	GIAGIGN	NEPENDICE DATA							PARAMETRIC BATA	: Data	
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10.230	25.339	.0241	.09571		.31750	-,00326	00649	200.	-,00363	1.70190	•••
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		·	Ę	C3.9-1670A7!	ARCS.9-16TOA75 B1941DTV7 NED		AIR OF YAN		(MESM 9)		(or 120 to)
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10.20	15,493	•1914	4550.	.22400	.1173	.01267	.0000	-,00220	-,00339	1.90923	291.22000
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10.290	30.29	.63230	.0997g	.68648	.47148	.02221	01991	-,00065	00578	1.46026	283.12370
10.130	35.412	1.03000	1130	. 62709	.06428	.02576	-,0800	-,00117	00746	1.24513	207.95700
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	(02)(6)(1)	PARAMETRIC DATA	86	-40.000	40.700	1.720	,	룡	-,00132	-,00062	-,00206	-,00266	-,00347	-, m409			9698		PARMETHIC DATA	8	4		000	1.720		ŧ	-,00029	00070	-,00119	-,00127	·		
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a							iton imek	ŧ	ל	Cardina.	1110			16910	600	36000									DIENT INTE	1	5						
1.5-167 COATI	ISAROTAT HE						1.96 CALDIENT INTERVAL = 20.00/ 30.00		8	.14267	1120	.1678	.23726	13627	.61410	27.120.	Meantry? H			٠					1.6		8				. ASTES	1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1 1	
TABLEATED SOURCE DATA - ARC 3,5-167 WATS)	ARCS. 9-1670A73 BIBMIOTY7 NED			.48CD 1W.		<u>i</u> B	* 7ª		đ	enu.	2080X	34290	.49107	. 62.534	75.58	SCHOOL	Oth Cyclin and Property and				AMO 194.	000 IX.	.130 IN.				ರ	mu.		E.	.62161	ret.	5 903 0.
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				-	COST CONTRACTOR		110 747 1				
		į	¥	3.5-1670A75	arcs.b-1670ats bigaiotyt mið		MIN ON ANY RIN		(22mcom)		
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10.290	30.284	. 75303	.070	.61500	.44024	.01787	.6060	-,0001	-,00016	1.39697	330,30500
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0.20	19.23	.82316	16040	.28365	.15304	.00313	05733	00120	-,00043	1.91612	329.00000
10.230	20.23	. 48541	.07624	.43696	. 24250	92800	07810	00147	00069	1.02257	
10.290	25.374	,7000v	.0870e	.98783	37990	16110.	10873	00165	00102	1.57392	329.66600
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DATE 13 NOV 73	2	TABULAT	TO SOURCE D	IATA - ARC	TABLATED SOUNCE DATA - ARC 3.5-167 COATS)	•				Ž	7. 23 P. 16. 24
			ARCS.5	-1670A73 B	ARCS.5-1670A73 BISMOTY7 HIS		AIR ON TAN SIN			(1859Rt.) (04 OCT 75)	. tr
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10.250	15.309	1					50.00	00117	-,00094	1.00109	330.52500
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			₹	9-1670A73 (ARCS, 9-16FOATS BESAGOTY HEE-NES AIR ON ROLL ALT	PARS AIR (N ROLL ALT	•	(526501)	55 (C C C T T T	c t
	HEPENDACE DATA	E DATA							PARMETRIC DATA	C DATA	
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24 AON 61 21VO	Q 73	TABLE	NTED SOURCE	DATA - ARC	TABILATED BOURCE DATA - ARC 3.9-167 (DATS)	ŝ				č	PAGE 23
			FFG3	.9-1670A73	BISMOTYT R	ARCS.S-1670ATS BISMOTYT NEZ-NES AIR ON ROLL ALT	N BOLL AL	-	(NESTRE)		(Be OCT 73 ?
	HEPENENCE DATA	E DATA							PARAVETRIC DATA	: DATA	
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10.230	30.317	.7400	38000		-	,01337	15004	11000	00411	1.54454	200.7950
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	CAADIENT	.03550	06000	.02893	.0210 2	16000	. מסט.	.00009	00002	D4742	.13473

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PAGE 28	X 73)		375.000 -14.250 390.000 2000.000		7. 28. 4. A. S. A.	394.45500	394.06400	395.01309	394.76600	394.66100	-	. e		375.000	-14.250	350.000	8000.000	1.190		¥	394.96000	394.29800	394.29000	394.35100	394.57800	394.63100	.01020
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Ę				60.00 INTERM. = 20.00/ 30.00	5	- DE 264	298.	9000	1100	10.00	2000.								CLOIDH DIFFINE =	t	101314	.01291	.01ee	Cross.	.0236	.02786	Stoop.
TABILATED BOUNCE BATA - ARC 8.8-107 (DATS)	ARCS.9-1670A73 BISHADTYT NEI			1.57 GEA	8		1		2107	. 98078		ARCO.O-1070ATS BROMBOTY NED							1.30 62.0	8	.11940	.11540	10001	18087	.436bt	. 40423	#130.
DATA - ARC	.9-1670A75		.4600 1M. .0000 1M. .1900 1M.	4	4	1970	OLASK.	.4698		. 758%	2007	6-1670A79		.4600 IX.	.44 0000	.1500 IM.			3	8	. 2196	Seatt.	. Soone	.46063		.74346	.05839
NTED BOARCE	SAC			. 74	5	q	-Deg-	0.	.D4588	.04543				₹	6.	.18			9	5	.08785	.08431	.00eez	.00007	19090	.04227	36000
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£		MOTOR	.6030 36 19.3500 1H. 14.0500 1H.		A S	15.49	20.23	13.13	30.137	36.360	MADIDA		ADDEDICE DATA	.T 0900.	19.3600 IN.		06 a5.			*	19.54	15.47	80.88	25.310	30,387	35.37	RACIENT
DATE 13 HOV 73					i i		10,250	10, 890	10,110	10,210	-						CAL .			ğ	10.250	10.830	10.230	10.230	10.00	10.290	•